

RESEARCH MEMORANDUM

EFFECTS OF SPOILER AILERONS ON THE AERODYNAMIC LOAD

DISTRIBUTION OVER A 45° SWEPTBACK WING AT

MACH NUMBERS FROM 0.60 TO 1.03

By Joseph M. Hallissy, Jr., F. E. West, Jr., and George Liner

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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

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SUMMARY

An investigation was conducted with 73-percent-semispan inboard spoiler ailerons, projecting 4 percent of the local chord from the wing surface, and located at the 70-percent-chord line of a 45° sweptbackwing—fuselage combination. The model consisted of a wing with an aspect ratio of 3.98, taper ratio of 0.61, and NACA 65A006 airfoil sections parallel to the plane of symmetry in combination with a fuselage of fineness ratio 10. Pressure data were measured on the wing and spoiler at several spanwise stations at Mach numbers from 0.60 (Reynolds number 5.1×10^6) to 1.03 (Reynolds number 6.2×10^6) for angles of attack that usually extended to 20° or more.

Upper-surface spoilers resulted in normal-force decrements which were largest in the 0.6 to 0.8 semispan area of the wing for low and moderate angles of attack. Most of this decrement was associated with increased upper-surface pressures ahead of the spoiler, but some of the decrement resulted from decreased lower-surface pressures. The addition of a gap through the wing behind the spoiler relieved the low pressure behind the spoiler on the upper surface and thus increased the rolling effectiveness. Lower-surface spoilers give reversed rolling-moment effectiveness at angles of attack higher than about 10° primarily because there is a large decrease in pressure behind the spoiler on the lower wing surface at the higher angles of attack.

Spoiler loads were highest at the inboard end. For upper-surface spoilers, the loads on the spoiler decreased rapidly with increasing angle of attack, but for lower-surface spoilers they increased to the highest test angles of attack.

INTRODUCTION

Recently there has been considerable interest in spoiler ailerons, primarily because they maintain rolling effectiveness through the transonic speed range (ref. 1) and because they provide high reversal speeds for thin flexible wings. In addition, they can be designed to have very low hinge moments. However, very few pressure data for spoiler configurations have been available at high subsonic and transonic speeds (refs. 2 and 3) for load calculations or for studying the effects of spoilers on the flow about a wing.

Hence, a systematic test program has been initiated in the Langley 16-foot transonic tunnel to provide pressure data for various spoiler configurations in the transonic speed range at moderately high Reynolds numbers for a large angle-of-attack range. The initial investigation of this program was conducted on a 6-percent-thick 45° sweptback-wing—fuselage combination at 0° yaw and Mach numbers from 0.60 to 1.03. The spoilers investigated on this wing were of the retractable type, and they extended along the 70-percent-chord line from the fuselage (14 percent of the wing semispan and had a projection from the wing surface of 4 percent of the local wing chord.

This paper presents results for some of the configurations that were tested during the initial investigation. The effects of upperand lower-surface spoilers and of a wing gap behind the spoilers are shown on the wing normal-force characteristics, chordwise pressure distributions, span-load distributions, and centers of load. Also included are tabulated wing static-pressure coefficients, spoiler pressure distributions, and spoiler span-load distributions. A limited amount of summary data from this investigation has already been published (ref. 4), and six-component force data obtained simultaneously with the pressure data are presented in reference 5.

SYMBOLS

| Ъ | wing span |
|----------|-----------------------------|
| c | section wing chord |
| <u>c</u> | average wing chord |
| c † | wing mean aerodynamic chord |

$$c_m \frac{c^2}{\overline{c}c'}$$

wing-section pitching-moment parameter with moment about the 25-percent position of the mean aerodynamic chord,

$$\frac{c^2}{cc}$$
, $\int_0^1 (P_L - P_U) \left(\frac{x_1}{c} - \frac{x}{c}\right) d\frac{x}{c}$

 $c_n \frac{c}{\overline{c}}$

wing-section normal-load parameter,

$$\frac{c}{c} \int_{0}^{1} (P_L - P_U) d\frac{x}{c}$$

 $c_{n_S} \, \frac{\underline{h}}{\overline{h}}$

spoiler-section load parameter which acts parallel to plane of symmetry and perpendicular to h at a given spanwise station,

$$\frac{h}{h} \int_{0}^{1} (P_F - P_R) d\frac{z}{h}$$

 $C_{\mathbf{N}}$

wing-panel normal-force coefficient,

$$\int_{0.135}^{1.0} c_n \frac{c}{\overline{c}} d\frac{y}{b/2}$$

C,

rolling-moment coefficient

h

length of chord of exposed front face of spoiler at any spanwise station

 $\overline{\mathtt{h}}$

average h, $\frac{(h \text{ at } 0.14b/2) + (h \text{ at } 0.87b/2)}{2}$

M

free-stream Mach number

Ρ

pressure coefficient, $\frac{p - p_0}{q}$

р

local static pressure

Po

free-stream static pressure

q

free-stream dynamic pressure

x

distance from wing leading edge at a given spanwise station, positive downstream

distance from wing leading edge at a given spanwise station to line perpendicular to plane of symmetry and passing through 25-percent position of mean aerodynamic chord, positive downstream

longitudinal location of wing-panel center of pressure, $\frac{\int_{0.135}^{1.0} \left(c_m \frac{c^2}{\overline{c}c'}\right) d\frac{y}{b/2}}{c_N}, \text{ fraction of mean aerodynamic chord}$

y spanwise distance from the plane of symmetry

lateral location of wing-panel center of pressure, $\frac{\int_{0.135}^{1.0} \left(c_n \frac{c}{\overline{c}}\right) \left(\frac{y}{b/2}\right) d\frac{y}{b/2}}{C_N}, \text{ fraction of semispan}$

distance measured from wing surface along h at a given spanwise station (not perpendicular to x- and y-axes)

angle of attack of fuselage center line relative to test-section center line

ΔP change in P across spoiler at a given spanwise station (P at 0.65c - P at 0.75c)

Subscripts:

F forward surface of spoiler

L lower surface of wing

R rear surface of spoiler

U upper surface of wing

APPARATUS AND TESTS

Tunnel.- The investigation was conducted in the Langley 16-foot transonic tunnel, which is a single-return wind tunnel having a slotted throat of octagonal cross section. The maximum variation of average Mach number was about ±0.002 along the test-section center line in the vicinity of the model. Additional details of the test-section configuration and of the calibration of the tunnel are given in reference 6.

Model.- Figures 1 and 2 show details of the basic model and the spoiler configurations included in this investigation. The steel wing had NACA 65A006 airfoil sections parallel to the plane of symmetry, quarter-chord line sweep of 45°, taper ratio of 0.61, and aspect ratio of 3.98. The wing was designed to have no incidence, dihedral, or twist and was mounted in a midwing position on the fuselage. The fuselage, constructed of steel, had a fineness ratio of 10, and the quarter chord of the wing mean aerodynamic chord was located at the longitudinal position of the maximum fuselage diameter.

The spoilers for these tests (fig. 1) simulated retractable spoiler-aileron configurations pivoted about the 50-percent-chord line. These spoilers were located along the 70-percent-chord line of the wing and were projected four percent of the local wing chord from the wing surface. They extended from the fuselage (14 percent of the wing semispan) to the 87-percent wing semispan and had a sweep angle of 41.6°. Spoilers were tested without and with a gap in the wing behind the spoiler. The gap, when used, extended outboard from the 15-percent to the 87-percent wing semispan station. The lower-surface spoiler with gap configuration was obtained by inverting the model with upper-surface spoiler and wing gap. The oppositely deflected spoiler configuration had one spoiler mounted on the upper surface of the left wing and one on the lower surface of the right wing with no gap behind the spoilers.

Figure 2 shows location of the wing static-pressure orifices which were distributed over the left wing at seven spanwise stations. The orifices at the inboard station (average $\frac{y}{b/2} = 0.135$) were actually located on the fuselage 0.1 inch from the wing surface. Pressure orifices were also located on the front and rear surfaces of the left wing spoiler and in the wing gap behind this spoiler at five spanwise stations as shown in figure 2. The pressures were transmitted by means of small tubing through the model support system to mercury manometer boards.

Model support system. - A cantilever strut, described in reference 7, supported the sting-mounted model. The model was near the tunnel center line at all angles of attack. A straight coupling between the sting and

the model permitted variations in the angle of attack from -4° to 15° ; a 10° coupling extended the range.

Tests.- Tests were generally made for all configurations at angles of attack of -2° to 26° for Mach numbers from 0.60 to 0.90. At Mach numbers from 0.94 to 1.03, the maximum angle of attack of these tests was limited by sting-support stresses or available tunnel power.

The Reynolds number variation over the Mach number range of the tests is shown in figure 3.

DATA REDUCTION

Data reduction methods.— Extensive use of a punched-card system greatly facilitated the reduction of data. Pressure data recorded with manometer board cameras were first transferred to cards by the use of a commercially available manual film reading device coupled to a card-punch machine. The data were then processed on electronic computing machines to obtain individual pressure coefficients as well as section normal-force and pitching-moment coefficients (using a rectangular step integration). The data cards were then fed to an automatic plotting device for the preparation of the chordwise pressure plots of this paper and were also used to prepare the tables of pressure data.

Corrections. The angles of attack presented include an adjustment for an incremental angle determined from static calibration of the model angular deflection as a function of pitching moment and normal-force loads. Based on the repeatability of deflection measurements made during the static calibrations, the estimated maximum error of the angle-of-attack measurements is to.1°. No corrections were made for tunnel-flow angularity. The cumulative effect of model asymmetry and tunnel flow angularity is shown to be small by the basic model normal-force curves in figure 5. No corrections were made for sting interference. Sting interference was not considered of importance for these tests because all lateral-control configuration changes were made on the wing, which was relatively remote from the sting. The data have not been corrected for tunnel boundary-interference effects since the results of reference 8 indicate that these effects would be small.

In calculating wing section and panel coefficients, the effect of the forces on the spoiler was neglected. The magnitude of the error thus introduced was checked and found to be within the accuracy of the data.

RESULTS AND DISCUSSION

Static-pressure measurements are given in coefficient form for the basic wing and the several spoiler configurations as follows:

| Table | Static-pressure coefficients for - | Page |
|-------|---|-----------|
| I | Basic wing | 19 to 43 |
| II | Wing with upper-surface spoiler (no gap) | 44 to 63 |
| III | Wing with upper-surface spoiler (with gap) | 64 to 88 |
| IV | Wing with lower-surface spoiler (with gap) | 89 to 104 |

Each table shows the pressure coefficients at seven spanwise wing stations for various Mach numbers and angles of attack. Some of the high-angle-of-attack data for the upper-surface spoiler without gap and for the lower-surface spoiler with gap have been omitted since these configurations are of less interest because of loss of effectiveness or reversal at high angles of attack. No data have been tabulated for the oppositely deflected spoiler configuration, since, as will be shown, these were only slightly different from the data presented in table II.

In discussing the test results, some of the more important effects of spoiler operation on panel and section characteristics are first noted. Subsequent discussion makes use of chordwise pressure distributions to illustrate the manner in which the various spoiler configurations affect the air flow and the load distribution over the swept wing. Finally, the effect of a lower-surface spoiler on the opposite wing loading is considered and loads on the spoilers are discussed.

In figures which show comparisons at one angle of attack, the angle of attack given is an average for the compared configurations. This average does not differ more than 10.15° from the extreme value for any of the compared configurations.

Effect of Spoiler on Wing-Panel Loading

Since the rolling-moment coefficients produced by the spoiler configurations are given in reference 5, there was no need to integrate the present pressure data for rolling moment. In order to be certain,

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however, that such integrations would indicate the same rolling moments as were measured with the strain-gage balance, integrations have been carried out for a few pressure-data points. A comparison with the balance measurements is made in figure 4. Even though the chordwise pressure force rolling-moment component (about the stability roll axis) has been neglected, as have crossover effects from the left wing to the right wing, agreement between the two sets of data is generally excellent. It is concluded, therefore, that an analysis of the pressure measurements made in these tests will help to provide a correct understanding of the functioning of a spoiler as a lateral control.

Figure 5 shows the variation of integrated wing normal-force coefficient with angle of attack at all test Mach numbers for the basic wing and three spoiler configurations. As in reference 5, the upper-surface spoilers caused decreases in normal-force coefficient which were largest in the region of 4° to 6° and which became small at the higher angles of attack. The presence of a gap through the wing behind the spoiler provided an additional decrement which was more or less constant throughout the angle-of-attack range. The beneficial effect of such a wing gap at transonic speeds has been previously shown. (For example, see ref. 9.) A lower-surface spoiler provided an increase in wing normal force at the lower angles of attack but a reversal was indicated for angles of attack above about 10° as was also shown in reference 5.

The increments in normal-force coefficient in figure 5 caused by operation of the spoiler were about twice the magnitude of the lift-coefficient increments shown on a similar figure in reference 5 for the same configurations. The reason for this effect was that the lift coefficient in reference 5 was based on total wing area, whereas the normal-force coefficient in this report was based on the semispan wing area.

Figures 6 and 7 show, respectively, the longitudinal and lateral locations of wing-panel center of pressure for the basic wing and three spoiler configurations at all test Mach numbers. These curves show discontinuities at some low angle of attack in both longitudinal and lateral center-of-pressure locations. These discontinuities indicate that, for an upper-surface spoiler, there was extensive positive loading inboard and negative loading outboard for the condition of zero panel lift.

Changes in center-of-pressure location caused by operation of the upper-surface spoiler were generally less for the spoiler with no gap than for the spoiler with a gap. At moderately high angles of attack, the center-of-pressure locations were affected only slightly by the spoilers, although the spoilers were still effective in reducing normal force at these angles. (See fig. 5.)

Effect of Spoiler on Wing Section Loadings

Figure 8 presents the semispan load distributions for the basic wing and the three spoiler configurations at all test Mach numbers. For the lower angles of attack, when the spoilers were most effective, the important loading changes caused by the spoiler occurred outboard of 0.3 semispan, the largest decrements being in the region between 0.6 and 0.8 semispan. This large influence over the outboard regions of the wing was the cause of the inboard position of the panel center of pressure for uppersurface spoilers (or outboard for lower-surface spoilers) at low positive load conditions. The loss in effectiveness for the upper-surface spoilers as the angle of attack was increased beyond 6° is evident as a reduction in the load decrement beginning near the tip.

Addition of a gap through the wing behind the spoiler produced an added decrement in section normal-load parameter which extended across most of the semispan.

Figure 9 shows the wing-section center-of-pressure locations across the span of the wing at all test Mach numbers. For a swept wing the line of section centers of pressure tends to be somewhat less swept than the wing itself as shown by the basic wing data in this figure. The effect of adding the upper-surface spoilers was to exaggerate this tendency, the local center of pressure being farther rearward inboard and farther forward outboard. This effect was most noticeable at low angles of attack ($\alpha = 4^{\circ}$ and 6°) but rapidly decreased at higher angles for two reasons: The spoiler decrements were smaller in absolute magnitude at high angles of attack and were a smaller proportionate amount of the total normal force.

The lower-surface spoiler at low angles of attack had the opposite effect, that is, the center of pressure was more forward inboard and more rearward outboard at low angles of attack ($\alpha = 4^{\circ}$) than it was for the basic wing.

Figure 9 also shows that the section center-of-pressure locations moved ahead of the leading edge at low angles of attack for the upper-surface-spoiler configurations. This movement occurred whenever the section normal force approached zero while a finite section moment remained.

Chordwise Pressure Distributions

In the discussion which follows, selected chordwise pressure distributions for the basic wing and the three spoiler configurations are used in conjunction with unpublished tuft photographs obtained on one spoiler configuration (upper-surface spoiler - no gap) to permit the study of the effect of the spoilers on the flow over the wing.

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Upper-surface spoiler without a gap. Figure 10 shows the chordwise pressure distributions obtained at seven spanwise stations at several Mach numbers. For a Mach number of 0.60 (fig. 10(a)) and low angles of attack, the presence of the spoiler resulted in increased pressures on the upper surface ahead of the spoiler. The highest pressures were immediately ahead of the spoiler, the peak being in the corner ahead of the spoiler-fuselage juncture. This peak was sharp, but elsewhere along the spoiler face a relatively flat peak about 0.1 chord in width indicated a separated flow region. Tuft studies confirmed this result and showed that the flow on the wing surface in this region was toward the tip and parallel to the spoiler.

Behind the spoiler on the upper surface the pressures (fig. 10(a)) and tuft studies also indicated a separated flow condition. Immediately adjacent to the fuselage, the separation extended only a few percent of the chord behind the spoiler and was followed by complete recovery of the flow. Since the expansion or turn made by the air over the top of the spoiler was very abrupt, the pressures reached at this point were quite low, being about the same as those reached at the leading edge at high angles of attack. Somewhat farther out on the wing (at 0.25 semispan), the extent of the separation was greater, recovery being made at about the trailing edge. Because of the increased extent of the separated region, the air turned down toward the surface less abruptly than at the more inward locations, and the pressures reached were not as low. Outboard of 0.25 semispan, complete separation of the flow behind the spoiler was indicated by the flat pressure distributions. The pressure in this region steadily increased toward the wing tip but was always less than for the basic wing.

Apparently the flow about the spoiler was somewhat similar to that described in reference 10 for a two-dimensional supersonic case and as illustrated in figure 11 but combined with the circulatory flow patterns in the boundary layer was the outboard movement of the boundary layer in the regions ahead and behind the spoiler. The pressure distributions were somewhat similar to the two-dimensional case, and the circulation in the separated areas also appeared to be present. Ahead of the spoiler this circulation was evidenced by the spoiler face pressure distributions to be discussed later, whereas behind the spoiler the direction of the tufts indicated that the air on the wing surface was moving toward the spoiler (in addition to moving outboard). Since this circulation was combined with a spanwise movement, the flow may be described as two vortex-type motions, one ahead of the spoiler and the other behind the spoiler.

The lower-surface pressures on the wing, as shown in figure 10(a), were not changed in the inboard areas because of the presence of a spoiler on the upper surface. At about 0.30 semispan, however, the separated area behind the spoiler on the upper surface reached the trailing edge and from

this point on the semispan to the tip the pressures on the lower surface were reduced because of the influence of the upper-surface flow. Outboard of 0.55 semispan the reduction in lower-surface pressure extended all the way to the leading edge.

As the angle of attack was increased, it was apparent that the effect of the spoiler on the wing did not change appreciably until angles of attack were reached where the flow separation on the basic wing began to progress inboard from the tip. (See fig. 10(a).) At these angles of attack, the spoiler effectiveness was reduced, as would be expected, since raising the spoiler into a separated flow region where the air is already moving parallel to the spoiler should not have any effect. (A discussion pertaining to the flow over the basic wing may be found in ref. 11.)

As the Mach number was increased from 0.6 at zero angle of attack (fig. 10(a)), there was no change apparent in the way which the spoiler affected the flow until about Mach number 1.0 (fig. 10(e)). Beginning at this speed the lower-surface influence was less extensive, being generally confined to the area behind the 0.70-chord line. At higher angles of attack the upper-surface influence ahead of the spoilers was also less extensive for the higher speeds. This effect was mainly noticeable over the inboard stations and was probably caused by the presence of a shock wave associated with the separation point ahead of the spoiler. The presence of this shock wave would have opposed the transmission of the pressure increase ahead of this point except outboard where the boundary layer was considerably thickened.

The effects of these pressure changes on the rolling moment were as follows: The pressures on the upper surface ahead of the spoiler and on all the lower surface changed in a direction to decrease the lift and thus contribute to the rolling moment, whereas the pressure decreases behind the spoiler on the upper surface were adverse. Inboard this adverse contribution was enough to completely offset the small favorable contribution which occurred at these stations. Thus, the span load distributions (fig. 8) show the largest normal-force decrements to have occurred outboard (between 0.6 and 0.8 semispan) and that the contribution of that part of the wing inboard of 0.3 semispan was generally unfavorable. These pressure changes also indicate the reasons for the discontinuities in the panel center-of-pressure locations shown in figures 6 and 7. Zero panel lift occurred at a small positive angle of attack and for each angle of attack the unaffected inboard sections were positively loaded, whereas the outboard, sections, where the spoiler was most effective, were negatively loaded.

The ineffectiveness of the spoiler inboard of 0.3 semispan is not too important since the roll moment arm is small, but the large pressure difference across the spoiler near the fuselage is undesirable since it

contributes heavily to the drag. These adverse roll and drag effects inboard may indicate that an improvement could be obtained by removing the inboard part of the spoiler, but there is a likelihood that the region of adverse effects would then move outboard. In other words, the trouble may be associated with the inner end of the spoiler more than with the inboard area of the wing.

Effect of a gap through the wing behind an upper-surface spoiler .-Figure 12 compares chordwise pressure distributions of the basic wing with those of the spoiler-with-gap configuration throughout the Mach number and angle-of-attack range, and figure 13 compares the configurations with and without gap at a few points.

It appeared that the gap served to relieve the pressure difference between the upper- and lower-surface trailing-edge regions but its presence did not affect the extent of flow separation behind the spoiler. Evidently, the quantity of flow through the gap was too small to affect the extent of flow separation or the flow was not directed through the gap properly to decrease the extent of flow separation.

At all angles of attack and Mach numbers the gap was effective in increasing the upper-surface pressures behind the spoiler. The gap was also effective in increasing upper-surface pressures ahead of the spoiler. but this effect was very small at low angles of attack. Both of these effects increased the rolling effectiveness. On the lower surface, effects of the gap were limited to localized pressure changes.

Lower-surface spoiler with a gap. - The effect of a lower-surface spoiler ahead of a wing gap on the wing pressures is shown in figure 14. At the lower angles of attack, the lower-surface spoiler was equivalent in its effect on the flow to the upper-surface spoiler, the basic configuration being completely symmetrical about the chord plane. The appropriate part of the discussion on upper-surface spoilers therefore applies. As the angle of attack was increased, however, the lower-surface spoiler became ineffective and above about 10° produced losses in lift rather than increases. The changes leading to these losses in lift were as follows:

- (1) The region of increased pressure ahead of the spoiler on the lower surface became less extensive.
- (2) Behind the spoiler, the lower-surface pressures became much less than corresponding basic wing pressures and hence there was a large reduction in normal force over the trailing-edge region.
- (3) When the angle of attack was reached where separation existed on the upper surface, the influence of the lower-surface spoiler in reducing upper-surface pressures vanished.

Thus, the two favorable influences noted for upper-surface spoilers both became less for the lower-surface spoiler as the angle of attack was increased, whereas the unfavorable influence became greater. This increase with angle of attack of the unfavorable trailing-edge loading did not occur with the spoiler on the upper surface.

Oppositely Deflected Spoilers

A configuration was tested in which the left wing had an uppersurface spoiler and the right wing had a lower-surface spoiler. There was no gap through the wing behind either spoiler.

Comparisons showing the effect of the lower-surface spoiler on spanload distributions for the wing having the upper-surface spoiler are presented in figure 15 for three representative Mach numbers. Study of this figure and unpublished comparisons at other Mach numbers showed that the presence of the right-wing lower-surface spoiler reduced the effectiveness of the left-wing upper-surface spoiler at an angle of attack of 0° for Mach numbers above 0.90. For all other conditions the reverse was true, although the differences appeared to be small in every case.

At about 11° there were inconsistencies in the span loadings throughout the speed range, possibly because at this angle of attack the extent of the separated flow on the wing was not always consistent. Small differences in surface conditions on the wing could probably have varied the extent of separation at this angle of attack enough to account for the differences shown.

In order to determine the magnitude of the rolling moment represented by the differences shown in the span-load plots of figure 15, the differences were plotted to a larger scale and integrated for moment about the plane of symmetry. The rolling-moment-coefficient increments so obtained in general were less than ±0.0016, with no consistent Mach number effects discernible. Thus, the carry-over effects between wings were quite small, probably because of the inability of the spoiler to affect the inboard wing section loads to any great extent. This indication should lend credence to the use of reflection-plane test techniques for obtaining data on this type of control on swept wings.

Chordwise pressure differences have been compared at one speed only (fig. 16) to show the effect of adding the lower-surface spoiler to the opposite wing. The generally small differences were typical of all speeds.

Spoiler Loadings

Pressures at each spanwise station on the spoilers were integrated in such a manner as to obtain the pressure force coefficient normal to the spoiler chord line as indicated in the list of symbols. Figure 17 shows distributions across the wing span of these spoiler section normal-force coefficients for the three spoiler configurations weighted for spoiler height.

For upper-surface spoilers, the highest loads occurred at zero angle of attack. As the angle of attack was increased, the spoiler loads, especially near the tip, fell rather rapidly. At about an angle of attack of 11°, when flow separation existed over a considerable portion of the outboard wing area (see figs. 10 and 12), the spoilers were unloaded outboard of 0.40 semispan for all Mach numbers. This loss of spoiler load contrasted with the wing span-load distributions (fig. 8) which show that lift decrements caused by the spoilers extended much farther outboard at this angle of attack. Apparently, the inboard portions of the spoiler had an influence on the separated flow over the outboard wing areas. At higher angles of attack, where flow separation extended over more of the wing, this influence diminished.

Addition of a gap through the wing behind the spoiler resulted in a reduction of the loads on the spoiler for nearly all conditions. Figure 18, which makes a few comparisons of pressures on the upper-surface spoiler with and without gap, shows that this reduction resulted from increases in the low pressure on the back of the spoiler. These increases were probably due to the flow of high pressure air from the lower surface through the gap. The largest relief, as might be expected, was inboard where the pressures behind the spoiler were lowest.

Figure 18 shows that, at 0.25 semispan for angles of attack up to 16° and at other stations for low angles, the front face pressures on the spoiler were somewhat lower in the center of the spoiler than at the top and bottom. This type of pressure distribution is similar to that shown in reference 12 and probably indicates that there was a circulation of the separated air ahead of the spoiler as illustrated in figure 11, since such a circulation would result in higher velocities (and lower pressures) near the center than at the top or bottom.

For a lower-surface spoiler, figure 17 indicates that the loads continued to increase to the highest angles of attack of these tests. Figure 19, which compares the pressures on upper- and lower-surface spoilers, shows that the load on the lower-surface spoiler increased with angle of attack because the pressure coefficient on the front face of the spoiler remained at an approximately constant positive value through the angle-of-attack range, whereas the rear face pressure coefficient became more negative with increasing angle of attack.

Inasmuch as the spoiler pressure distributions were quite rectangular for most conditions, it would seem that a satisfactory measure of the spoiler loads could be obtained from a pair of orifices at each station located in the wing at the base of the spoiler, one orifice being immediately ahead of the spoiler and the other immediately behind. In order to test this supposition, the loads on the spoiler have been determined by this method at two speeds, Mach number 0.60 and 0.98, by using the closest available wing orifices, which were at 0.65- and 0.75-chord locations. The pressure at the 0.65-chord location was assumed to be the same as that on the front face of the spoiler and the pressure at 0.75-chord location was assumed to be the same as that on the rear face. The results are shown by the symbols in figures 17(a) and 17(f). Agreement was good for most conditions but would probably have been better if orifices closer to the spoiler had been available. This is especially true at the inboard stations where the extent of the separated flow ahead and behind the spoiler was small. It is therefore believed that a pair of properly located orifices on the wing at each station would give loads on this type spoiler as good as those obtained in the present tests with seven orifices on the spoiler at each station.

CONCLUSIONS

An investigation was conducted with 73-percent-semispan inboard-spoiler ailerons having heights of 4 percent of the local chord and located on the 70-percent-chord line of a 45° sweptback-wing—fuselage combination. Pressure data were measured on the wing and spoiler at several spanwise stations at Mach numbers from 0.60 (Reynolds number 5.1×10^6) to 1.03 (Reynolds number 6.2×10^6) for angles of attack that usually extended to 20° or more. The results of the investigation indicate the following conclusions:

- (1) Operation of upper-surface spoilers at low and moderate angles of attack produces normal-force decrements which are largest in the 0.6 to 0.8 semispan area of the wing. Most of the normal-force decrement is associated with increases in the upper-surface pressures ahead of the spoiler due to a deceleration of the air approaching the spoiler. An additional contributing factor is a decrease in lower-surface pressures resulting from transmission of upper-surface pressure changes around the trailing edge.
- (2) Rolling-moment effectiveness is reduced for upper-surface spoilers at high angles of attack because they do not have much effect on the separated flow which occurs on the basic swept wing at these angles of attack.
- (3) A gap through the wing behind the spoiler is effective in increasing the rolling-moment effectiveness because it permits a relief of the pressure difference between the upper- and lower-wing surfaces.

- (4) Lower-surface spoilers give reversed rolling-moment effectiveness at angles of attack higher than about 10° primarily because there is a large decrease in pressure behind the spoiler on the lower-wing surface at the higher angles of attack.
- (5) A lower-surface spoiler has only a small effect on the rollingmoment effectiveness of a spoiler located on the upper surface of the opposite wing.
- (6) Spoiler section loadings are highest at the inboard end. For upper-surface spoilers, the spoiler loading decreases rapidly with increasing angle of attack, especially outboard. For a lower-surface spoiler, however, the spoiler loading increases with angle of attack.
- (7) Spoiler section load at any point along a spoiler of this type can be determined by measurement at the wing surface of the pressure drop across the spoiler.

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National Advisory Committee for Aeronautics,
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TABLE I

BASIC WING

| ſ | | PRESSURE COEFFICIENT, P, AT: | | | | | | |
|---------------|---|--|--|---|--|--|---|---|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95ь/2 |
| 1 | M = 0. | .60 α = 0.0° | | | | <u> </u> | | |
| UPPER SURFACE | 12.5000 12.5000 12.5000 10.500 | 0 19 .1 37 .0 028 .0 028 .0 0389 .0 0811 .1130 .1149 .1157 .1157 .1157 .1157 .1149 .1199 | . 485 1207 097 1095 1124 1134 1154 1159 1704 1664 146 146 183 183 183 183 183 183 184 185 | - 44 2 - 114 0 - 119 4 - 119 2 - 119 2 - 116 6 - 116 6 | . 684 - 093 - 098 - 0991 - 107 - 125 - 134 - 139 - 147 - 150 - 154 - 155 - 150 - 149 - 099 - 074 - 044 - 029 - 009 | . 449 043 1009 110 1126 133 138 135 138 136 136 136 136 137 128 128 1099 0618 0022 048 | . 467 091 138 071 1090 101 125 125 134 137 137 125 126 137 125 126 137 1204 104 104 104 105 10 | - 44 3 - 00 8 8 16 - 10 11 13 20 42 20 11 12 34 20 11 12 34 20 11 11 14 4 10 10 10 10 10 10 10 10 10 10 10 10 10 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 550.000 65.000 850.000 850.000 | .134 .069 .0011 0018 0053 0065 1124 124 124 141 1639 1588 1581 1581 1036 1 | 092 069 071 087 1321 1523 162 178 178 178 155 136 178 100 136 100 | 1258 098 1065 11284 115483 16337 1369 1369 13637 1369 1369 | 134 098 047 080 106 139 144 145 147 147 148 148 148 129 115 069 008 008 008 008 008 | 124 091 114 116 122 132 142 143 147 137 128 110 086 048 0031 004 | 0741071141031118137139139136139130100000000000 | 14091376136013761406118811551108099900750003090031900317 |
| | M = 0. | $.60 \qquad \alpha = 4.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 25.00 25.00 40.00 45.00 55.00 60.00 70.00 85.00 95.00 | . 013 - 1743 - 2757 - 264 - 2661 - 2661 - 2670 - 2560 - 2670 - 2560 - 2739 - 2451 - 1988 - 1988 - 1988 - 1788 - 1988 - 19 | 241 - 1.0925884654003733413223107300297291283252224171131056012 | 471 - 1.3128718454684293955332632102855224442178094500455 | 349 - 1.2387665614794483953317302265246241172114077007 | 905 - 1.471 - 844 541 446 370 343 315 296 281 244 224 225 177 149 057 013 057 013 039 | 969 - 1.344901555247243702682432201186135105051027040067 | 673 - 1.3766792440437542512101110111011015007702770273 |
| LOWER SURFACE | 1.25 2.50 5.50 10.00 15.000 25.000 35.000 35.000 45.000 45.000 65.000 65.000 75.000 85.000 90.000 | 308 2217 1184 1120 1075 1027 10027 10027 10027 100665 100665 100655 100655 | .357 .280 .201 .149 .069 .004 024 053 075 083 077 075 075 075 075 075 075 075 | .3777.2904.1637.0841.01133.00455.0769.00558 | 362 293 238 194 152 087 002 002 - 017 - 031 - 052 - 052 - 052 - 007 007 0016 | .413 .331 .240 .145 .103 .0633 .0033 .009 015 045 032 005 .007 .005 .0133 .057 | .353 .244 .187 .1652 .059 .027 003 1 .026 055 056 056 056 056 056 056 056 056 056 056 056 056 002 002 002 002 003 003 005 00 | 0456 0863 0863 0756 0756 0756 0756 0759 0876 0759 0759 0759 0759 0759 |

TABLE I

BASIC WING

| Ţ | | | <u>-</u> | | PRESSU | RE COEFFICIENT, | P, AT: | | |
|---------------|---|--------|---|---|--|--|--|---|---|
| | PERCENT CHORD | 0.1 | 35b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ъ/2 | 0.85ъ/2 | 0.95b/2 |
| ı | М = | 0.60 | $\alpha = 6.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.500 10.000 15.000 15.000 25.000 35.000 45.000 45.000 60.000 70.000 85.000 90.000 | | 019 7448 74151 | - 12924 - 1296132 - 265825 - 2 | - 1:082 - 1:2347 - 1:2647 - 1:9088 - :7566 - :6860 - :4900 - :4401 - :3348 - :3338 - :2799 - :250 - :212 | - 1:511 - 1:238 - :967 - :767 - :663 - :466 - :426 - :338 - :349 - :349 - :293 - :293 - :208 - :199 - :199 - :069 | - 1.570 - 1.160 - 1.69336757439955534399335102413241318010620062 | 1.099633 | 104527723983114682779816554 444542667632198682779816554 1111111111111111111111111111111111 |
| LOWER SURFACE | 12.500 57.500 10.500 10.500 23.000 23.000 45.000 45.000 45.000 65.000 75.000 85.000 85.000 | | 6660125545454143667545174554517456767545174567675451747676754551747676767676767676767676767676767676767 | 4591 4391 4391 4311 4311 4311 4011 | .4546 .3946 .31596 .21599 .2169 .1177 .0060 .0017 00137 00134 | *447 *3396 *3377 *278 *233 *171 *125 *0070 *0010 *0011 *0011 *0011 *0012 | .121 .091 .062 .032 .018 002 012 022 013 .000 .007 | 2319 2235 103 1071 0037 0014 - 0014 | ###################################### |
| | М | = 0.60 | $\alpha = 8.0^{\circ}$ | 0 | | | | | |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 10.00 20.00 20.00 25.00 40.00 40.00 50.00 70.00 60.00 80.00 90.00 | | 60034 60534 5948 5036 44594 4508 44105 441 | - 1.213 - 1.1399 - 1.0305 829 7135 5610 4412 4412 4412 3603 297 225 132 035 | - 1.282 - 1.221 - 1.137 - 1.965 836 674 573 573 457 441 349 349 281 281 081 082 | - 1.042 - 1.0066 9056 9057 779 657 653 490 438 389 268 252 142 0955 040 | 953 944 906 859 647 5591 524 449 323 231 188 118 094 064 0034 0034 | 936 907 862 833 798 730 668 601 551 485 485 367 315 267 2267 120 | - 1.483 - 1.172 - 1.9825 8635 5262 3337 2275 2443 2281 2208 1912 11688 11688 11688 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 20.00 30.00 35.00 45.00 65.00 65.00 75.00 80.00 95.00 | 1 11 | 368 353 369 365 3337 288 220 1129 1111 0511 0017 0017 0017 0010 0011 0010 | . 484 . 484 . 382 . 327 . 285 . 146 . 1082 . 061 . 007 . 007 . 007 . 007 . 007 . 007 . 007 | . 449 .381 .387 .289 .234 .183 .117 .060 .029 .021 .005 .017 .006 | .439 .389 .336 .304 .232 .185 .121 .093 .071 .049 .030 .020 .014 .014 | .449 .388 .334 .301 .238 .190 .0151 .0168 .0077 .043 .029 .0177 .008 .013 | .461 .380 .322 .302 .216 .165 .124 .089 .053 .026 .004 013 .026 033 026 036 | . 425 .393 .308 .222 .166 .083 .017 .0030 035 055 056 058 061 062 044 039 039 |

TABLE I

BASIC WING

| | PERCENT | | PRESSURE COEFFICIENT, P, AT: | | | | | | |
|---------------|---|---|---|---|---|--|---|--|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 | |
| | M = | 0.60 α = 11.2 | 0 | | <u> </u> | | <u> </u> | I | |
| UPPER SURFACE | 12.5000 12.5000 10.5000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.00000 10.00 | - 1.0573 - 1.05239 - 1.09488 72522 66329 66329 66329 66329 4199 41992 33251 2252 2252 2252 2252 2252 2252 | - 2.007 - 1.304 - 1.3538 - 1.3538 - 1.3845 - 1.3845 - 1.0846 597 3767 | - 1.729 - 1.2717 - 1.2617 - 1.2631 - 1.1236 - 1.1959 - 1.11489788787869910552023571096 | - 1 . 385 - 1 . 2239 - 1 . 2246 - 1 . 2811 - 1 . 2011 - 1 . 0246 | 820 794 799 799 780 757 7195 689 661 661 661 548 548 548 548 548 538 5338 376 463 3338 | 536 5325 55104 55104 4865 44012 44012 4322 44012 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 32694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 33694 | 552459146354399297122570221502150187118711899 | |
| LOWER SURFACE | 125.000000000000000000000000000000000000 | 38194044814363448145363448145363448145363448194648481999773999739885 | .491 .592 .445 .445 .340 .2980 .2173 .1546 .1011 .0074 .068 | .438 .4959 .4182 .3825 .2720 .1968 .1373 .0871 .0713 .0753 | .500 .4972 .4932 .4332 .3317 .2258 .1940 .1132 .1080 .0046 .0046 .0055 | .469 .499 .451 .395 .302 .244 .205 .1731 .1051 .0051 .008 .005 0657 0687 136 | .477 .414 .345 .2672 .1072 .0174 .043 .0120 .0555 -0757 -0928 -1155 | .416 .3919 .324959 .0578 .10578 001557 001557 00766 00776 00778 009874 1024 | |
| | M = 0.6 | 60 α = 15.5° | | | | | - · · · · · · · · · · · · · · · · · · · | | |
| UPPER SURFACE | 1.25 2.500 70.000 15.000 25.000 35.000 40.000 40.000 55.000 60.000 70.000 85.000 85.000 95.000 | 7007 1 1 2 5 9 1 9 1 1 2 8 8 8 8 9 4 1 2 8 8 8 9 4 1 2 9 9 7 0 7 2 8 7 6 6 2 9 6 6 6 0 6 6 6 0 6 6 6 0 6 0 6 | - 27 324 - 129817 - 129817 - 129817 - 12993 - 2221998 - 1281998 - 1281998 - 1281998 - 1281998 - 1281998 - 1281998 - 1281998 - 234499 - 23449 - 2 | - 1 : 23 3 4 - 1 : 23 3 4 - 1 : 23 3 3 - 1 : 23 3 3 - 1 : 23 3 4 - 1 : 23 3 3 - 1 : 23 3 5 - 1 : 23 5 - 1 : 23 5 - 2 : 23 5 - 3 | 1788574 - 188574 - 188574 - 188597 - 188597 - 188699 - 188999 - 187787 - 1877777 - 1877777 - 187777 - 187777 - 187777 - 1877777 - 187777 - 187777 - 18777 - 18 | 6666 6479 6428 6328 6218 5984 5575 5557 5557 5547 5460 518 518 433 | - 44543 - 44543 - 44403 - 44403 - 44014 - 74401 - 74014 - 73945 - 73773 - 737680 - 73773 - 737 | 15341088948509991743343777 553535353535322228282889887777 | |
| LOWER SURFACE | 1.25 2.500 7.500 15.000 25.000 35.000 40.000 50.000 70.000 80.000 80.000 95.000 | \$2575375810828755558108888888888888888888888888888888 | *424 *5557 *5575 *5575 *5543 *352 *352 *2277 *2275 *2275 *21752 *11752 *11312 *0075 *0001 | #3512844128995512844273175144427522440721141553 1144420049 | # 45314 # 45314 # 45314 # 45314 # 45314 # 45318 # 15318 # 1 | .433 .500 .489 .452 .415 .354 .303 .254 .167 .132 .091 .020 029 029 106 136 143 | 7487 74452 74462 73869 7246 71936 70958 70958 70958 70977 71309 71110 711309 71171 71197 | # 44590253437407594457466666666666666666666666666666666 | |

BASIC WING

| Ţ | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|---|---|---|--|--|---|
| - | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85ь/2 | 0.95b/2 |
| \dashv | M = (| $0.60 \alpha = 19.5^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.500 7.500 15.000 25.000 25.000 35.000 45.000 50.000 70.000 80.000 90.000 | 021930 - 1.363 - 1.363 - 1.363 - 1.3745 - 1.234 - 1.1940 - 1.0812 - 1.082 - 1.08285067829661955881 | 1.246 1.2246 1.2231 1.2232 1.2239 1.2219 1.1209 1.1167 1.1144 1.10963 | - 1.070 - 1.0376 - 1.0286 - 1.0286 - 1.9286 9975 9957 9957 9930 9957 9930 9956 9861 8861 8861 8861 8861 8861 | - 1.060 801 793 793 783 767 740 721 705 705 705 706 70 | | | 453 428 421 421 421 421 422 422 423 433 - 433 - 43 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 25.00 35.00 45.00 45.00 65.00 70.00 85.00 85.00 95.00 | . 155 .242 .5123 .6633 .5653 .5653 .5471 .4187 .3445 .2642 .2309 .1718 .1116 .0024 | .408 .571 .609 .566 .5508 .466 .4729 .2256 .184 .127 .057 .0021 080 | .331 .506 .548 .5315 .5117 .411 .3229 .238 .2038 .1258 .1253 .0952 0108 1080 1080 | .430 .381 .332 .287 .243 .204 .116 .080 .048 .012 | .387 .495 .507 .485 .3477 .3497 .2492 .164 .1183 .0440 -012 | . 486 . 470 . 441 . 426 . 353 . 2941 . 186 . 139 . 094 . 0015 0248 075 105 160 185 22 | .358 .408 .374 .322 .273 .200 .116 .0217 023 017 093 119 128 139 156 164 181 181 |
| | | $0.60 \alpha = 25.8^{\circ}$ | | | 0.2.* | 740 | . 604 | 524 |
| UPPER SURFACE | .00 1.25 5.00 7.50 10.00 20.00 20.00 30.00 40.00 45.00 55.00 665.00 75.00 850.00 | 914 - 1.073 - 1.034 - 1.022 - 1.023 - 1.022 - 1.023 - 1.024 - 1.008 9984 9961 9961 9954 9918 9918 8918 | - 1.015 - 1.023 - 1.023 - 1.023 - 1.0335 - 1.0335 - 1.0335 - 1.0327 - 1.0327 - 1.0327 - 1.0327 | 982 987 987 995 995 995 996 963 9427 9427 8633 8538 | 879 8880 8774 8663 8473 8433 8259 81123 777 777 7724 7124 70123 | 735 729 725 723 710 705 690 680 661 6546 65646 65646 6589 5899 589 | - 5990 - 55911 - 55917 - 55821 - 557416 - 557436 - 557436 - 55434 - 55434 - 55434 - 5474 - 54 | 5195115125095095125135145085084094764450 |
| LOWER SURFACE | 55.00 | 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 | .658 .6680 .6199 .534 .5374 .4954 .4162 .3726 .2254 .213 .2264 .213 | . 55 9 . 46 9 . 42 3 . 3 3 7 . 2 9 7 . 2 5 1 | . 468 . 538 . 554 . 547 . 540 . 424 . 333 . 2245 . 2245 . 2166 . 1166 . 00161 . 00161 . 00161 . 00161 . 00161 | .445 .523 .5317 .480 .3434 .3434 .2258 .1599 .1785 .043 .033 0812 0812 | . 451 .500 .500 .491 .385 .2832 .2832 .187 .0187 .014 020 | .239 .377 .475 .375 .274 .196 .1196 .081 .0016 |

TABLE I

BASIC WING

| Ī | | | PRESSURE COEFFICIENT, P, AT: | | | | | | | |
|---------------|---|--|---|--|---|--|--|--|--|--|
| ١ | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 | | |
| \exists | M = 0 | .85 $\alpha = 0.0^{\circ}$ | | | | | | - | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 25.00 30.00 35.00 45.00 55.00 60.00 60.00 60.00 85.00 95.00 | 116 136 187 188 188 206 159 149 185 1745 | .516 097 080 078 093 107 138 157 170 180 212 2217 2217 196 1635 088 055 008 | . 458 127 1150 138 1728 1728 2010 2215 2216 184 185 184 185 185 1075 0219 0227 | .712144138146141179190191193193195127091126127 | .467 -048 -113 -0820 -1355 -1567 -1770 -1776 -1776 -1775 -1643 -118 -072 -143 -118 | 487 | . 454 098 098 1145 1762 1763 154 154 139 130 100 100 056 013 0013 0084 | | |
| LOWER SURFACE | 1.25 2.50 7.50 10.50 10.50 20.00 30.00 40.00 45.00 60.00 60.00 70.00 80.00 90.00 | 005 038 085 126 134 146 204 209 219 195 185 153 | 102 073 073 093 104 134 159 176 203 223 223 213 - | 16 0 1179 1326 1651 18954 2212 22215 22215 16858 1185 1185 1185 1185 1085 | 167138712411813191319951995199619971619161908705530201 .046 | 183147161163173185189189179185186179100179100179100179100 - | 117140150157156177182187184187166127169039035066 | - 219 - 1170 - 1204 - 2216 - 1169 - 1169 - 1151 - 1132 - 1102 - 0752 - 0058 - 00149 - 077 | | |
| | M = 0 | $0.85 \alpha = 4.0^{\circ}$ | | <u></u> | | | | | | |
| UPPER SURFACE | . 00 1.25 2.50 5.00 7.50 10.00 20.00 30.00 40.00 40.00 55.00 665.00 665.00 805.00 90.00 | | .098998863424380338338338334533613613693399223 | 057 - 1.176 - 1.0247 484 435 416 406 386 387 3559 2262 220 1401 046 018 048 | . 175 - 1.243 - 1.166743512455455433404377349330281257186186114073036002019 | 012 | 410 - 1.205 - 1.106 843 695 593 472 408 3627 285 255 224 200 174 141 052 019 .042 .066 | 281 - 1.170 - 1.1882688935662299219918631355103610690222 .0010 .020 | | |
| LOWER SURFACE | 30.00 35.00 40.00 45.00 50.00 | .3099 .2232 .1949 .0046 .0046 .0053 0053 0064 0080 0080 0080 | 015 036 051 072 092 | .149 .114 .069 .027 030 048 065 077 087 | .342 .276 .245 .187 .130 .070 .040 .009 - 012 - 031 - 061 - 069 - 062 - 053 - 053 - 034 - 013 .008 | .387 .307 .219 .164 .123 .088 .042 .014 069 031 064 064 064 067 055 028 019 036 | .328 .230 .173 .160 .084 .0039 .0030 060 077 096 104 093 076 076 | .360 .279 .094 .043 043 103 108 107 105 108 070 050 050 038 070 038 024 010 | | |

BASIC WING

| I | | | | PRESSU | RE COEFFICIENT, | P, AT: | | ··· · |
|---------------|---|---|--|---|---|---|--|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0 | $.85 \qquad \alpha = 6.0^{\circ}$ | Ţ-· | | | | | |
| UPPER SURFACE | 12.57.00000000000000000000000000000000000 | \$6834659630138895538088276 \$2574664466567171445553083 \$1000000000000000000000000000000000000 | - 1.1998 - 1.1998 6283 6283 4486 4456 4456 4479 44510 4479 44510 4479 4319 219319 219319 219319 219319 | - 1.313768 - 1.3135768 - 1.55200935 55200930 55200930 5520990 5520990 5520990 5520990 5520990 5520990 | 33193448245195775122 59136 0313074652692505775122 111.222444351222 1196220 111.222444351222 1196220 | - 1.463.844.5667.60.60.60.60.60.60.60.60.60.60.60.60.60. | 2638363343663 1.209027332663 1.1090273326 1.1090273326 1.1090273325 1.10902733334363 1.10902733334363 1.10902733334363 1.1090273 | 107409001 107409001 1099001 1099001 1099001 1099001 1099001 1099001 1099001 1099001 1099001 1099001 1099001 1099001 |
| LOWER SURFACE | 1.2500 7.500 10.500 20.000 20.000 20.000 40.000 50.000 60.000 75.000 85.000 95.000 | *381588 | # 4 1 3 7 4 7 3 3 9 2 3 3 3 4 7 7 3 3 9 2 3 3 3 3 4 7 7 3 3 7 3 2 3 3 3 3 3 3 3 3 3 3 3 3 3 | 437 3684 2884 2835 11445 1067 2013 - 0028 - 0041 - 0040 - 0033 - 0033 - 0034 - 0044 | 155858754122278778779 457821111000200128779 111100000000000000000000000000000000 | . 453 .3907 .2509 .163 .114 .080 .051 .027 .0015 .027 .0037 .036 .036 .031 .0011 .0029 | 4117 42169 711638 10681 10036 10036 100741 1007 | 20464415522174455221100462247288411000997648410010101100009976484100100110001100011000110001100011000 |
| | M = 0 | .85 $\alpha = 8.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.5.00 7.5.00 10.00 20.00 25.00 30.00 35.00 40.00 50.00 50.00 60.00 70.00 85.00 90.00 | 429 423 467 527 560 452 410 238 | - 1.215 - 1.1980 - 1.1397 - 1.097565834206544857175511826891264 | 126 074 047 | 127 091 050 | 960 - 1.277 - 1.262 - 1.288 - 1.1972 - 1.0913967830753673585498402701158113024 | 83474974371366166285859523744935631927725011991 | 957 - 1.1052 - 1.0675 - 1.0975 - 1.11196423162336224332243322072001186617001652 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 25.00 35.00 45.00 45.00 66.00 66.00 67.00 80.00 90.00 95.00 | .403 .427 .419 .403 .317 .245 .183 .128 .062 .062 .0014 | .520 .4623 .3867 .2226 .1856 .1146 .1144 .0614 .01333 .0008 .0008 | . 484 . 431 . 357 . 306 . 276 . 164 . 125 . 097 . 068 . 046 . 030 . 001 . 000 | .460 .411 .368 .318 .268 .216 | .478 .440 .369 .314 .268 .221 .169 .099 .073 .047 .025 | . 448 . 365 . 306 . 287 . 146 . 100 . 064 . 033 . 062 . 079 . 075 . 079 . 075 . 069 . 069 . 087 | . 427 .378 .289 .2153 .0118 0518 103 1104 1097 1114 0997 0084 0097 0064 |

TABLE I

BASIC WING

| Γ | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | |
|---------------|--|--|---|--|--|--|---|--|--|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 | | |
| T | M = 0.8 | $5 \alpha = 11.3^{\circ}$ | | | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 25.00 35.00 40.00 45.00 55.00 65.00 75.00 85.00 85.00 85.00 | 007 - 4517 - 7649 - 7649 - 77370 - 6427 - 66427 - 66427 - 6528 - 6579 - 55884 - 65799 - 4979 - 3770 | 887 - 1.479 - 1.499 - 1.499 - 1.3330 - 1.268 546 5566 5566 520 6177 490 2403 2403 1111 | - 1.015 - 1.384 - 1.247 - 1.246 - 1.203 - 1.193 - 1.111 - 1.063 - 1.017 9612 850 762 637 581 3881 3810 243 190 | - 1.1468 - 1.1458 - 1.1364 - 1.1393 - 1.0507 - 1.095079621829978577371642 | 873 7873 7645 7645 7747 7311 6675 6630 6630 6630 5982 59866 59866 59866 | 514 483 465 465 463 454 441 425 405 391 3877 3767 3767 3567 | | | |
| LOWER SURFACE | 95.00 12.50 5.00 10.00 10.00 10.00 25.00 25.00 45.00 45.00 65.00 65.00 85.00 95.00 | 1 8 6 . 3 8 3 3 . 4 5 1 9 1 | . 576 . 556 . 501 . 448 . 411 . 343 . 298 . 2256 . 220 . 1186 . 0276 . 039 . 065 . 039 . 0118 . 008 | .511 .507 .456 .413 .376 .269 .269 .159 .130 .104 .079 .066 .039 .023 .023 | .515 .490 .464 .418 .3708 .265 .190 .125 .097 .047 .047 .047 .047 .047 .047 .047 .04 | .489 .491 .441 .389 .351 .297 .2496 .161 .122 .090 .027 .021 -021 -021 -0521 -122 .090 .021 -122 .090 .021 | . 4822 . 4822 . 3629 . 2600 . 1947 . 0053 . 0120 . 00688 . 1211 . 1628 . 1212 . 1887 | .437 .4131 .2552 .2103 .00159 0059 1124 1144 1146 1569 11577 1184 1184 | | |
| | M = 0.85 | $\alpha = 15.8^{\circ}$ | | | | | | - 3419 | | |
| UPPER SURFACE | 1.00 2.50 5.50 5.50 105.000 125.0000 125.000 125.000 125.000 125.000 125.000 125.000 125.0 | - \$601 - \$545 - \$449 - \$360 | - 1 - 294 - 1 - 529 - 1 - 529 - 1 - 5482 - 1 - 5482 - 1 - 5484 - 1 - 1 - 269 - 1 - 1 - 271 - 1 - 1 - 1031 - 1 - 1031 | - 1:026 - 1:047 - 1:043 - 1 | 9591 | 560 5544 546 512 503 488 475 462 | - 2382 | | | |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 40.000 60.000 70.000 75.000 85.000 85.000 | \$ 415 \$ 445 \$ 4461 \$ 3338 \$ 2245 \$ 2146 \$ 1163 \$ 0099 \$ 0022 | 55549 555593 543618 543618 522454 5417462 7017462 70040 | .5496 .5496 .4463 .3314 .2370 .21986 .119976 .003976 .00459 | *5509 *474 *5509 *474 *474 *53 | .502 .476 .436 .3347 .249 .196 .1196 .029 .0007 .0038 .0007 .0038 | 4404124905354490535449053544905354108354211083542111111111111111111111111111111111111 | 1583468003333 458210006246800333 5582100555555555555555555555555555555555 | | |

BASIC WING

| | | | | PRESSUF | RE COEFFICIENT, | P, AT: | | |
|---------------|--|--|--|--|--|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40ь/2 | 0.55b/2 | 0.70b/2 | 0.85ъ/2 | 0.95b/2 |
| | M = 0. | 85 $\alpha = 19.6^{\circ}$ | | | | | | |
| UPPER SURFACE | 125000000000000000000000000000000000000 | 0 6 4 8 8 8 1 8 1 8 1 8 1 8 1 8 1 8 1 8 1 8 | - 1.020 - 1.0108 - 1.098 - 1.997 984 9654 9456 9466 9466 8576 8586 8586 8646 7640 7640 | 8701 86701 86446 88446 8845 88273 88273 88273 777783 77783 772039 | 841 737 734 742 742 720 715 709 715 696 686 686 686 686 686 683 646 631 | | 5558 5558 5544669 554461 553280 | 49767 48747 466129 466129 46699 46690 46690 46630 - |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 25.00 35.00 45.00 45.00 65.00 65.00 65.00 85.00 90.00 | 217 402 6674 7754 66210 6820 4637 3520 4637 3116 2252 2154 11115 0052 | .573 .661 .6640 .612 .505 .415 .3738 .299 .260 .2196 .165 | . 4573 . 5839 . 55392 . 443932 . 443932 . 35081 . 11286 . 0078 - 00855 - 12530 | .509 .5442 .5522 .5044 .3949 .3076 .22199 .11376 .0064 -04824 -1284 | . 421 . 506 . 512 . 486 . 459 . 3536 . 260 . 2173 . 1287 . 0451 - 025 | . 492 . 475 . 446 . 3961 . 305 . 2248 . 1938 . 1938 . 1938 . 1090 . 0314 1600 141 1284 2267 2319 | . 4214 . 3337 . 202008 . 10514 . 0017452 . 0017452 . 1190412 . 120412 . 224422 . 22742 . 228856 |
| | M = 0 | | | <u> </u> | <u> </u> | <u> </u> | ! | L |
| UPPER SURFACE | 1.25 2.50 7.50 10.00 20.00 25.00 30.00 45.00 55.00 65.00 65.00 65.00 80.00 85.00 95.00 | 898 811 804 783 736 | 881 885 884 884 887 887 887 887 887 887 885 885 885 885 885 | 840 830 826 823 | 776 767 758 753 | 780 791 798 799 788 799 769 769 765 753 753 753 753 753 753 753 | 69769470973073073168266266566566466466466467655644594594 | 658967 58967 58967 557678 557678 557678 557678 557678 557678 557678 557678 557688 |
| LOWER SURFACE | 1.25 2.500 7.000 10.000 20.000 25.000 35.000 45.000 65.000 70.000 75.000 85.000 85.000 95.000 | 717 668 619 583 5541 5500 4423 3322 2259 210 | .404 .369 .335 .298 .222 .185 .1037 | .358 .567 .640 .632 .600 .557 .518 .441 .398 .316 .247 .240 .192 .096 .001 -081 -081 | . 420 .5776 .5771 .5765 .5405 .4622 .3834 .2941 .208 .125 .0041 | . 295 . 475 . 541 . 5528 . 499 . 454 . 336 . 285 . 238 . 1193 . 107 . 070 - 064 - 1145 - 274 | .460 .498 .498 .471 .444 .396 .347 .292 .138 .083 .083 .015 -015 -143 -1289 -341 | . 256 . 398 . 398 . 398 . 301 . 201 . 301 |

TABLE I

BASIC WING

| Γ | | | | PRESSUE | E COEFFICIENT, | P, AT: | | |
|---------------|--|--|--|---|--|--|---|--|
| ļ | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | $\alpha = 0.0^{\circ}$ | | | - | | | |
| UPPER SURFACE | 1 25 25 25 25 25 25 25 25 25 25 25 25 25 | .030 .216 .072 .0145 .032 .0014 .027 .050 .104 .108 .118 .1190 .158 .1204 .229 .177 .2204 .229 .2177 | .530 078 064 0669 0794 1312 1722 1912 2247 | . 459 1127 1032 11310 11791 11791 1291 22427 22427 22427 22427 22427 22427 10666 0004 | 720 - 1114 - 121 - 121 - 128 - 178 - 178 - 193 - 197 - 192 - 1185 - 1718 - 1122 - 084 - 0853 - 0176 - 034 | . 464 052 087 148 148 1681 189 189 184 1754 130 0847 0027 0054 | . 487 079 161 088 113 163 166 176 189 198 197 178 126 178 126 178 126 | 45999739945533462729934002578 |
| LOWER SURFACE | 1.25 2.50 7.50 10.000 20.000 30.000 40.000 45.000 65.000 75.000 85.000 85.000 95.000 | .194 .131 .061 .039 .007 .007 0130 1130 1130 1130 1215 224 224 224 224 2215 2215 2215 | 093 063 084 094 137 138 162 207 245 245 245 234 237 246 237 246 237 238 - | 15310811212416216320021724125219610790395041 | 172 141 082 124 191 205 209 209 209 197 156 151 082 015 .058 | 2081677188717891999199419941586196804440152 | 137 154 161 178 195 2015 215 215 2157 1277 1276 036 034 046 075 | 2 4 0 2 7 8 2 7 4 6 2 2 4 6 2 2 4 6 1 8 3 7 1 5 3 1 5 3 1 5 3 1 5 3 1 6 7 1 6 7 0 7 6 0 7 7 6 0 7 7 6 0 7 7 6 0 7 7 7 7 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 |
| | M = 0 | $.90 \cdot \alpha = 4.0^{\circ}$ | | L | | | 1 | T |
| UPPER SURFACE | 1.25 2.50 7.50 10.00 20.00 30.00 35.00 45.00 45.00 50.00 65.00 70.00 80.00 80.00 80.00 | 270 298 394 300 392 328 340 3405 3405 3405 3405 3405 | .1659218203953613283253413563794164304214024133019089 | 094 040 011 | - 1.081 - 1.103 966 542 438 473 466 506 522 489 3823 156 155 104 008 008 008 | - 1.1215 - 1.04753 - 1.0975 0975 0988 0975 5886 28510 28510 115 0025 0025 0025 | 309 - 1.131 - 1.199 - 1.070 - 1.0309866803472862747204171155128104015019045 | 091 078 053 030 006 |
| LOWER SURFACE | 55.00 | .3054 .2224 .1149 .0927 .0927 .0037 0036 0036 11030 11030 | 102 1180 120 122 107 083 047 | 068 089 115 111 101 096 045 003 | .241 .165 .108 .015 .015 .015 .029 .029 .084 .085 .086 .080 .080 | . 280 .193 .143 .071 .023 027 041 085 085 089 089 089 090 | 089 038 008 .029 | .268 .1790 .03807 .03807 .11262 .11264 .11247 .11106 .00799 .00465 |

TABLE I

BASIC WING

| Ī | PERCENT | | PRESSURE COEFFICIENT, P, AT: | | | | | | |
|---------------|--|--|---|---|---|--|---|---|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | |
| | M = 0.90 | $\alpha = 6.0^{\circ}$ | | | | | | | |
| UPPER SURFACE | 1250 5.050 10.000 15.0000 15.000 15.000 15.000 15.000 15.000 15.000 15.000 15.000 15.0000 15.000 15.000 15.000 15.000 15.000 15.000 15.000 15.000 15.0000 15.000 15 | 2579 208799 208799 208799 208799 20879 208 | 072 - 10008 - 17352 76263 76263 7426 | 10.2004 -1.2004 -1.2004 -1.2006 -1.77735 -1.49917 -1.53286 -1.55655 -1.5565 -1.5565 -1.55655 -1.556 | 0 86 - 1 237 - 1 2321 - 1 2321 - 1 2058 - 2544 - 25548 - 25548 - 255378 - 26163 - 2616 | - 1.3117 - 1.3117 - 1.2599 - 1.1572 - 1.0527 - 1.0521 6242 5963 3867 1159 1159 0047 0008 0008 0008 | 627 -1.297 -1.363 -1.208 -1.208 -1.207 -1.127 -1 | 81880 511100957286090681141009572844459068111111111111111111111111111111111111 | |
| LOWER SURFACE | 1.250000 1.25000000000000000000000000000000000000 | 38854 33854 3385445 3385445 33738345 30737365 30737375 30737365 3073756 | 2448 23788 22413 2149 21103 200444 - 20057 - 20057 - 200683 - 20763 - 20763 - 20763 - 20763 | .426 7350 .2770 .219 .1182 .0085 .0026 .0026 .0047 .0061 .0027 .0005 | 33172 33172 3236 31722 30030 11222 30030 10049 10049 10049 10023 10023 100248 | .435 .367 .284 .1286 .147 .0965 .0315 0046 0256 0433 0043 0043 0043 00443 00443 00443 00443 00443 00443 00443 00443 00443 00443 00443 00443 00444 0044 | | 43345719222442 43345719222442 110001354311097759611046 | |
| | M = 0. | $.90 \qquad \alpha = 7.9^{\circ}$ | | | | | | | |
| UPPER SURFACE | 1.25 2.5.00 7.5.00 10.000 15.000 25.000 30.000 30.000 40.000 50.000 50.000 70.000 70.000 70.000 85.000 85.000 90.000 | 523 464 502 533 523 420 | 330 - 1.139 - 1.119 - 1.031795600468468468516516557564199092 | - 1.3516 - 1.2556 - 1.275 - 8889 - 6809 - 66127 - 6636 - 6634 - 6642 - 592 - 286 - 1799 - 013 | 095 - 1.411 - 1.433 - 1.380 - 1.3225 - 1.1275 - 1.225 - 1.155786671676671675195044024022013 | 738 - 1.403 - 1.401 - 1.355 - 1.2187 - 1.232 - 1.190 - 1.133 - 1.023 - 1.023 - 1.031 | 876 - 1.079 - 1.098 989 1.9975 925 827 764 7663 663 565 525 494 460 418 356 317 231 231 196 | 750 7987 80822 7524 5274 5274 5274 4521 4521 33538 33538 33538 3264 2260 2260 2260 2260 2260 2260 2260 3320 | |
| LOWER SURFACE | 1.25 2.5.00 7.5.00 20.00 30.00 30.00 35.00 45.00 55.00 65.00 65.00 70.00 80.00 80.00 90.00 | . 413 . 444 . 428 . 407 . 376 . 323 . 247 . 148 . 127 . 092 . 058 . 014 . 004 . 006 . 006 | .517 .4522 .288 .221 .117 .105 .075 .022 .003 .023 .034 .034 .036 .038 .039 | . 489 . 427 . 351 . 352 . 265 . 210 . 1158 . 1255 . 088 . 059 . 035 . 0018 . 0018 . 0029 | . 449 .386 .356 .296 .242 .193 .1153 .118 .086 .060 .035 .016 0012 0014 0012 0014 0014 | . 422 .348 .292 .248 .2154 .116 .087 .062 .037 .014 .002 .021 .017 | .435 .353 .295 .247 .200 .147 .100 .062 .022 -011 -044 -075 -089 -092 -089 -092 -089 -092 -089 -092 -089 -092 -092 -092 -092 -096 -096 -096 -096 -096 -096 -096 -096 | .380 .294 .213 .154 | |

BASIC WING

| ſ | | | · · · · · · · · · · · · · · · · · · · | PRESSUR | E COEFFICIENT, | P, AT: | · · · · · · · · · · · · · · · · · · · | |
|---------------|---|---|--|---|--|--|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | $\alpha = 11.4^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.500 7.500 10.000 25.000 25.000 35.000 45.000 50.000 70.000 85.000 70.000 85.000 | 0 0 5 - 3 4 6 - 4 9 5 - 6 3 6 4 - 6 5 7 - 6 5 0 4 - 5 7 6 - 5 7 6 - 5 7 6 - 5 7 7 - 7 7 7 7 - 7 7 7 7 - 7 7 7 7 - 7 7 7 7 - 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 | 713 - 1.354 - 1.354 - 1.348 - 1.348 - 1.303 - 1.204 - 1.204 - 1.204 - 1.205 - 1.205 | - 1.4269 - 1.3582 - 1.3582 - 1.3582 - 1.2849 - 1.18317 - 1.0047 - 1.0047 - 1.09719273828671249182837 | 436 -1.073 -1.073 -1.185 -1.115 -1.129 -1.064 -1.048 -1.019 981 981 983 786 708 708 546 467 | - \$68 .523 .463 .463 .367 .299 .246 .203 .165 .125 .089 .059 .059 .009 .006 | - 361 | 4784024174073853703537318311307312312312312312312312312312312312312 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 30.00 40.00 40.00 50.00 60.00 60.00 60.00 85.00 90.00 | 388 4740 5557 5524 3863 222288 11500 1098 10000 10000 10000 10000 10000 10000 10000 | .587 .558 .498 .444 .410 .339 .250 .2176 .148 .082 .0647 .0 29 .008 .0108 .008 .008 .008 .008 .008 .00 | .522 .508 .448 .404 .313 .216 .176 .1158 .059 .029 .029 .029 .017 .0185 .0296 | . 508 . 4475 . 3963 . 2888 . 2406 . 1052 . 0099 . 0423 . 0005 . 0018 . 0710 . 0710 . 1155 | - 824 - 8175 - 7658 - 77466 - 7702 - 6667 - 6624 - 6624 - 558 - 598 - 488 - 488 | 489 506 506 501 487 446 426 426 426 426 426 426 426 436 - | . 439 . 414 . 2334 . 2033 . 1009 . 00387 1252 1255 1785 1785 1809 2213 2244 2243 2245 |
| | M = 0. | .90 $\alpha = 15.6^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 15.00 20.00 30.00 40.00 45.00 45.00 60.00 65.00 60.00 75.00 80.00 90.00 | 7180 6722 6666 5537 6593 6593 6619 6619 4448 | - 14444357 - 114444357 - 11444435916 - 11444435916 - 11444435916 - 11441000003955 - 11441000003955 - 11441000003955 - 114410000003955 - 11441000000000000000000000000000000000 | 1 | - 267310 - 277710 - 27779820 - 27779820 - 27779830 - 27774318 - 27774318 - 27774318 - 27774318 - 27774318 - 2774318 - 2774318 | - 6840 - 66666 - 665650 - 665650 - 66292 - 66222 - 66104 - 55878 - 55778 - 55778 - 55778 | 75969972288885584368684368429 344724444444444444444444444444444444444 | 443555544508445193545676 |
| LOWER SURFACE | 55.00 | 3483 34843 376124 365124 34935 35143 35143 35143 37308 319662 31015 30043 | .621 .646 .618 .575 .4475 .4475 .4475 .298 .298 .2299 .239 .2191 .148 .121 .080 .063 .022 -086 | 216174384416755447238443624111005 003636 003636 003636 | 15316 1444815 1444835 12328508 1444835 12328508 1628 1628 1628 1628 1628 1628 1628 162 | .471 .510 .481 .4406 .3518 .2445 .1653 .0406 .0055 .1080 .0055 .1589 .292 | - 498 4448 4406 -3991 -3250 -1197 -0843 -0163 -1099 -1187 -2855 -3334 | 1982334173234477344773447734477344773447734 |

TABLE I

BASIC WING

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | | · |
|---------------|---|--|--|---|---|---|--|---|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 90 α = 19.8 ⁰ | | | | | | |
| UPPER SURFACE | 12.5000000000000000000000000000000000000 | 0 4 3 0 6 0 3 4 - 1 . 1 2 7 - 1 . 1 2 7 - 1 . 0 8 3 - 1 . 9 8 0 - 9 8 5 9 8 - 7 9 8 5 - 7 3 8 6 - 7 3 8 6 - 6 6 2 8 8 6 1 4 - 6 6 5 5 - 6 6 5 8 6 6 1 4 - 6 6 5 5 | - 1.086 - 1.0867 - 1.0862 - 1.0862 - 1.0760 - 1.0717 9879 9754 99536 8766 8766 | 9115 9057 8887 8887 8853 88487 85291 85487 822131 8008 7748 77323 | 859 7663 7762 7782 7752 7552 7495 7445 7737 7736 7726 77217 7715 6899 6820 664 | 7 2 4 4 7 7 2 8 0 7 2 | 5 5 9 9 8 0 4 6 2 3 5 5 7 1 6 5 3 2 8 9 8 5 2 7 1 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 | 520 5218 5222 5222 5222 5007 5009 5113 5114 5144 5046 5 |
| LOWER SURFACE | 1.25 2.500 70.500 10.500 15.000 25.000 35.000 40.000 50.000 65.000 75.000 65.000 75.000 85.000 95.000 | 2 449079 - 849079 - 7194 - 66046 - 64466 - 44607 - 44607 - 33069 - 28528 - 1881 - 1085 - 0852 | .604 .686 .6890 .6636 .5749 .488 .440 .366 .2887 .2887 .2280 .118 .0042 -1104 | .487 .5901 .5800 .514 .417 .3737 .2961 .218 .115 .0402 -0402 -1110 | 533 5553 5553 5537 4572 43636 2844 2197 1157 10846 - 02599 - 01632 | . 423 .5118 .4956 .44618 .3207 .239 .1195 .10627 .0007 .0007 .11600 .288 | . 499 . 484 . 4557 . 373 . 3147 . 2609 . 1059 . 1055 - 0051 - 098 - 1398 - 2591 - 2591 - 3372 | . 378 . 4316 . 4316 . 3513 . 228 . 1062 . 1062 . 1155 . 11922 . 1292 . 2368 . 291 . 2368 . 291 . 3162 . 327 . 327 . 328 . 328 |
| | M = 0. | | r | _ | T | | <u> </u> | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 65.00 65.00 65.00 75.00 85.00 | 161 47252 98108 98909 99045 88498 86672 884985 864985 864985 864985 864985 864985 | - '.920 903 898 8990 905 905 906 906 906 906 906 906 906 906 906 906 906 908 | - 901 - 881 - 862 - 887 - 876 - 8864 - 877 - 876 - 8864 - 874 - 861 - 8867 - 8861 - 8827 - 845 | - 804 - 7780 - 7780 - 7892 - 8050 - 8117 - 81187 - 81187 - 81187 - 81187 - 81187 - 81187 - 7787 - 7787 - 7787 - 7787 | 811 793 783 783 783 7774 7776 7776 7776 7769 738 7318 7318 | 781 769 7893 8593 8593 767 701 6678 6775 6744 6744 6742 6424 650 | 64 1 65 154 58 95 59 98 59 98 66 1134 66 1133 65 79 90 66 1134 65 79 90 65 79 90 55 75 80 55 |
| LOWER SURFACE | 1 .25 2 .50 0 .75 0 .00 10 .00 25 .00 35 .00 40 .00 45 .00 55 .00 66 .00 67 .00 80 .00 90 .00 | . 0 627 . 6 63 8 . 8 777 . 6 6 6 3 8 . 777 . 6 8 8 9 8 . 777 . 6 8 9 . 6 4 1 6 6 0 6 3 6 8 4 8 6 9 4 3 7 1 9 9 2 8 3 6 8 1 1 2 | .539 .7668 .7628 .7505 .6688 .5408 .5408 .5408 .3944 .3311 .2539 .0888 .0988 | .626 .583 .585 .505 .470 .393 .350 | .560 .603 .598 .533 .491 .414 .373 | 320 492 4557 5648 5118 44353 3354 266 2226 1182 115 0327 - 0927 | .474 .518 .518 .470 .472 .373 .3727 .278 .227 .174 .174 .174 .174 .186 .237 .313 | 273 4437 4137 4137 30346 1659 00531 - 0061 - 11474 - 11862 - 2653 - 2259 - 3352 - 3352 |

TABLE I

BASIC WING

| Ī | - | PRESSURE COEFFICIENT, P, AT: | | | | | | |
|---------------|--|---|---|---|--|--|--|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | 94 α = 0.0° | | | | | 1 | |
| UPPER SURFACE | 1.250 57.500 105.000 1 | 027 02366 0054 0055 00355 | - 1042 - 1042 - 1042 - 1042 - 1042 - 1076 - 1119 - 1141 - 1167 - 1208 - 1208 - 1257 - 1267 - 1274 - 1274 - 12136 - 12136 | - 468 - 0984 - 0984 - 11026 - 11745 - 11745 - 1225 - 1225 | 7 10 20 5 7 10 20 5 7 10 20 5 7 10 20 5 7 10 20 5 7 10 20 20 20 20 20 20 20 20 20 20 20 20 20 | 445 0750 1267 12910 2359 2499 2759 2172 1715 1611 138 1062 0149 0029 .062 | - 4852 - 1771 - 1978 - 1131 - 1157 - 1157 - 1157 - 1157 - 1202 - 12224 - 1224 - | 510777809778099080 408897778099045890985897044594 |
| LOWER SURFACE | 1.25.000 70.000 115.000 125.00 | 2091 21078 2091 2091 2002 2002 2002 2003 | - v086 - v057 - v057 - v0723 - v116 - v1152 - v1177 - v202 - v247 - v247 - v278 - v297 - v297 - v272 - v272 | - 155 - 1111 - 11289 - 1712 - 12117 - 12117 - 12255 - 12867 - 13335 - 13336 - 13336 - 13336 - 13006 - 10006 - 10006 | - 1888 - 11019 - 11649 - 121649 - 1226436 - 12 | 244 205 2318 2247 22804 22804 22804 22109 11601 11601 0054 0020 | - 171 - 181 - 184 - 184 - 184 - 189 - 1227 - 12221 - 12221 - 12251 - 1169 - 1015 - 1016 - 101 | 64686204605728866714144 564686204605728866714144 8888888811110000000000000000000000 |
| | M = 0.9 | $\alpha = 4.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 1000 1.255 5.000 5.050 10.000 20.000 20.000 35.000 45.000 45.000 45.000 65.000 75.000 85.000 95.000 | 028 - 0367 - 0118 - 1150 - 1150 - 1159 - 1282 - 2231 - 2244 - 2261 - 2371 - 2371 - 2376 | 244 - 1823 - 1726 - 1312 - 1310 - 1287 - 1287 - 1287 - 1287 - 1287 - 1302 - 1302 - 1316 - 1404 - 1373 - 1404 - 1373 | 120 - 19351 - 18551 - 15396 - 13881 - 137384 - 13784 - 1453 - 1466 - 1472 - 1472 - 1477 - 1472 - 1477 - 1475 - 1477 - 1477 - 1475 - 1477 - 1477 - 1477 - 1477 - 1477 - 1477 - 1477 - 1477 - 1475 - 1477 - 1475 - 1477 - 1475 - 1477 - 1475 - 1477 - 1475 - 1475 - 1477 - 147 | 7365 - 7974 - 7974 - 74858 - 74432 - 74456 - 74556 - 74556 - 74564 - 75521 - 75521 - 75537 - 74487 - 74487 - 7464 - 75021 - 75 | 057 946 976 895 895 8713 479 520 538 5576 5793 5199 160 043 043 0038 078 | 1279747 - 1.9313 - 1868 - 1767 - 15514 - 15568 - 15568 - 1557 - 1009 - 1009 - 1009 | - 1 48 47 2 |
| LOWER SURFACE | 1.25 2.500 70.500 15.000 25.000 35.000 40.000 50.000 60.000 70.000 80.000 80.000 90.000 | *3481 *32644 *22654 *22057 *1503 *10046 *00121 *00322 *10049 *112241 *11250 *11250 *112543 | **339 **269 **2269 **201 **1533 **0732 **0028 **0028 **0080 **0112 **1143 **143 **143 **143 **145 **1156 ** | 318 4245 4167 1118 1087 1044 - 0036 - 0074 - 1093 - 1142 - 1158 - 1162 - 1158 - 121 - 1087 - | 7272 2208 2208 21922 11925 10079 1018 1018 1134 1134 1134 1104 1004 1004 1004 1004 1005 1006 100 | .312 .236 .149 .098 .058 .032 014 064 080 092 111 113 103 083 044 048 | 278 176 126 126 1093 0045 - 0058 - 1092 - 1118 - 1153 - 1189 - 1205 - 1194 - 1006 10031 1053 | 2977612276122762224984085387715249846782224985387711298467822249853877112984678222498467 |

TABLE I

BASIC WING

| Ī | | - | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|--|--|---|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 94 α = 5.9 ⁰ | | | | | | |
| UPPER SURFACE | 12.5000000000000000000000000000000000000 | 0245 00347 | 0176 - 10702 - 6678 - 65492 - 4428 - 44054 - 33990 - 44245 - 44650 - 4439 - 4848 - 4837 - 3382 | 0 9 8 6 4 6 9 9 1 1 1 1 4 9 9 1 1 1 1 1 4 9 9 1 1 1 1 | . 2045 - 1.1927 - 1.1022 - 1.00167 - 1.5312 - 1.55254 55254 55254 552599 6550 2973 10099 | | 410 - 1.181 - 1.243 - 1.1417 - 1.085 - 1.025 - 1.0990 948 9447 2649 4473 1133 .0061 .0744 .0856 .1086 | - 1.269 - 1.269 - 1.269 - 1.261 - 1.175 - 1.152 - 1.106 - 1.0769 - 1.2522 - 1.426 - 1.3499 - 1.3497 - 1.0768 - 1.068 - 1.0647 - 1.034 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 45.000 65.000 75.000 85.000 85.000 85.000 | .3986 .3986 .3396 .2246 .1881 .1212 .0068 .00006 .00006 .0065 .00782 .00782 .1110 | .452 .3809 .2531 .154 .1114 .041 .041 .013 074 .0712 080 090 095 1099 1099 | .425 .345 .214 .1265 .078 .078 .0021 047 091 106 113 | . 387 .307 .2287 .2287 .222 .160 .104 .031 .0029 .053 .076 .089 .096 .096 .096 .096 | . 4 09 .334 .250 .1953 .1112 .028 .0061 .024 045 065 074 065 079 065 079 065 | .360 .262 .2062 .118 .025 007 0476 113 1179 1759 075 0015 0016 | - 10008948899568996512006564000000000000000000000000000000000 |
| | M = 0 | $.94 \alpha = 7.9^{\circ}$ | | | ' | T | | 1 |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 15.00 20.00 25.00 35.00 40.00 45.00 55.00 60.00 65.00 75.00 | 0 2 2 117 2 142 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 | 202 - 1.073 - 1.067 - 1.023 908 528 441 425 4459 4459 490 5022 483 528 | 282 - 1.275 - 1.219 - 1.209 - 1.099 921 5544 5537 5548 5683 5683 5883 5883 5883 | .033 -1.253 -1.282 -1.282 -1.173 -1.092 -1.020 979 626 623 623 623 623 623 649 | 561 - 1.286 - 1.296 - 1.241 - 1.196 - 1.173 - 1.1079 - 1.028 - 1.005 978 849 754 666 483 342 130 077 054 | 675 - 1.259 - 1.317 - 1.228 - 1.186 - 1.152 - 1.104985793715656652652582554493426380328 | - 627 - 7314 - 7592 - 7726 - 7229 - 7229 - 6835 - 6835 - 6935 - 4968 - 4968 - 4188 - 33788 - 3398 - 3398 |
| | 85.00 90.00 95.00 | 492 | 492 359 200 | 274 190 127 | 125 060 019 | 041 | 262 | 312 |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 25.00 35.00 45.00 45.00 55.00 65.00 | . 430 . 460 . 443 . 424 . 338 . 259 . 195 . 1151 . 133 . 096 . 061 . 011 | . 528 . 466 . 395 . 335 . 301 . 229 . 185 . 1145 . 109 . 077 . 053 . 021 . 021 . 034 | . 490 . 424 . 345 . 297 . 257 . 257 . 154 . 076 . 046 . 020 - 0037 - 045 - 056 | .0960 .0333 .0021 044 0604 064 | .394 .258 .217 .168 .118 .078 .049 .049 .0050 0050 0050 | .034 005 041 079 120 145 161 160 | .284 .150 .043 126 1286 229 246 246 245 163 117 |
| | 75.00 80.00 85.00 90.00 95.00 | 019 050 063 | 060 085 095 | 072 063 063 | 038 020 011 | 038 028 019 | 082 074 067 | 080 |

BASIC WING

| 1 | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|--|---|--|--|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 94 $\alpha = 11.4^{\circ}$ | <u> </u> | | | | | |
| UPPER SURFACE | 12.5000000000000000000000000000000000000 | 0112 644 055 4795 775 8365 164 9644 9944 8304 8305 3745 53745 53745 53745 53745 5374 | 585 - 1.2428 - 1.22612 - 1.2622 - 1.1774 - 1.0019 8514 514 4985 5641 5641 56764 5697 4044 | 648 - 1.279 - 1.208 - 1.1089 - 1.038 991 9941 871 871 840 813 763 765 765 765 460 405 405 405 405 | 253 - 1.156 - 1.155 - 1.142 - 1.142 - 1.145 - 1.101 - 1.059 - 1.027 986 9830 847 731 849 753 | 871 99547 99547 99519 99519 99519 88341 74402 66445 66445 55475 4477 | 738 699 680 6670 6670 6638 6677 6538 5543 5543 5523 5503 4400 4426 41397 | 5605822459052390890040661 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 30.00 40.00 45.00 55.00 55.00 60.00 65.00 70.00 75.00 85.00 | 4 08 5 56 5 66 5 68 6 1 6 4 88 4 116 6 318 7 2 5 09 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 | .607 .574 .513 .461 .4266 .3366 .2281 .1627 .1097 .0974 .0053 .0158 .0158 .0158 .0990 | .540 .5158 .4104 .3718 .2620 .1829 .1146 .0537 .0330 .0337 .03416 .098 | .521 .479 .449 .397 .348 .289 .242 .197 .1627 .093 .003 .009 -009 -007 -007 -1077 -1137 | .492 .476 .4167 .3656 .2271 .2174 .1379 .0658 .0001 .0353 .072 | .469 .396 .344 .295 .247 .135 .0889 0054 1054 1054 1212 2460 270 2911 315 | 4454 44154 44154 44154 413595 4111290 41169191 411691 41691 |
| | M = 0 | .94 α = 15.7° | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 15.00 20.00 25.00 40.00 45.00 55.00 60.00 70.00 80.00 95.00 | 820 714 6743 66318 6631 66147 5940 5774 6103 6505 | 555 397 339 | - 1.047 - 1.1065 - 1.1080 - 1.0881 - 1.0981 - 1.0918 - 1.099782 99452 99452 99880 98880 98880 7550 7550 | | 755875587451734473347320370966681466567661056610566930 | 69946 59946 59831 558831 555711 55545 5543413 553341 553367 554343 5543443 55434443 55434443 55434443 55434444 55434444 55434444 5543444 5543444 5543444 5543444 5543444 5543444 554344 554444 554 | 22993034 21101100930781 555554993781 |
| LOWER SURFACE | 1.255 2.500 5.000 10.000 25.000 25.000 35.000 40.000 55.000 60.000 85.000 90.000 95.000 | .350 .500 .500 .7315 .6029 .5311 .33726 .33726 .2852 .22076 .1120 .0043 .0043 | .66326 .66326 .55856 .44386 .35555 .44386 .3755 .22437 .11634 .11634 .11634 .1176 .1 | .5481 .5855 .5855 .4440 .3838 .2634 .1525 .1074 .00034 .0055 | . 5642 . 5427 . 5427 . 4466 . 3346 . 2269 . 2280 . 1144 . 1074 . 0417 . 0417 . 0417 . 0417 . 1058 . 2282 | .4717 .4857 .44107 .35529 .22594 .117342 .00516 .00140 .00140 .00140 .117416 | - 496 - 451 - 3161 - 3160 - 2086 - 1057 - 0052 - 1182 - 2809 - 3393 | # 44381671 # 44381671 # 5218851 # 52 |

TABLE I

BASIC WING

| | PRESSURE COEFFICIENT, P, AT: | | | | | | | |
|---------------|--|--|--|---|--|---|---|---|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 94 $\alpha = 20.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.55 5.00 10.000 15.000 25.000 35.000 45.000 45.000 55.000 65.000 85.000 85.000 | | 1 . 17 0 1 . 25 17 1 . 22 18 1 . 23 18 1 . 21 18 1 . 12 10 1 . 11 4 0 1 . 11 4 0 1 . 11 0 0 1 . 10 0 2 7 1 . 95 8 2 8 3 6 8 3 6 8 6 7 6 0 1 . 76 0 1 . 76 0 1 . 76 0 1 . 76 0 | 9851 9951 9953 9955 9955 9955 9955 9957 9957 9957 | 899 7990 7990 8200 8200 8111 8091 7794 7758 7758 7758 7759 7749 7749 7720 7200 7200 7152 793 | 821 818 808 7425 718 715 705 705 6892 6770 675 6485 675 6485 675 | 638 632 641 625 636 631 620 602 612 605 605 605 605 605 605 598 578 578 578 578 558 | |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 25.00 35.00 40.00 40.00 50.00 65.00 65.00 85.00 85.00 95.00 | 2 58 4 5 5 6 6 4 5 5 5 9 5 4 4 1 1 4 4 5 5 4 2 2 1 4 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 | .633 .7116 .688 .658 .658 .658 .4730 .398 .3213 .263 .2263 .1643 .1052 .039 | .512 .6624 .6027 .5339 .4446 .3770 .3394 .2526 .1970 .0510 .00177 .125 | .564 .579 .5581 .55569 .4846 .34963 .32737 .1963 .1294 .0207 .1024 | .448 .5344 .5237 .4529 .4529 .3557 .3111 .2286 .1487 .00735 | .516 .504 .4730 .4038 .2999 .2447 .1945 .00410 00548 1657 2977 2597 | . 390 . 4534 . 3844 . 3844 . 3846 . 1693 . 0038 . 0038 . 0137 . 1771 . 2258 . 2258 . 3928 . 3928 . 3928 . 3928 . 3928 . 3928 . 3928 . 3928 . 3928 . 3928 |
| | M = 0. | | | T | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 25.00 35.00 45.00 45.00 65.00 65.00 65.00 85.00 85.00 | 1 | l | l l | | - 7896 - 7551 - 7551 - 7548 - 7448 - 7445 - 7745 - 77327 - 77221 - 77221 - 7221 - 7221 | 717 695 694 7138 702 6740 6659 6659 6654 6554 6554 6514 6349 5976 | 647 606 567 579 588 588 596 601 601 601 597 557 557 557 557 557 557 |
| LOWER SURFACE | 1.25 5.00 7.50 10.00 20.00 25.00 35.00 40.00 45.00 60.00 75.00 60.00 75.00 80.00 95.00 | . 548 . 512 . 437 . 424 . 365 . 314 | .601 .7734 .773 .761 .742 .691 .609 .528 .496 .414 .384 .3154 .328 .228 .228 .228 .159 | .616 .570 .529 .493 .455 | .500 .500 .507 .509 .559 .521 .481 .447 .404 .363 .323 .241 .241 .272 .095 .016 .055 .126 | .348 .307 .262 .224 | .507 .532 .524 .493 .464 .368 .321 .271 .224 .174 .122 .071 .003 003 | .330 .440 .454 .424 .3340 .251 .1636 .0546 -050 -11436 -1176 -1252 -2252 -2366 -3362 -3362 |

BASIC WING

| | DEDOENM | | | PRESSUF | RE COEFFICIENT, | P, AT: | | | |
|---------------|--|---|---|---|--|---|---|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | |
| | M = 0. | 98 α = 0.0 ⁰ | | | | | L., | <u> </u> | |
| UPPER SURFACE | 12.5500000000000000000000000000000000000 | .036 .2259 .10825 .00825 .00827 .00877 008921 11400 11587 11587 123802 28802 2864 | 526146307420842126412641459181422542276522765238762408 | . 4 4 2 201 1132 1136 1236 1236 2256 2256 23014 3205 320 | . 7 0 7 22 0 4 175 8 175 8 23 11 22 63 3 32 0 8 33 4 6 33 4 6 33 4 6 34 6 35 6 6 35 | . 37 5 2028 238 1218 219 2347 3126 3469 3469 3450 4550 4550 4550 123 123 079 | . 38 3 220 320 0 224 4 303 3 318 1 367 7 3919 1 4407 1 4407 1 493 1 343 1 146 1 045 1 045 1 045 1 045 1 | . 37 7 7 . 2866 | |
| LOWER SURFACE | 25000000000000000000000000000000000000 | 262 2145 2145 2145 20449 20023 200527 | .038 .042 .0021 .0021 .0058 .0714 .1540 .1228 .1540 .2346 .2357 .2453 | 022 0039 0543 05743 1353 1808 2367 2367 2367 2966 2367 2966 2367 2966 2367 2966 | 1 072 072 0520 071 1155 1285 1280 1281 1337 1349 1349 1349 1349 1349 1349 1349 1349 1349 1349 1356 1349 1367 1377 1 | 1141035179123582806336183689406435111853218532 | 105 163 1914 232 287 325 378 408 444 449 449 449 449 182 036 .006 .013 | 1877 18343 2521 3604 4257 4485 4455 4455 4450 0470 00729 00470 00470 1317 | |
| | M = 0.9 | 98 α = 3.9° | | | | · | | | |
| UPPER SURFACE | 00 5 5 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 | .041 .107 .0168 .1108 .1108 .1346 .1346 .11794 .117 | .305 750 659 277 2660 2532 2541 3012 3122 3359 3359 3359 3359 3359 3359 | . 191 848 7828 3209 3219 3319 3345 3462 3402 4429 4429 4429 4429 4429 4429 | . 423 - 809 - 856 - 751 - 457 - 3749 - 3749 - 3749 - 4412 - 4465 - 4810 - 483 - 4 | . 0 4 2 . 8 2 3 . 8 6 3 . 7 6 2 9 . 7 6 2 9 . 4 0 3 0 . 4 5 7 3 . 4 5 7 3 . 5 5 0 3 . 5 5 6 3 . 5 5 6 3 . 5 5 6 3 . 5 6 9 . 1 2 2 8 . 1 2 8 | . 027 867 9019 7746 646 474 5513 5576 559 66559 172 1055 1055 0046 | . 1057 9103 7783 7773 7443 4483 4883 55744 55764 55764 1522 10042 10042 10042 | |
| LOWER SURFACE | 1.25 2.50 | .368 .348 .292 .261 .1830 .1350 .0059 .0010 0045 0089 1093 1119 1149 1449 210 | . 358 .288 .280 .176 .087 .087 .024 035 035 124 154 154 158 158 158 | .332 .259 .184 .107 .0515 023 057 179 166 170 182 190 156 15 | . 284 . 231 . 204 . 1043 . 0025 . 0015 0048 0771 1081 1611 185 2002 1145 1144 0073 0046 | .290 .215 .039 .007 082 111 1407 199 199 1755 118 110 110 | .222 .121 .068 .030 .0150060092111013581283229722972837292273207 | . 278 . 118 . 0406 . 0968 1469 | |

TABLE I

BASIC WING

| ſ | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|--|---|--|--|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| П | M = 0.9 | 8 $\alpha = 5.9^{\circ}$ | 1 | | <u></u> -L | | <u>.</u> | |
| UPPER SURFACE | 1.250 5.000 10.000 15.000 25.000 25.000 45.000 45.000 65.000 65.000 75.000 85.000 85.000 | .038 .017 .087 .175 .219 .2219 .2284 .2264 .2264 .2262 .2272 .331 .331 .3335 .3331 .3323 .3331 .3331 .336 | - 102 - 9924 - 9926 - 1586 - 4196 - 3515 - 343 - 3585 - 3852 - 3852 - 3852 - 3899 - 4432 - 3431 - 3599 | 006 - 1.044 - 1.014 957 880 380 422 380 409 444 464 464 494 450 482 882 | . 311 - 1:0153 - :9919 - :872 - :812 - :514 - :430 - :458 - :476 - :502 - :517 - :543 - :573 - :574 - :566 | 206 - 1.032 - 1.038 - 1.0058 - 1.000 927 875 875 875 5715 533 560 625 646 414 216 161 123 | - 238 - 1:0382 - 1:0382 - 9872 - 9943 - 9943 - 8956 - 8666 - 8666 - 8666 - 7661 - 76620 - 76620 - 76824 - 7682 | 1007845 10079876456 111999764568 111999764568 1119999999999999999999999999999999999 |
| LOWER SURFACE | 1.25.00000000000000000000000000000000000 | 413 4418 4375 33447 22999 119869 1098 | 4588674 4588674 45886674 45886 4588 | 9222336566227 2396566227 21836566227 21836566227 21836566227 21836566227 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 2183656627 21836657 21836657 21836657 21836657 21836657 2183667 218367 | 379 3282 3282 32825 32052 3205 | .387 .3226 .1729 .087 .032 0361 0861 1136 1140 1140 115 116 | 322743743 .00366377336 .00366376316499 .007031649 .007031649 | 2021164304899157425280 7917819060246875287140 782100022533333333321100 |
| | M = 0. | 98 α = 7.8° | | | | | | |
| UPPER SURFACE | .00 1.25 2.500 7.500 15.00 25.00 35.00 40.00 40.00 65.00 65.00 75.00 85.00 | .0 3 0 0 614 2 85 3 18 3 20 3 35 3 35 3 35 3 35 3 36 3 46 3 48 3 4 | 106 - 1.030 - 1.023910749674557428400423437457457457457457457457457457457457457457457457453 | - 181 - 1.148 - 1.105 - 1.028 - 1.028 - 1.02800190563449546847749153115344549541548465303154 | - 1 . 111 - 1 . 149 - 1 . 094 - 1 . 006 974 9917 9912 717 584 5557 574 647 260 2191 121 | 429 -1.139 -1.160 -1.108 -1.064 -1.0059939279968943714666674666435238190 | 488 - 1.116 - 1.176 - 1.088 - 1.052 - 1.0379849972961956956956956956936936936936 | 440 - 1.180 - 1.084 - 1.069 - 1.0423 - 1.0423 - 1.0423 9750 9955 9955 86593 66629 6642 66431 5500 462 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 25.000 35.000 45.000 55.000 65.000 75.000 85.000 90.000 95.000 | . 4 4 5 . 4 76 5 . 4 36 3 . 3 4 6 . 2 7 2 . 2 0 4 . 1 6 3 5 . 1 0 5 . 0 6 9 . 0 0 1 . 0 0 9 . 0 0 9 . 0 0 9 | .539 .479 .396 .342 .308 .240 .155 .120 .086 .058 .025 -014 -018 -033 -044 | . 488 . 423 . 346 . 298 . 208 . 116 . 079 . 0149 . 010 . 035 . 060 . 060 . 085 . 079 . 083 . 082 | . 456 .379 .336 .284 .2832 .1731 .089 .024 .010 .005 .006 .0076 .006 .0090 .0090 .0082 .0082 .0082 | . 446 .382 .300 .2463 .1562 .063 .031 028 028 060 080 099 104 105 115 115 115 | .387 .296 .243 .193 .199 .094 .012 -023 -106 -147 -191 -2245 -245 -245 -2114 -137 -041 | .415 .350 .261 .1822 .0346 1525 2733 3356 3766 3386 3386 174 135 |

TABLE I

BASIC WING

| į | | | | PRESSU | RE COEFFICIENT, | P, AT: | | ··· |
|---------------|--|---|--|--|--|---|--|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | $8 \alpha = 11.5^{\circ}$ | | | | | | |
| UPPER SURFACE | . 0 0 1 2 5 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 | . 0 6 6 180 | 465 - 1.125 - 1.11425 - 1.11425 - 1.0522 - 1.0522 9627 8650 4414 5520 4414 5520 5433 5433 5435 4555 | - 1.1769 - 1.17523 - 1.0587 - 1.058798617986177759774477719771666094804351 | - 121 - 1.284 - 1.2514 - 1.2514 - 1.2514 - 1.1927 - 1.0766 - 1.0998 9928 8534 8534 8556 5319 3856 3276 | 758 - 1.3103 - 1.2812 - 1.2832 - 1.28390 - 1.1447 - 1.10937 - 1.06359564832883288328781376681 | 861 - 1.086 - 1.086 - 1.0457 - 1.0486 - 1.0457 - 1.0486 - 1.030 9333 8944 798 7655 6653 | 7917789807990777898379983799837998379983799 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 35.00 45.00 45.00 65.00 65.00 85.00 95.00 | 4 3 9 5 3 4 5 8 5 6 1 7 8 5 5 6 1 7 8 5 5 4 4 0 5 5 2 8 8 9 8 9 8 9 8 9 8 9 8 9 8 9 8 9 8 9 | .640 .602 .541 .487 .4543 .338 .252 .188 .159 .100 .087 | .563 .536 .471 .427 .3917 .282 .2241 .203 .1377 .109 .0627 .027 .030 .051 | .558 .495 .405 .408 .359 .2599 .1740 .104 .072 .0043 .019 .002 .014 | .503 .478 .417 .364 .273 .219 .175 .140 .072 .033 .005 -022 -043 -059 | . 481 . 408 . 3565 . 2559 . 2059 . 1577 . 1165 . 0022 00737 1154 181 2259 2887 344 | .460 .4356 .281 .23145 .0046 1176 2147 22778 23000 3126 33126 33564 392 |
| | M = 0. | | | | l., | J | 1 | |
| UPPER SURFACE | 1.25 2.50 7.50 10.00 20.00 35.00 35.00 40.00 45.00 45.00 60.00 65.00 70.00 75.00 80.00 90.00 95.00 | 157 325 5304 741 7329 6731 66731 5566 5566 5581 5800 6634 6634 6634 6634 6634 6634 6634 | 839 - 1.263 - 1.263 - 1.263 - 1.267 - 1.167 - 1.167 - 1.157 - 1.158 - 1.759 - 596 - 630 - 630 - 491 | 933 - 1.193 - 1.177 - 1.176 - 1.150 - 1.177 - 1.177 - 1.173 - 1.177 - 1.162 - 1.1144 - 1.1157 - 1.124 - 1.1124 - 1.156962962750 | | - 1.040 937 933 933 933 925 911 921 878 858 811 770 750 759 687 687 6641 502 | 665 660 6641 647 642 632 625 602 619 610 605 604 611 610 604 610 604 597 | |
| LOWER SURFACE | 1.25 2.50 7.50 7.50 20.00 20.00 30.00 30.00 40.00 65.00 60.00 60.00 80.00 90.00 95.00 | 3789 5702 7729 6580 5463 4401 3722 2725 2725 2725 2725 2725 2725 2725 | .678 .694 .660 .617 .583 .518 .471 .427 .385 .346 .276 .276 .244 .219 .196 .175 | .574 .604 .578 .546 .513 .462 .410 .366 .328 .292 .257 .224 .192 .144 .112 .062 .049 .039 .030 .007 | .589 .565 .550 .509 .421 .374 .333 .296 .255 .222 .186 .155 .127 .106 .082 .039 .018 .039 .018 | .537 .513 .475 .441 .395 .342 .300 .224 .190 .147 .017 .084 .056 .030 | .529 .488 .448 .436 .362 .307 .255 .208 .162 .016 .066 .014 074 113 179 193 292 341 | |

TABLE I

BASIC WING

| Γ | | | | PRESSURI | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|--|--|--|--|---|
| | CHORD | 0.135ь/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0,95b/2 |
| 寸 | M = 0.98 | 8 α = 20.2° | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 20.00 25.00 30.00 35.00 40.00 45.00 55.00 65.00 70.00 85.00 85.00 85.00 | - 258 - 408 - 685 - 9351 - 9351 - 939 - 918 - 8469 - 7614 - 6654 - 6654 - 6664 - 6674 - 6674 - 6674 - 6674 - 6673 - 6633 - 6634 | - 1.067 - 1.217 - 1.206 - 1.2110 - 1.219 - 1.183 - 1.1910 - 1.152 - 1.160 - 1.158 - 1.128 - 1.128 - 1.128 - 1.014 669 660 583 544 460 385 | - 1.195 - 1.070 - 1.028 - 1.064 - 1.047 - 1.050 - 1.050 - 1.057 - 1.030 - 1.03 | 931 8550 8556 8553 8656 8656 8656 8656 8647 8440 8258 8191 7981 77661 | 808 787 7800 780 780 7732 7773 786 7770 77570 75570 75570 75570 75570 75570 75570 75570 7644 75570 75570 75570 75570 7646 77570 7646 77570 | 72771047067086699868976810668766816687668766876687668766876687668766876687668766876687668766876687 | 2 8 5 2 5 2 7 5 1 3 1 0 0 2 2 3 9 3 8 0 3 2 5 2 6 6 6 2 2 2 1 3 1 0 0 2 2 3 9 3 8 0 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 |
| LOWER SURFACE | 95.00 1.250 5.000 10.000 15.000 25.000 45.000 45.000 65.000 85.000 85.000 | 279 2517 7668 8368 8368 7765 6655 6658 5525 4489 4076 3364 3364 3364 3364 3364 3364 3364 33 | .665 .740 .742 .6836 .5842 .5842 .5900 .428 .3851 .3258 .227 .221 .2434 .010 | .542 .641 .650 .631 .604 .5164 .5173 .435 .391 .3291 .262 .235 .290 .131 .100 .0648 .061 | .569 .603 .580 .586 .513 .473 .391 .313 .237 .200 .171 .141 .071 .042 .005 .116 | .464 .5537 .5547 .5437 .476 .4267 .383 .3405 .264 .222 .184 .147 .111 .089 .024 021 120 180 | .534 .528 .501 .463 .429 .379 .326 .282 .236 .184 .003 .004 .004 .011 .181 .214 .234 .234 | .410 .479 .479 .4176 .305 .202 .134 .0066 0555 1349 1210 2570 2720 2730 3353 389 |
| | M = 1. | $00 \alpha = 0.0^{\circ}$ | | · | | | 1 | 1' |
| UPPER SURFACE | 90.00 | .074 .288 .2257 .1157 .0957 .0050 .0 | .582 .006 .011 .005 .021 .025 .036 .057 .1029 .1127 .1528 .1127 .222 .222 .222 .222 .2237 .247 .247 .247 .247 | .5030450340550961211475196248259265286298259164 | .7470740820971041641952222452853393316184115085 | 0471210881631632492632307307373390350350350350360360360360 | 183 125 071 027 | 022 .030 .066 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 35.000 45.000 65.000 65.000 70.000 85.000 90.000 95.000 | 058 0697 136 151 1697 1997 1904 184 237 | 013 .012 .011 007 024 052 064 083 1138 1158 1224 233 235 235 235 | 045 069 069 1138 1155 179 224 225 2281 2290 299 294 1219 | - 103 - 093 - 1164 - 187 - 227 - 227 - 279 - 3306 - 345 - 345 - 293 - 126 - 126 | - 144 - 159 - 2015 - 2015 - 2015 - 2015 - 2015 - 2015 - 2015 - 3016 - 30 | 160 197 211 228 248 278 374 352 373 403 403 473 418 227 149 | 213 202 245 3374 378 391 426 444 459 420 204 204 204 204 204 |

TABLE I

BASIC WING

| Ī | | | | PRESSUF | RE COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|---|---|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 1. | 00 α = 3.9° | | | | | | |
| UPPER SURFACE | 10.500 | 003141 - 003748 - 007788 - 10188 - 116639 - 116639 - 12639 - 1 | | - 221 - 804 - 759 - 296 - 3296 - 3296 - 3286 - 3766 - 3766 - 409 - 409 - 409 - 426 - 328 - 328 - 328 - 3296 - 3576 - | 456 768 813 441 441 355 3785 3785 3785 4464 490 460 460 480 281 186 186 | .073 773 817 686 596 3706 4372 463 452 463 503 558 558 558 558 558 372 | . 053 - 821 - 854 - 765 - 735 - 735 - 449 - 458 - 553 - 6639 - 689 - 250 - 136 - 092 - 092 | |
| LOWER SURFACE | 1.25 2.50 5.50 10.00 20.00 35.00 35.00 45.00 45.00 45.00 60.00 65.00 75.00 85.00 | .385 .368 .3281 .2547 .2547 .1516 .0959 .0049 .0024 0049 0089 0089 0089 1004 1300 1300 1300 | .372 .307 .242 .195 .169 .079 .074 .016 038 074 105 1136 136 136 136 136 141 | .340 .271 .199 .153 .121 .077 .029 093 063 063 093 147 157 160 169 189 199 059 | .292 .240 .213 .149 .1033 0039 066 098 124 154 173 185 200 200 | .295 .219 .136 .083 .041 -0378 -1057 -1134 -162 -212 -212 -212 -2174 -174 -1179 -1075 -1075 -1075 -1075 | . 225 . 127 . 0740 . 0007 - 0057 1253 1644 1893 2755 2833 275 2915 1915 101 | 281 .2034 .10516 .0078 1236 2324 2324 3336 3550 3550 35797 1408 1036 |
| | M = 1 | <u> </u> | | | 1 | | | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 15.00 30.00 35.00 40.00 55.00 60.00 75.00 80.00 80.00 | - \$343 - \$375 - \$371 - \$360 | - \$390 - \$314 - \$174 | - :030 - :975 - :898 - :809 - :755 - :383 - :349 - :379 - :379 - :443 - :443 - :445 - :477 - :477 - :477 - :477 - :477 - :275 | - \$154 | 154 950 984 927 8860 804 7437 449 523 5547 612 602 2914 2914 2914 2914 2914 | - 187 - 1961 - 1991 - 1891 - 18826 - 1 | - 1 00151 - 1 00 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 25.00 30.00 35.00 40.00 55.00 60.00 75.00 85.00 95.00 | | 2555 1188 1143 1080 2045 2045 - 2014 - 2055 - 2069 - 2086 - 2086 - 2086 - 2087 - 2086 | - 105 - 109 - 109 | 7395 7325 7299 7249 72179 72174 70114 7020 7020 7020 7077 7116 7116 7131 7131 7131 7131 7131 71 | .396 .325 .241 .1144 .1144 .1049 .010 057 1103 1124 121 116 099 103 099 103 | 73340 73340 7114954 7114954 7114957 71169557 71169557 7116957 716957 716957 716957 716957 716957 716957 716957 716957 | # 12727360211779229926111 # 512100773899135544226371 # 5 2 2 2 2 3 3 3 3 3 3 2 2 2 1 1 1 1 1 1 1 |

TABLE I

BASIC WING

| ſ | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|---|---|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| 1 | M = 1. | $\alpha = 7.8^{\circ}$ | | - 1 | | I | 1 | |
| UPPER SURFACE | 1.5500 70.000 1.5500 1.50000 1.50000 1.50000 1.50000 1.50000 1.50000 1.50000 1.50000 1 | 0 0 2 6 3 3 4 3 6 6 6 6 7 6 7 6 7 6 7 6 7 6 7 6 7 6 7 | 383 382 395 423 423 423 | - 136 - 1.076 - 1.025 - 1.025 9627 8307 4634 437 475 475 506 506 506 506 205 205 | . 168 - 1.042 - 1.079 - 1.031984943915858842552455245745765653952666134 | 370 - 1.066 - 1.082 - 1.035 978 918 918 918 985 837 780 6435 6435 627 | 873 893 | - 1.1099 - 1.0995 - 1.00125 - 1.00125 - 1.997637 967637 86682 867799 6663293 6663293 6663293 6663293 6663293 |
| LOWER SURFACE | 1 . 25 2 . 50 5 . 50 10 . 000 20 . 000 30 . 000 40 . 000 40 . 000 45 . 000 65 . 000 75 . 000 85 . 000 95 . 000 | . 463 . 494 . 471 . 4418 . 367 . 301 . 291 . 228 | .552 .483 .416 .360 .3252 .2143 .1352 .076 .0012 0017 0029 0017 0031 .0079 0089 | .505 .441 .361 .3217 .220 .162 .082 .0318 018 043 043 059 087 087 073 064 | .463 .394 .348 .348 .245 .143 .140 .073 .0053 054 076 078 078 078 075 075 075 075 | .450 .388 .306 .2512 .162 .1162 .1072 .0388 0196 067 086 0942 0942 097 1014 126 | .395 .302 .2501 .1584 .1062 .0315 0486 1288 1267 2022 2022 2044 2022 2044 2022 | .435 .371 .288 .2128 .1600 .0051 1293 243 243 344 344 344 344 344 344 344 |
| | M = 1 | .00 $\alpha = 11.5^{\circ}$ | · · · · · · · · · · · · · · · · · · · | · · · · · · · · · · · · · · · · · · · | | · · · · · · · · · · · · · · · · · · · | 1 | |
| UPPER SURFACE | 25.00 30.00 45.00 45.00 55.00 60.00 75.00 85.00 85.00 90.00 | - 164 - 296 - 4475 - 488 - 450 - 4150 - 4133 - 4146 - 415 - 415 - 416 - 415 - 493 - | - 1.103 - 1.090 - 1.115 - 1.084 884 886 826 432 443 463 463 514 514 533 533 525 454 525 | - 1.141 - 1.0926 - 1.1226 - 1.0377 9703 8302 7755 7411 7256 7083 7008 4842 4048 | - 1.227 - 1.240 - 1.203 - 1.141 - 1.090 - 1.031 - 1.995 - 8839 - 8839 - 887 - 800 - 774 - 724 - 724 - 724 - 724 - 724 | - 1.268 - 1.267 - 1.238 - 1.238 - 1.267 - 1.184 - 1.156 - 1.106 - 1.093 - 1.081 - 1.093 - 2.081 - 1.077 - 777 - 777 - 777 - 7780 - 759 - 629 | - 1.210 - 1.231 - 1.166 - 1.185 - 1.157 - 1.080 - 1.016 965 895 840 777 754 | - 1.08 - 1.07 - 1.05 - 1.01 98 93 93 90 87 78 73 73 76 68 |
| LOWER SURFACE | 1.25 2.50 5.50 10.000 10.000 20.000 30.000 45.000 45.000 65.000 65.000 75.000 85.000 95.000 | .615 .579 .520 .4416 .3152 .279 .2208 .171 .1303 .0357 .0357 | .297 .2260 .224 .192 .156 .123 .1084 .064 .045 | .180 .148 .121 .090 .073 .058 .031 017 017 | .570 .508 .477 .428 .364 .311 .265 .226 .154 .121 .087 .018 .004 .018 .004 | .516 .490 .426 .375 .338 .287 .189 .154 .119 .087 .053 .023 .024 043 043 | .485 .409 .360 .313 .267 .210 .162 .017 .072 .031 014 107 146 176 226 226 226 2355 297 | |

TABLE I

BASIC WING

| Ī | | | | PRESSU | RE COEFFICIENT, | P, AT: | | |
|---------------|--|---|---|--|---|--|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.0 | $\alpha = 15.9^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 115.00 205.00 300.00 400.00 400.00 600.00 800.00 800.00 900.00 | 28 56 60 - 46 69 97 - 46 69 97 - 46 69 97 - 55 55 55 80 00 77 - 55 65 56 54 57 - 55 65 65 65 57 - 55 65 65 65 7 - 55 65 65 65 7 - 55 65 65 | 788 - 1.207 - 1.189 - 1.202 - 1.137 - 1.137 - 1.1105 - 1.1105 - 1.1092 - 1.0332 - 1.5645 5665 - | 879 - 1.138 - 1.074 - 1.195 - 1.122 - 1.124 - 1.1099 - 1.0981 - 1.067 - 1.066 - 1.064 - 1.0166 - 1.064 - 1.0941 - 1.0941 - 1.0941 | - 1.077 - 1.077 - 1.038 947 941 852 831 808 | - 1.0039753975396909640943392048850819477371268186753 | 835 7848 7542 7553 731 7113 66799 6645 6634 6618 6618 6613 66095 6699 | 40 80 22 1 |
| LOWER SURFACE | 1.25 2.50 5.50 10.00 10.00 25.00 35.00 35.00 45.00 45.00 65.00 70.00 85.00 85.00 90.00 | 305 | .397 .359 .328 | .588 .621 .559 .557 .476 .423 .342 .305 .206 .188 .218 .218 .218 .218 .218 .218 .218 | . 556 . 521 . 480 . 427 . 384 . 341 . 307 . 267 . 269 . 193 . 158 | .522 | .540 .500 .461 .418 .377 .320 .274 .229 .180 .038 -0087 -052 -052 -087 -145 -204 -241 -228 -302 | 482 494 494 3841 23583 10833 10833 10932 11700 1000 10 |
| | M = 1 | $.03 \alpha = 0.0^{\circ}$ | | | | | 1 | |
| UPPER SURFACE | 1.25 2.50 7.50 10.00 20.00 25.00 35.00 40.00 55.00 60.00 70.00 85.00 90.00 | .240 .163 .1183 .065 .035 .035 .035 .044 .044 .060 .1140 .1140 .1140 .179 .1154 .1154 .1207 .214 | 099 124 149 167 194 219 224 205 | 034 029 0361 0709 1166 139 1259 2533 2559 263 | 042 066 080 078 147 178 129 221 253 274 290 300 167 167 100 | 017 086 059 1306 137 147 210 231 250 274 345 314 345 316 164 102 057 | .015 133 079 115 139 202 224 252 274 299 328 361 423 423 423 | 370 395 411 432 408 271 189 |
| LOWER SURFACE | 1.25 2.50 57.50 10.00 25.00 25.00 30.00 45.00 45.00 65.00 65.00 65.00 95.00 | .207 .164 .095 .069 .024 .027 .007 .073 | 048017021028037069074114157183233233 | 060 087 135 178 197 1921 246 276 276 285 297 297 | 106 052 091 157 178 2133 268 325 325 331 326 327 3 | 137 140 163 178 206 233 258 272 293 348 367 344 159 109 | - 163 - 177 - 195 - 2354 - 282 - 3309 - 3364 - 3364 - 432 - 4432 - 4432 - 1419 - 181 - 181 - 083 | - 189 - 216 - 2736 - 3145 - 361 - 3684 - 3684 - 4133 - 4619 - 306 - 1023 - 1023 - 1046 |

TABLE I

BASIC WING

| Ī | DEDGERAM | _ | <u> </u> | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|---|---|---|--|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1. | $\alpha = 3.9^{\circ}$ | | | | | | |
| UPPER SURFACE | 125000000000000000000000000000000000000 | | | 250 -7496 -3774 -3774 -3774 -32878 -32878 -3287 -33490 -3340 -3340 -3340 -33400 -33400 -33400 -33400 -33400 -33400 -33400 -33400 -33400 | . 497 698 750 681 529 312 342 345 345 345 403 443 443 443 444 447 447 448 | .116 703 746 684 636 353 3545 386 396 413 446 471 505 520 506 | . 1052 7578 66958 66466 66466 6443 4447 4493 55751 | . 20896 |
| LOWER SURFACE | 1.25 8.50 7.50 10.000 25.00 35.00 45.00 55.00 65.00 65.00 65.00 75.00 85.00 | .3305 .2862 .2385 .1455 .0959 .0049 .0015 .0061 .0072 .0072 .1079 .1015 | .361 .299 .233 .194 .167 .111 .0850 .0107 .0087 0867 109 109 123 123 111 | .344 .276 .204 .161 .088 .043 .010 .025 .075 .135 .135 .135 .135 .135 .135 .0856 .081 | .300 .253 .2531 .1728 .053 .0265 0865 0865 1145 1146 1146 1150 | .329 .2576 .1236 .0857 .0127 .0127 .00563 .1067 .11553 .1144 .1147 .0875 .00566 .0050 | .272 .174 .1237 .0845 .0045 0041 1076 1127 1144 1734 2236 225 1640 1640 1640 | .307 .2352 .0775 .0352 1207 2267 3011 3299 33161 32990 33161 3367 1363 13 |
| | M = 1 | .03 $\alpha = 5.9^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 15.00 20.00 25.00 40.00 45.00 55.00 66.00 75.00 85.00 85.00 | - 1180 - 22120 - 22120 - 22236478 - 2234359 - 223189 - 233339 - 33336 | 276 | .074 - 896 - 8875 - 829 - 7409 - 428 - 3346 - 3346 - 3358 - 4167 - 428 - 4364 - 3833 - 4167 - 4384 - 3833 - 3795 | .378 - :8612 - :859 - :7393 - :7092 - :3774 - :3795 - :442 - :4493 - :4499 - :4499 - :4731 - :1100 | 1058749068558155731873187318731844674967496749674967496749674967496749674967496749674967496749674967496749675555 | - #134 - #886 - #832 - #8306 - #7753 - #77352 - #77314 - #5773 - #5 | - 1040 - 19921 - 19921 - 19821 - 19821 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 25.00 40.00 45.00 55.00 60.00 85.00 75.00 80.00 85.00 | .316 .34189 .3742 .3297 .2238 .2277 .21365 .01369 .0006 0069 0069 0069 | . 481 . 414 . 346 . 294 . 262 . 199 . 129 . 095 . 070 . 047 . 012 - 014 - 058 - 063 - 063 - 056 - 058 | 4582 3033 2256 11767 20926 200523 - 00420 - 00450 - 00642 - 0074 - 0074 - 0053 - 0043 | 417 3533 2646 11446 1070 0042 00197 - 00188 - 0058 - 00895 - 0077 - 0058 - 0058 | . 426 .354 .270 .216 .175 .139 .050 .019 .019 .0019 .0019 .0071 .097 .0986 .066 .066 .0650 .0659 | .358 .265 .2174 .1274 .035 .0032 .0032 .0032 .0032 .0091 .1302 .1302 .1303 .1402 .1602 .1603 .1603 .1603 | 4413508 |

TABLE I

BASIC WING

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|--|---|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 1. | $03 \qquad \alpha = 7.9^{\circ}$ | | | | | | |
| UPPER SURFACE | 000 12.50 57.50 10.00 12.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10.00 10 | | 019 - 1.007229221808453584344934593769377940224022402739773164 | 094 - 1.0349919819818606834563993994274614614654734654691626 | . 211 - 1.0043 9448 9910 8478 8479 5767 4807 487 513 462 513 513 513 513 513 513 513 513 513 513 513 513 513 513 513 513 514 - | | 368 - 1.0056 9766 9789 9821 8642 8666 8667 8666 8667 8663 5543 5443 5443 336 | - 1.0566 - 1.05818 99635 99635 99635 996315 88224 88351 85129 85129 66424 66421 |
| LOWER SURFACE | 10.500 57.500 10.000 205.000 305.000 45.000 45.000 45.000 655.000 755.000 850.000 850.000 95.000 | . 4 09 4 484 4 489 4 489 . 329 . 318 . 2563 . 196 . 1635 . 0069 . 0064 . 0024 . 0024 . 0036 | .508 .508 .4363 .3845 .2446 .1742 .1144 .0851 .0319 .0019 .0020 .0020 .0020 .0023 .0042 | . 534 . 5469 . 3347 . 3160 . 2080 . 11343 . 1072 . 0012 . 0014 . 0036 . 0036 . 0031 . 0020 | 495 423 3331 2825 1843 11163 00568 1023 1033 1034 1034 1034 1032 1033 1033 1033 1033 1033 1033 1033 | . 4 9 0 7 . 3 9 5 6 6 7 . 3 9 5 6 6 7 . 3 9 5 6 0 7 . 3 9 5 6 0 7 . 3 9 5 6 0 7 . 3 9 5 6 7 . 3 9 6 7 6 7 6 7 6 7 6 7 6 7 6 7 6 7 6 7 6 | . 428 .3419 .245 .1499 .1462 .0701 .0000 040 080 1177 167 | . 465 .4009 .246 .198 .109 .014 081 1460 1224 2582 2982 3004 304 304 2302 2302 |
| | M = 1 | .03 $\alpha = 11.5^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 7.500 10.00 20.00 30.00 30.00 40.00 50.00 65.00 65.00 65.00 65.00 65.00 95.00 | - 238 - 389 - 4435 - 4440 - 3981 - 3581 - 3583 - 363 - 363 - 363 - 4087 - 4408 - 447 - 447 - 443 | 402 | | 026 - 1.148 - 1.157 - 1.123 - 1.093 - 1.016958887887718774753718718718718718718718 | - 1.147 - 1.115 - 1.096 - 1.065 - 1.020 - 1.002 - 1.002 - 994 984 984 733 733 732 715 728 707 669 601 | 741 - 1.099 - 1.1069 - 1.0868 - 1.047 - 1.09761 9873 7208 7208 7208 66450 66450 5591 559 | 678 - 1.028 - 1.012 3862 9410 9410 9436 7536 7345 7345 7345 7345 7556 7345 7556 7345 55682 56082 56082 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 25.000 35.000 40.000 45.000 665.000 675.000 855.000 855.000 955.000 | . 5628 . 6450 . 6450 . 5526 . 4488 . 3338 . 31338 . 2753 . 2436 . 1768 . 1372 . 082 | .634 .575 .525 .482 .374 .331 .2530 .191 | .157 | .595 .596 .504 .449 .340 .340 .322 .186 .153 .118 .089 .067 .053 .039 | .542 .516 .453 .403 .366 .314 .222 .186 .153 .121 .087 .061 .001 .011 .004 .028 .028 .029 .011 .004 | .512 .438 .391 .338 .2945 .1156 .1151 .023 -0642 -129 -1787 -2252 .289 | 4798787899995787898989899995799999999999 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| Ī | | | | PRESSURI | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|---|---|--|--|--|---|
| | PERCENT CHORD | 0.135ь/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | $60 \alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 00 1.50 2.50 5.00 1.50 2.50 2.50 2.50 2.50 2.50 2.50 2.50 2 | 2 3 3 3 5 4 3 5 4 5 6 5 6 5 6 6 6 6 6 6 6 6 6 6 6 6 6 | . 476 080 066 0675 084 087 0887 0881 0687 0681 0639 076 1842 2630 272 2630 297 | . 451 - 0227 - 0341 - 0554 - 0554 - 0051 - 0042 - 0032 - 0038 - 1033 - 225 - 312 - 312 - 312 - 343 - 345 - 4451 - 4414 | . 687 . 039 . 016 . 004 . 0010 . 005 - 011 - 009 - 002 . 010 . 033 . 054 . 094 . 151 . 2310 . 317 . 317 . 317 . 317 . 317 . 316 . 336 . 336 . 336 . 336 . 339 | . 437 .171 .077 .061 .033 .015 .019 .009 .019 .042 .102 .102 .240 .3154 .269 .267 .268 .267 .267 | . 435 .185 .085 .085 .005 .0014 .0119 .0019 .0019 .0034 .1254 .2248 .2248 .274 .2488 .1744 .1538 | 3 3 0 1 5 4 1 0 3 0 0 1 6 - 0 3 0 - 0 4 5 - 0 3 3 0 - 0 4 2 - 0 4 2 - 0 4 2 - 0 3 4 - 0 3 3 - 0 4 2 - 0 3 5 - 0 3 5 - 0 4 5 - 0 3 6 - 0 4 7 - 0 6 8 - 0 7 9 |
| LOWER SURFACE | 1.25 2.50 7.500 10.000 15.000 20.000 30.000 40.000 45.000 55.000 66.000 65.000 70.000 75.000 90.000 95.000 | . 126 . 068 . 068 . 0133 . 0069 . 0074 . 1235 . 1416 . 1777 . 1779 . 178 . 165 . 159 . 159 . 139 . 109 . 109 | 1270940911051111301461521641761781801801521081091072003 | - 188 - 135 - 120 - 133 - 139 - 158 - 164 - 167 - 179 - 189 - 191 - 191 - 155 - 157 - 124 - 142 - 142 - 1457 - 167 | - 236 - 172 - 106 - 133 - 149 - 174 - 176 - 183 - 184 - 197 - 206 - 208 - 208 - 196 - 186 - 177 - 158 - 158 - 158 - 178 | 364 255 218 2104 1912 2123 2113 216 2225 2225 225 225 214 169 164 134 176 | 448 2955 2353 2167 2167 21122 2218 2114 1978 1200 1227 1221 | 467 324 252 248 216 275 171 175 153 154 117 017 008 - |
| | M = 0 | $\alpha = 4.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 20.00 30.00 35.00 40.00 55.00 60.00 65.00 65.00 75.00 80.00 80.00 90.00 | 231 230 209 209 171 094 008 . 201 1390 1779 362 1362 045 | 764 889 837 693 473 | 071014066134152160398419428405 | 07192034139124001651240016580457188931403530 | 8567 8567 87668 27669 2069 11067 0146 117880 117880 11880 224483 224483 22437 22437 22437 | 23886654431326619491100870470470491155164125144136 | 684 456 3100 2229 1641 0188 0073 0537 0244 026 027 028 027 0336 0336 0342 0336 |
| LOWER SURFACE | 55.00 | .2722 .2729 .11796 .11552 .0652 .01672 00452 00452 00452 00893 00893 00893 | . 264 . 140 . 140 . 107 . 062 . 030 . 023 058 079 093 099 099 099 096 056 056 056 | 059 074 086 102 106 | .331 .257 .191 .140 .099 .043 .007 027 048 071 128 129 134 139 134 139 134 | .341 .257 .168 .116 .075 .041 .003 038 076 076 126 136 136 134 132 134 132 | .330 .246 .147 .107 .090 .028 049 074 1127 1127 134 142 133 080 088 097 096 | 107 121 1193 123 1193 1019 094 065 064 |

TABLE II

| | program | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|--|---|---|---|--|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85ъ/2 | 0.95b/2 |
| | M = 0 | .60 α = 6.0° | | | | | | |
| UPPER SURFACE | 25.5000 25.500 | 138 337 404 3934 3455 3265 3276 263 263 2763 | 852 - 1 .463 912 676 5708 422 370 289 248 126 065 005 | 743 877 807 723 6270 3699 3048 1967 0805 137 0805 135 3806 3806 4119 392 | 331 758 716 558 588 582 433 352 227 177 126 066 011 .046 015 .110 .130 314 331 331 331 | - 1.080 - 1.023 903 595 497 357 278 165 1166 014 084 1820 1350 259 2670 2661 | 834 817 6550 4969 444 3572 201 153 097 004 116 116 116 126 1346 1346 1346 | 7327 9288 53584 43535 43535 13087 13087 0064 00440 00440 00440 004 |
| LOWER SURFACE | 1.25 2.500 7.500 10.00 15.00 25.00 25.00 35.00 45.00 55.00 65.00 65.00 85.00 95.00 | 364026 340763 340763 324047 344047 34 | . 450 .3894 .2043 .1407 .0704 .011 .02417 .0050 .04477 .0050 .0477 .0050 | 44768 44768 2888 21996 11996 003097 00554 00057 00057 00067 | . 426 . 263 . 296 . 248 . 197 . 1389 . 049 . 0026 . 0027 . 0051 . 0087 . 0087 . 108 . 1002 . 1002 . 1106 . 1146 | . 440 .3288 .3281 .1891 .0412 .0412 .0413 .0413 .0413 .0413 .0915 .0999 .0999 .1034 .1044 .134 | . 4370 . 2598 . 1924 . 10297 . 10297 . 10297 . 10379 . 11113 . 1108 . 11113 . 10664 . 10822 . 10922 . 10922 | 30354 30054 10714 107577 100097 1000984 10009884 10009884 100066642 10009884 1000988 10009884 1000988 1000988 1000988 100098 10 |
| | M = 0. | 60 α = 8.0° | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 15.00 20.00 30.00 40.00 40.00 55.00 60.00 65.00 75.00 80.00 80.00 90.00 | - 43197 - 3357 - 33537 - 3286 - 187 - 078 - 1166 - 1893 - 1935 - 2354 - 160 | 696 566 431 | - 931 - 8897 - 76339 - 6379 - 6379 - 6379 - 6379 - 6379 - 6667 - 6667 - 6667 - 6730 - 6667 - 6730 - 67300 - 67 | 324 280 253 | 291 263 239 | 96501 66501 5661964 5514737 4887497692 4887497692 1222 007369 1222 007369 12637 12637 12637 12637 12637 | 105 095 099 089 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 25.00 35.00 40.00 55.00 65.00 65.00 75.00 80.00 95.00 | .070 .042 .022 .005 006 023 012 | .484 .4444 .370 .321 .278 .218 .174 .130 .003 .003 .003 .003 .003 .003 .003 | .087 .060 .027 | 473 434 383 326 279 203 1133 074 0054 002 0133 - 0452 - 065 - 065 - 065 - 065 - 065 - 065 | 461 436 365 304 265 211 1110 0080 0080 0080 - 0040 - 0040 - 0049 - 0056 - 0070 - 0056 - 104 | . 458 . 428 . 340 . 278 . 268 . 1829 . 0747 . 0019 - 046 - 0602 - 083 - 072 - 072 - 072 - 079 - 079 | . 431 . 285 . 2038 . 0014 0038 0521 068 0663 0664 0644 0448 0448 0448 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|---|---|---|--|---|
| _ | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85ъ/2 | 0.95b/2 |
| | M = 0. | $60 \alpha = 11.3^{\circ}$ | | | , | | | |
| UPPER SURFACE | 1.00 1.250 2.500 5.000 1.50.000 2.50.000 2.50.000 4.50.000 5.50.000 6.50.000 7.50.000 8.50.000 9.50.000 | - 0 34 - 1 0 634 - 1 0 780 - 0 8657 - 0 657 - 0 657 | - 1.770 - 1.392 - 1.423 - 1.4533 - 1.6402 - 1.3965 - 1.558 - 1.3965 - 1.0658 - 1.334 - 1.117 - 0074 - 0678867886824390 | 1.529 1.1079 1.079 1.079 1.0740 1.0740 1.0776 1.07776 1.07 | - 1 . 12 4 | | - 470 - 439 - 434 - 425 - 420 - 420 - 3977 - 3760 - 3453 - 3219 - 2874 - 2274 - 2274 - 250 - 245 | 4127 4127 4127 4127 4127 4127 4127 4127 |
| LOWER SURFACE | 1.25 2.5.00 7.5.00 10.00 15.00 20.00 25.00 40.00 45.00 605.00 70.00 85.00 90.00 90.00 | 3682 3334 4463 4463 3385 2374 1128 2277 0083 0070 0033 0039 0013 | .487 .529 .4406 .44087 .3287 .2207 | .461 .462 .467 .425 .232 .276 .232 .137 .131 .070 .013 .0213 .0013 .0013 | 501 502 429 382 221 185 142 091 045 021 - 019 - 025 - 091 | .476 .498 .3948 .2840 .1942 .1151 .00552 .0015 .0013 .00797 .00797 | . 477 . 477 . 4031 . 3350 . 194 . 1400 . 0628 . 0030 . 00527 . 0073 . 0612 . 0082 . 00 | .411 .3710 .3107 .1760 .0462 .00141 00566 0071 0084 0087 0087 0087 0087 0087 |
| | M = (| 1 | 1 | | | | | |
| UPPER SURFACE | 10.00 1.25 2.50 7.50 10.00 20.00 20.00 30.00 40.00 45.00 60.00 60.00 60.00 80.00 80.00 90.00 | 681 580 5364 472 2665 2731 731 4968 | 198 | - 1.203 - 1.167 - 1.184 - 1.182 - 1.163 - 1.1666 - 1.157 - 1.1666 - 1.129 - 1.013 - 1.0156 - 1.129 - 1.073 - 1.0156841966556565 | 673 6505 623 5514 551 543 529 | 435 434 429 418 404 | 462 448 443 451 452 452 442 442 442 421 403 403 403 403 303 319 | 4573943987388338833867355132103200320032003200 |
| LOWER SURFACE | 40.00 45.00 50.00 55.00 | .317 .6133 .530 .5693 .459 .459 .459 .431 .330 .236 .236 .195 .1106 .106 | .556 .456 .416 .362 .2857 .2857 .2229 .1169 .1148 .1148 .1087 .0577 | .396 .529 .510 .480 .371 .324 .286 .213 .1180 .151 .127 .108 .076 | .211 .173 .135 .070 .039 .014 062 076 104 | .502 .477 .436 .401 .343 .289 .242 .202 .158 .121 .0017 017 018 | .430 .436 .436 .392 .377 .297 .235 .184 .140 .092 .051 .008 074 093 074 093 112 1475 1475 1475 | 072 |

TABLE II

| | éspectary | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|---|---|--|---|--|---|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85ь/2 | 0.95b/2 |
| | M = 0.8 | $\delta \alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 25.50 25.50 10.50 1 | .177 .209 .1033 .0612 .0027 .0050 0050 0086 0 | .512 068 0658 0671 087 0890 0990 0980 0563 0554 22779 7647 7715 7715 | . 478 041 033 039 051 051 053 019 003 033 156 253 325 3345 397 421 4210 416 | 7 4 9 - 0005 - 0015 - 0015 - 0017 - 0037 - 1124 - 2347 - 3339 - 3226 - 3226 - 3226 | .476 .1666 .0545 .0009 .001117 .00585 .0098 .01127 .00586 | . 465 . 2087 . 0087 . 00962 . 00418 . 002262 . 004615 . 0 | 3672622458484198293771107 |
| LOWER SURFACE | 25 25 25 25 25 25 25 25 25 25 25 25 25 2 | 179 1198 0023 0040 0046 0064 1367 1472 1204 12233 12211 12211 12211 12211 1236 1237 1367 1367 | 107 075 076 1095 1247 1643 2008 2249 2437 2319 180 155 1103 107 | - 168 - 121 - 1345 - 1469 - 12018 - 12018 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 - 12028 | - 219 - 183 - 110 - 114 - 170 - 211 - 221 - 2231 - 242 - 253 - 265 - 260 - 225 - 241 - 221 - 221 - 221 - 221 | - 370 - 229 - 2333 - 2180 - 2457 - 2770 - 2770 - 2770 - 2773 - 2855 - 2731 - 2057 - 1148 - 205 | - 478 - 3279 - 22712 - 22506 - 2278 - 28806 - 22534 - 22534 - 22534 - 22534 - 2377 - 1328 - 13375 | 560 4103 31213 2251 2251 22079 20079 12104 11046 11046 0776 |
| | M = 0 | $0.85 \alpha = 4.0^{\circ}$ | | | | | T | r |
| UPPER SURFACE | 25.000 25.000 25.000 25.000 25.000 25.000 25.000 45.000 45.000 65.000 65.000 75.000 85.000 85.000 | - 182 - 071 - 050 - 250 - 457 - 1 038 - 743 - 621 - 407 - 241 | 290 2359 1184 016 093 .1735 .0693 721 7711 633 7311 | 437 | . 2 51 - 1.006 812 358 358 2111 173 083 035 035 035 035 035 328 328 328 3329 | 917 784 304 304 224 1495 1035 0055 001 . 1165 . 1165 . 1296 2247 2481 2481 252 | 040 .009 .062 .111 .145 .151 .133 164 150 156 | 04367 826749 33717 1133661 107596 004419 004419 004485 00485 00563 |
| LOWER SURFACE | 1.25 2.500 70.000 10.000 25.000 350.000 450.000 605.000 605.000 800.000 800.000 | .3059 .2110 .1380 .0830 .0283 .0283 .0283 .0041 .0785 .0955 .1055 .1177 .1177 .1177 | .264 .193 .142 .113 .066 .031027072076115121123107107107107 | . 269 - 182 - 196 - 098 - 098 - 098 - 098 - 115 - 1135 - 1135 - 1135 - 1144 - 1164 - 1 | .259 .143 .097 .040 001 001 085 107 131 147 163 | .261 .168 .1075 .075 .041 .012 .046 .075 .101 .121 .148 .178 .178 .178 .178 .178 | . 258 . 147 . 096 . 095 . 093 . 033 . 0372 . 102 . 1184 . 1184 . 1194 . 1177 . 1281 . 1204 . 11204 | .1905 .027 025 1335 1463 1552 1552 1263 1105 1066 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| ſ | DUDGENE | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|---|---|---|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| \neg | M = 0.0 | 85 α = 5.9° | | | | | | |
| UPPER SURFACE | 25 25 25 25 25 25 25 25 25 25 25 25 25 2 | . 206 139 2521 3286 3378 3271 3316 3317 3112 2472 2474 1012 2474 1012 6443 2495 | 420 | 275 - 1 .1348955665048223700239313750148080304441444194439413 | 665 607 551 473 386 315 194 194 091 074 085 3347 357 350 338 | - 1.0469 7326 63662 4538 4538 2490 1394 003197 0157 09158 2995 2989 2889 | 444 742 628 574 499 435 288 285 1054 0038 0038 0051 0051 0085 1836 149 | 368821726683944751515095400490043100570758077807880788 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 10.00 15.00 25.00 35.00 45.00 55.00 65.00 65.00 70.00 75.00 85.00 | .3773 .3377 .263 .263 .154 .098 .068 | 063 071 077 084 077 086 109 | . 436 .360 .276 .2276 .2286 .136 .048 .013 -011 -037 -060 -081 -093 -100 -1130 -1130 -1130 -1130 -1162 -1162 | . 420 . 344 . 292 . 241 . 193 . 116 . 074 . 034 . 008 . 023 . 051 . 077 . 096 . 115 . 122 . 130 . 147 . 1440 . 1442 . 157 . 176 | .429 .359 .2644 .262 .1172 .0228 0406 0922 1120 1357 1357 1341 1445 1059 | . 428 .355 .247 .189 .0939 .0034 .038 .0718 .1156 .1156 .1167 .1634 .1634 .102 | . 386 .303 .207 .1289 .0312 .1069 .1103 .1284 .1134 .1134 .1136 .1098 .0993 .068 .068 .0646 .0041 |
| | M = 0 | .85 $\alpha = 8.0^{\circ}$ | | | | | | |
| UPPER SURFACE | . 00 1.25 2.50 7.50 10.00 20.00 30.00 30.00 40.00 55.00 65.00 65.00 80.00 80.00 90.00 | 407 500 522 484 452 4452 431 319 1361 13 | - 1.388 - 1.243 8423 6407 6301 4243 2437 068 .0157 016 .023 0755 716 727 6827 529 | 956 8579 7733 6689 5186 3899 3209 259 2259 1188 1122 1162 4451 4451 4451 4451 | | 943262 8767853635436280354362803226699016292331872257 | 420 378 324 271 225 186 178 | 785 825 805 805 768 7624 53072 16372 1635 1440 1447 1447 1443 1286 1296 1296 1296 1296 1296 1496 1443 |
| LOWER SURFACE | 40.00 45.00 50.00 55.00 | . 434 .394 .356 .311 .228 .170 .107 .001 .001 .001 | . 452 .326 .3286 .218 .174 .174 .104 .069 .044 .017 .008 .017 .008 .017 .017 .018 .019 .019 .019 .019 .019 .019 .019 .019 | .114 .079 .055 .024 001 028 041 048 059 107 107 | . 367 . 314 . 268 . 1191 . 142 . 102 . 071 . 036 . 026 . 026 . 063 . 063 . 089 | . 428 .2847 .2487 .2483 .1138 .1060 .0032 .0037 0681 081 09966 | . 428 .334 .277 .267 .179 .075 .033 .010 .045 .079 .105 .128 .128 .128 .128 | .375 .285 .210 .143 .051 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| Γ | | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | |
|---------------|--|---|--|--|---|---|--|--|--|--|--|
| ł | PERCENT CHORD | 0.135ь/2 | 0.25b/2 | 0.40ь/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | | | |
| | M = 0.8 | $\alpha = 11.3^{\circ}$ | | | | | | | | | |
| UPPER SURFACE | 125000 125000 105000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 1050000 10500000 1050000000 10500000000 | . 170 - 450 - 614 - 735 - 749 - 723 - 621 - 621 - 598 - 582 - 361 - 1365 - 2965 - 7991 - 6554 - 554 | - 1.4466 - 1.4486 - 1.4486 - 1.4470 - 1.3450 - 1.3450 - 1.3450 - 1.350 - 1.350 | 990 1.1350 1.1350 1.1080 1.10854 9.6620 1.08550 1.88550 1.88550 1.88550 1.8668 1.762777 1.55777 1.55473 1.4543 1.3811 | 5375 -9770 -9563 -9563 -9544 -99347 - | 7 3 5 8 4 0 2 6 6 5 5 4 0 4 6 6 5 9 9 1 9 1 9 1 9 1 9 1 9 1 9 1 9 1 9 1 | - 459 - 422 - 4182 - 4118 - 4124 - 33777 - 33661 - 33661 - 33661 - 33728 - 33681 - 33728 - 33681 - 33728 - 337 | 45599531455959595959595959595959595959595959595 | | | |
| LOWER SURFACE | 1.25 2.500 7.500 15.000 25.000 35.000 45.000 45.000 55.000 60.000 75.000 80.000 90.000 | 4 11 4 4 8 8 4 5 5 16 4 6 6 12 5 5 0 6 12 5 5 0 6 12 5 5 0 6 12 5 13 13 10 6 12 6 12 6 13 6 13 6 13 6 13 6 13 6 13 6 13 6 13 | .568 .4919 .4379 .3392 .2445 .2209 .1179 .1144 .0065 .0035 .0035 .0035 .0035 .0035 .0035 .0035 | .515 .5159 .4150 .4178 .3188 .2215 .1320 .0922 .00477 .0077 .0077 | . 522 495 466 416 .369 .305 .251 .178 .141 .109 .077 .048 .009 .009 .051 .042 .051 .090 .137 | . 498 .489 .434 .386 .288 .288 .146 .113 .079 .014 .014 034 054 054 054 140 140 140 223 | . 489 . 479 . 3409 . 3437 . 253 . 141 . 092 . 0038 . 0770 . 11286 . 1134 . 168 . 134 . 169 . 191 | .441 .4029 .2329 .2577 .0094 .00329 .11310 11421 11542 11542 11542 11543 11543 11543 11543 | | | |
| | M = 0 | .85 $\alpha = 15.5^{\circ}$ | | | | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 20.00 30.00 30.00 45.00 45.00 50.00 60.00 60.00 70.00 80.00 90.00 | . 055 - 640 - 1040 - 1065 - 1040 - 1065 - 1040 - 1040 - 1040 - 1040 - 866 - 805 - 717 - 628 - 717 - 628 - 503 - 420 - 318 - 1040 - 1040 | 591 457 371 | - 1.085 - 1.0035 - 1.0043 - 1.042 - 1.048 - 1.0369919799578638638637755708663 | 664 646 645 601 504 556 556 576 555 576 555 | 559556056695733567355625548554855275241473047304730 | 418 406 387 | 454 454 456 456 429 4234 4234 4234 4234 4234 4234 4311 4013 3571 3334 3334 | | | |
| LOWER SURFACE | 30.00 35.00 40.00 45.00 50.00 | .476 .599 .698 .650 .597 .512 .478 .424 .3298 .233 .273 .1145 .088 .088 .088 | .575 .6142 .5946 .5449 .4494 .4358 .3188 .32743 .2774 .1299 .0745 .0152 | .500 .5539 .504 .472 .423 .365 .284 .209 .175 .141 .116 .091 .060 | . 4417 .393 .342 .295 .254 .174 .095 .061 .032 086 086 0927 127 | .199 .154 .113 .072 .034 005 | .383 .302 .242 .191 .082 .037 0637 | 187 | | | |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| - | PERCENT | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|---|--|--|---|---|---|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55ს/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | $0 \alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 15.00 25.00 35.00 35.00 45.00 45.00 65.00 65.00 75.00 90.00 | 251 221 147 074 034 0014 - 0246 - 0707 - 087 - 088 - 069 012 150 159 215 215 221 231 241 251 251 251 251 251 251 251 251 251 25 | .522 071 061 061 074 084 094 095 096 052 052 073 274 2810 2743 793 8029 8029 8029 | . 484 050 044 044 051 037 037 012 . 011 . 040 . 085 . 251 . 314 . 334 385 420 432 432 432 | .754 .001 -009 -009 -009 -0027 -027 -0018 -0027 -0034 -0034 -0034 -0358 -1070 -2568 -3373 -336 -3373 -336 -338 -338 -338 -338 -338 -338 -33 | .082 .126 .127 .286 .3729 .281 .285 .285 | .057 .007 .012 .022 .022 .035 .058 .098 .148 .226 .257 .228 .230 .200 .193 .183 .183 | 372 -152 -101 -041 -022 -036 -077 -087 -0971 -0544 -0365 -0766 -0766 -0766 -077 -1071 -1071 -1071 -1115 |
| LOWER SURFACE | 60.00 | 022 | 074 0561 0743 1158 1137 1258 1899 2155 2652 2652 2652 2117 | 125 1002 1201 131 150 181 226 240 2 | - 1948 - 105 - 137 - 164 - 221 - 224 - 224 - 2256 - 267 - 277 - 267 - 2234 - 2347 - 2347 - 2347 - 2347 - 2347 - 2334 | 325 243 223 223 223 2258 258 2578 278 278 2974 2275 | 404 301 265 262 276 276 397 3121 3271 3271 244 201 128 128 128 | 503 387 298 335 335 208 2121 2121 2121 1134 1134 1134 1134 081 |
| | M = 0 | .90 $\alpha = 3.9^{\circ}$ | 1 | | | | | |
| UPPER SURFACE | 1.25 2.50 7.50 10.00 15.00 20.00 25.00 35.00 40.00 55.00 65.00 75.00 85.00 85.00 90.00 | 246 237 194 064 .268 .467 - 1.082 668 586 454 306 | 307 298 183 091 .015 .127 .192 .167 .021 755 767 733 672 583 | - 1.039 - 926 - 1419 - 1283 - 2217 - 1147 - 1060 - 0022 - 128 - 144 - 163 - 433 - 442 - 4436 | 100 045 .019 .081 .138 .191 .181 .193 339 340 344 | 896 415 352 327 256 207 161 016 016 014 107 148 172 269 269 264 | 001 .052 .102 .137 .150 .140 .191 182 176 176 | 08 05 05 06 06 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 35.000 35.000 45.000 60.000 75.000 85.000 95.000 95.000 | .093 .089 .035 .0011 036 067 104 121 113 131 131 | .200 .148 .121 .073 .036 .004005100119123134119119123 | .058 .013 022 051 075 0799 120 137 143 144 144 184 173 180 | . 3249 . 197 . 142 . 098 . 037 . 037 . 038 . 0159 . 1155 . 1172 . 1179 . 1184 . 1185 . 1222 . 1185 . 2209 . 219 | 055 079 113 127 175 189 202 202 198 198 | . 338 . 255 . 150 . 097 . 097 . 011 - 038 - 081 - 113 - 1153 - 126 - 227 - 239 - 223 - 223 - 223 - 128 - 128 - 128 - 128 | .20 .11 .03 .02 .117 .157 .187 .179 .179 .179 .116 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| | | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | |
|---------------|---|--|---|---|--|---|--|---|--|--|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | | | |
| | M = 0.90 | $\alpha = 5.9^{\circ}$ | | | | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.500 10.000 20.000 30.000 40.000 450.000 450.000 655.000 750.000 850.000 850.000 | 2 5 6 5 5 6 7 2 8 8 3 1 0 3 1 0 3 1 0 3 1 0 3 1 0 3 1 0 3 1 0 3 1 0 1 0 | 082 - 1.067 - 1.093 782 636 549 427 411 363 160 031 074 134 020 7568 740 7568 740 7592 | 189 - 1.206 - 1.12107766603386620581192017600650065447444544454 | .165 -1.086 -1.0269197886574962972471891370147121313435663633351 | 440 - 1.130 - 1.022 836 712 609 396 281 282 147 053 0125 .0612 287 287 287 290 273 | 404 748 622 576 526 466 396 3262 207 152 098 004 .028 .049 209 - | 8457 87158 7407 64111 00539 00444 00678 00998 00988 00 | | | |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 25.00 35.00 45.00 55.00 65.00 65.00 75.00 85.00 85.00 | . 394 . 3964 . 3285 . 285 . 242 . 161 . 173 . 1155 . 083 . 062 . 001 . 001 . 002 . 005 . 0 | . 452 .357 .300 .248 .216 .155 .175 .078 .048 .016 .007 .058 .074 .085 .078 .089 .1022 .115 .171 | . 439 .362 .278 .224 .187 .085 .047 .016 - 011 - 039 - 062 - 1001 - 1107 - 1107 - 143 - 153 - 165 - 192 211 | .417 .293 .241 .191 .1156 .0306 .027 0586 1118 1440 1440 1452 1623 1623 185 | . 430 .362 .271 .211 .127 .027 .025 .038 .038 .1007 .1138 .1470 .150 .150 .162 .162 .178 | . 44 2 .368 .260 .201 .197 .107 .048 .0046 079 117 1182 194 191 171 | .307 .2131 .064 036 100 154 162 1662 145 145 145 107 1085 0770 0557 | | | |
| | M = 0 | $.90 \alpha = 8.0^{\circ}$ | | · - · - · - · · - · · · · · · · · | | T | | Τ | | | |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 20.00 30.00 35.00 45.00 45.00 55.00 65.00 75.00 85.00 | - 445a - 441a - 416a - 425a - 425a - 425a - 435a - 435a - 436a - 1065a - 1065a - 1066a - 6000 - 6000 | - 1.165 - 1.176 - 1.109 - 1.0039267786375384944813351067747537753602 | 358294245175175175146405 | - 1.172 - 1.100 - 1.000 - 8455 - 682 - 6500 - 1.293 - 1.184 - 1.184 - 1.026 026 0364 363 363 363 | 326 2855 2255 277 3566 340 340 278 244 | 765 837 7720 720 662 662 587 5514 2816 2816 2816 152 152 152 152 152 152 154 248 233 233 244 248 | 148 135 | | | |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 30.00 35.00 40.00 55.00 65.00 65.00 85.00 85.00 90.00 | 437 447 325 225 325 325 325 325 325 325 325 325 | 385 325 56 325 68 184 22 184 22 114 23 107 25 184 27 107 28 107 2 | .421 .35. .290 .266 .200 .155 .010 .011 .011 .011 .012 .013 .013 .013 .013 .013 .013 .013 .013 | | . 418 .3379 .2379 .2389 .183 .127 .0052 .017 .0047 .0047 .1080 .1108 .1128 .1128 .1128 .1128 .1128 .1128 | . 420 .323 .265 .259 .170 .059 .022 063 063 131 154 154 154 154 | .371 .2804 .1436 .0917 1237 1543 1543 1092 1647 0867 0667 0657 | | | |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| Γ | | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | | | |
|---------------|---|--|--|--|---|--|---|---|--|--|--|--|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 | | | | | |
| 1 | M = 0.90 | 0 α = 11.3° | | | | | | | | | | | |
| OPPER SORFACE | .00 1.25 2.50 5.00 7.50 10.00 20.00 25.00 30.00 45.00 45.00 65.00 65.00 65.00 85.00 85.00 90.00 | . 214 - 344 - 497 - 619 - 659 - 648 - 640 - 601 - 578 - 5546 - 5545 - 543 - 575 - 443 - 289 - 110 - 1015 - 654 - 6 | 694 - 1.328 - 1.3146 - 1.3275 - 1.1822 - 1.044156513311706420236124779447660617 | 777 - 1 .252 - 1 .113 - 1 .085 - 1 .039 - 1 .039942644644644649618595502510450454404365 | 352 - 1.209 - 1.168 - 1.168 - 1.1405 - 1.1405 - 1.039 - 1.0039 - 1.0039 - 1.039 - 1.039 - 1.039 - 1.039 - 1.039 - 1.3957 3953 361 3973 3567 379 3521 297 | 917 8404 825 826 818 8104 7787 7759 7759 762 6158 5408 410 | 548504505491487470470459438438438438364364364366340 | 462 380 377 377 377 376 368 365 365 356 350 350 350 369 3209 2259 2259 2254 244 | | | | | |
| LOWER SURFACE | 1.25 2.500 5.500 10.000 25.000 35.000 45.000 45.000 65.000 75.000 85.000 90.000 | .419 .512 .590 .567 .517 .391 .391 .3061 .309 .231 .140 .135 .097 .083 .023 .023 .023 | .588 .560 .499 .448 .341 .2249 .2177 .1142 .082 .063 .045 .027 | .530 .511 .452 .407 .314 .259 .216 .178 .144 .111 .082 .051 .010 -014 | . \$45 . 484 . 453 . 400 . 353 . 290 . 241 . 195 . 163 . 127 . 090 . 028 018 035 081 070 081 107 107 | - :117 | .009 036 079 110 134 157 124 150 186 | . 414 .346 .276 .211 .012 .028 .083 .125 .167 .167 .182 .176 .1773 .164 .176 .169 .169 | | | | | |
| | M = 0 | .90 $\alpha = 15.5^{\circ}$ | | | | | 510 | 510 | | | | | |
| UPPER SURFACE | 250 5.000 105.000 105.000 105.000 350.000 350.000 450.000 670.000 670.000 850.000 995.000 | 725 689 6631 631 478 354 027 - 1.127 6893 | - 1.00% 551 415 599 434 8872 7590 453 | 673 | 514 546 560 577 581 571 58 | | 502 498 5198 5198 5111 5111 509 500 4918 4918 4918 4419 44378 44378 44378 | - 467 - 467 - 470 - 470 - 450 - 451 - 454 - 453 - 553 - 553 | | | | | |
| LOWER SURFACE | 1.25 2.50 5.50 5.50 1.50.00 20.00 20.00 30.00 40.00 40.00 40.00 60.00 75.00 85.00 85.00 995.00 | 55 | 6411 5711 5712 4725 | .574 .548 .515 .432 .333 .295 .225 .128 .150 .128 .129 .099 .063 | .54 .53 .49 .45 .34 .34 .32 .32 .32 .33 .33 .30 .31 .30 .31 .30 .31 .30 .31 .30 .30 .30 .30 .30 .30 .30 .30 .30 .30 | 3 | 4 4 448 4 448 5 3910 6 251 6 251 7 201 7 201 1 149 1 149 | .438 .327 .178 .001 .001 .011 .011 .022 .017 .022 .017 .022 .017 .022 .033 .033 .033 .033 .033 .033 .033 | | | | | |

TABLE II

| Ī | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|--|--|--|---|---|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | $4 \alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 1250 1250 1250 1250 1250 1250 1250 1250 | 291 2410 10963 100367 100370 100899 100899 100899 100889 10089 | 5 3 6 0 7 6 0 7 7 6 0 7 7 8 9 7 8 9 7 8 9 9 9 1 8 9 8 9 9 9 9 9 9 9 9 9 9 9 9 | . 489 068 049 049 045 052 015 015 019 016 088 156 248 302 302 302 424 465 466 | 7 7 9014901890180036029029029029036 | .479 .116 .026 .018029029029029029029029029029031029031029031031031031031031031 | . 471 .180 .050 .072 .037 .025 .009 .0007 .007 .037 .074 .135 .211 .216 .225 .221 .224 .225 .198 | . 378 .145 .140 .042 .023 140 106 106 106 078 056 078 1203 088 1203 120 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 450.000 560.000 750.000 850.000 850.000 950.000 | 21610 .0159500255550000000000000000000000000000 | 054 038 0592 1045 125 1444 1664 2301 2587 26694 2792 2597 | 100 079 084 1018 147 172 197 224 224 227 319 329 329 329 329 329 329 251 251 258 2278 229 | 1753 1956 1340 2147 2262 2262 3272 3273 3774 308 3270 3268 3268 3268 | | 412 3199 2899 2940 2297 3562 3562 3784 3946 3946 3946 127 1135 1148 | |
| | M = 0. | $94 \alpha = 3.9^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 15.00 20.00 25.00 35.00 40.00 55.00 65.00 75.00 85.00 85.00 85.00 | 268 273 0866 .2666 .473 - 1.218 717 6887 525 | 305 306 304 069 .167 .212 .168 .135 .024 863 854 866 | 973 893 434 433 291 103 081 040 081 142 154 497 486 486 486 | . 352 - 847 - 705 - 393 - 3939 - 2835 - 2018 - 1192 - 0712 - 0712 - 1173 - 1173 - 1173 - 3883 - 3883 | 285 | 028 9072 85607 3576 2174 1174 00405 1239 11399 11906 20037 1971 3331 | . 97893 97893 543349 217949 109803 007757 00992407 009857 00857 00857 00857 00857 00857 00857 00857 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 25.00 30.00 35.00 40.00 45.00 55.00 60.00 70.00 75.00 85.00 90.00 95.00 | .347 .326 .279 .235 .200 .153 .102 .016 .004 .0016 .003 .0086 .0086 .0086 .0097 .009 | .342 .273 .204 .156 .080 .047 .0022 .052 .072 104 .135 .147 147 136 136 .136 | .358 .279 .139 .092 .052 060 085 113 154 155 155 155 155 243 243 | 323 246 188 1389 1090 0288 - 011 - 048 - 070 - 120 - 1247 - 177 - 1787 - 199 - 216 - 253 - 255 - 258 | . 3 2 3 4 8 . 160 4 . 106 6 . 0 30 5 . 0 5 9 1 . 118 9 . 213 5 . 24 7 7 . 23 9 . 230 4 . 26 5 | . 3351 . 1409 . 0904 - 0500 - 1277 - 1666 - 2253 - 2373 - 3347 - 321 - 1450 - 1333 - 1430 - 1333 - 1442 | . 299260 . 103299 . 1032995 . 0022995 . 0022995 . 00229960 . 0022996060 . 0022996060 . 0022996060 . 0022996060 |

CONFIDENTIAL

TABLE II

| ſ | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | |
|---------------|---|---|---|--|--|---|---|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 | | |
| | м = 0.9 | 4 α = 5.9° | | | | | | | | |
| UPPER SURFACE | 12.5000 12.5000 10.500 | 289 28289 0239 22676 22793 23005 33026 33036 33036 3136 213 | . 014 - 997 - 1.013 - 6542 - 4476 - 2398 - 2398 - 2388 - 2388 - 2387 - 0084 - 1147 - 0074 - 6839 - 8376 - 8376 - 8376 - 8376 - 8376 - 653 | - 1.1837 - 1.0002 - 1.1837 - 1.0002 8684 4466 1771 0612 0136 1791 0106 1791 5938 4888 | 2553537 1.183537 1.106538 1.106538 1.106538 1.106538 1.106538 1.10638 | - 1.176 - 1.1776 - 1.1031 9668 3157 22535 0941 0171 1142 2884 2889 2886 | - 381 - 1.0947 - 8759 - 4696 - 4996 - 318 - 2197 - 137 - 137 - 1080 - 2025 - 2055 - 2055 - 2055 - 2068 - 20 | 396742 598742 598742 598752 598752 598752 598752 59976 59 | | |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 50.000 65.000 75.000 85.000 95.000 | .406 .44071 .33370 .2573 .1825 .10918 .00366 .000144 .00548 .00548 .00777 .00777 .009919 | 458 3311 22597 11665 10863 10024 100570 100770 100786 10085 | :437 :359 :279 :224 :188 :134 :049 :015 :013 :0064 :0086 :0966 :0986 :130 :1499 :11491 :2157 | 4126 328365 4107310 40000581 4000000000000000000000000000000000000 | . 4256 .32655 .22168 .12266 .0252 .00252 .00463 .17424 .11889 .1208 .2198 .2198 .2198 | .431 .3249 .1186 .0036 .0050 .0050 .0050 .0050 .0050 .0050 .01381 .0267 .0267 .0391 | 8567354656458 332137324566458 | | |
| | M = 0. | .94 $\alpha = 7.9^{\circ}$ | | T | | T | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.50 7.500 10.00 15.000 20.000 25.000 40.000 55.000 65.000 75.000 75.000 90.000 90.000 | 310 131 .355 - 1.166 714 567 456 431 | 864 788 677 | 280 - 1.244 - 1.191 - 1.157 - 1.031773614492380200243315813313313581346098535446588 | - 1.288 - 1.316 - 1.261 - 1.278 - 1.0647524643492331480790590651044304274274274274397 | - 1.297 - 1.290 - 1.246 - 1.187 - 1.0778387628561489414345285194178 | 690 640 640 573 538 506 479 459 439 439 367 328 261 226 226 | 5998435831972972954463261220522092118621162911629117 | | |
| LOWER SURFACE | 1.25 2.5000 10.000 15.000 25.000 35.000 40.000 40.000 65.000 65.000 80.000 95.000 | .193 .129 .095 .061 .037 .010 006 006 037 034 058 | . 528 . 463 . 389 . 335 . 299 . 237 . 191 . 150 . 082 . 0054 . 020 0043 042 043 042 079 - 079 - | .106 .070 .040 .010 017 | . 451 .376 .345 .289 .244 .171 .127 .085 .020 -0145 -0798 -120 -1135 -179 -1826 -184 | .398 .316 .257 .216 .162 .109 .027 006 140 140 148 150 | . 400 . 303 . 242 . 242 . 148 . 090 . 038 003 088 136 224 255 275 275 | .350 .1267 .1293 .1215 .0519 .0519 .2555 .2288 .2280 .2808 .1502 .1502 .0813 .0813 | | |

TABLE II

| ſ | | | - | PRESSUR | E COEFFICIENT, | P, AT: | - | |
|---------------|--|---|--|---|--|---|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.255/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.9 | 94 α = 11.3° | | | | | | |
| UPPER SURFACE | 12.50 5.000 105.000 12 | 2 4 3 - 2 5 8 - 4 06 - 5 2 8 - 5 6 9 - 5 6 9 - 5 13 - 5 6 9 - 5 13 - 4 9 2 - 4 9 2 - 4 2 7 1 - 113 - 2 7 1 - 120 8 - 1 21 8 - 5 3 10 - 4 7 2 | 563 - 1.216 - 1.197 - 1.228 - 1.201 - 1.149996998624432057035508192539253925393106731 | - 625 - 1.229 - 1.101 - 1.048 9513 8119 5361 5505 5297 636 6669 6669 7333 5518 5518 | 181 1,357 1,357 1,357 1,379 1,279 1,229 1,29 | | 73867536675366563664259635544552545525448714411411613182 | 521444436418410401400397387387363363367335367335367335 |
| LOWER SURFACE | 1.25 2.500 10.500 10.500 25.000 25.000 35.000 45.000 45.000 65.000 65.000 805.000 905.000 | 4 2 67 . 6 3 4 69 . 5 5 7 7 . 4 4 8 8 8 . 3 2 6 6 9 . 2 2 6 9 . 1 4 7 0 . 0 7 2 4 5 . 0 7 2 4 5 . 0 7 2 4 5 . 0 0 0 0 0 | .607 .507 .507 .427 .3518 .2635 .1859 .121 .0071 .0071 .0036 | .546 .518 .451 .451 .375 .264 .222 .184 .153 .087 .037 .037 .009 093 113 160 202 | . 552 . 483 . 483 . 495 . 344 . 2838 . 195 . 177 . 041 . 020 . 046 . 072 . 115 . 125 . 125 | . 488 . 470 . 408 . 3510 . 259 . 203 . 157 . 119 . 078 . 004 - 024 - 073 - 073 - 126 - 146 - 157 - 137 - 203 | . 478 . 478 . 391 . 3330 . 322 . 238 . 177 . 125 . 079 . 030 . 012 . 108 . 1470 . 1207 . 1300 . 2267 . 299 | . 456 . 415 . 343 . 276 . 211 . 112 050 117 162 192 234 234 235 249 249 249 |
| | -M = 0 | .94 $\alpha = 15.6^{\circ}$ | | | | | | |
| UPPER SURFACE | 10.00 1.25 20.50 7.50 10.00 20.00 25.00 35.00 45.00 65.00 60.00 70.00 80.00 90.00 | 608 777 818 800 735 6653 6653 608 608 636 4307 097 102 857 6537 654 465 | 925 - 1.349 - 1.322 - 1.354 - 1.351 - 1.258 - 1.258 - 1.213 - 1.202729243085327102950898788617 | - 1.231 - 1.152 - 1.162 - 1.162 - 1.154 - 1.101 - 1.101 - 1.071 - 1.071 - 1.923 - 895 - 895 - 838 - 7780 - 7780 - 7780 - 723 - 684 | 741741737737712712712740738727740738668668668668668668668668 | 556 553 550 | 522 514 495 | 556 5279 5146 516 516 516 516 520 |
| LOWER SURFACE | 55.00 | 5468 6688 6743 66355 653557 4463 3377 2271 22146 21122 1122 1123 1124 1126 | .645 .667 .635 .592 .559 .494 .446 .399 .322 .288 .219 .171 .152 .171 .152 | .114 | .587 .558 .545 .501 .461 .403 .351 .309 .271 .233 .1193 .1153 .1164 .056 .028 -057 -099 -139 -139 | . 493 .522 .490 .450 .412 .365 .314 .264 .123 .101 .067 .001 .0023 .001 .0023 .0023 .1266 .167 .276 | . 456 . 411 . 324 . 266 . 212 . 164 . 113 . 066 . 0143 | . 435 . 441 . 342 . 286 . 1991 . 00153 1159 1259 1259 2287 287 3016 377 |

TABLE II

| Γ | | | | | | | PRESSURE (| COEFF | ICIENT, I | P, AT: | | | | | |
|---------------|--|---|--|------|--|------|---|-------|--|--------|---|------|--|-----|--|
| | PERCENT CHORD | 0.13 | 5b/2 | 0.25 | b/2 | 0.40 | b/2 | 0.5 | 5b/2 | 0.70 | b/2 | 0.85 | b/2 | 0.9 | 5b/2 |
| 7 | M = 0.9 | 8 ar= | = -0.1° | | | | ·-·· | | | | | | . 486 | | . 430 |
| UPPER SURFACE | 1 25 25 25 5 00 7 0 00 15 00 15 00 00 1 | - 1 | 344 2701 2201 2201 2201 2201 2201 2201 2201 | | .561 .034 .019 .019 .054 .068 .079 .0262 .138 .3295 .3277 .9413 .945 .8644 | - | 5 0 6 4 9 3 9 1 8 3 1 9 1 9 1 9 1 9 1 9 1 9 1 9 1 9 1 9 1 | = | .803 .0105 .0105 .0105 .012 .012 .0012 .0014 .0025 .048 .048 .048 .048 .048 .048 .048 .048 | | .494 .034 .034 .0020 .0020 .0020 .0020 .0021 .00 | | .145 .0223 .018 .0032 .018 .0032 .00213 .0017 .0019 .0146 .12213 .2110 .2213 .2110 .2213 .2213 .2213 .2213 .2213 | | 10728 10728 10728 10728 10728 10728 11623 11 |
| LOWER SURFACE | 1.25 2.50 7.500 10.000 15.000 30.000 35.000 40.000 55.000 60.000 670.000 80.000 90.000 90.000 | | 246234904 1943 109627 100627 1007 1007 1007 1147 11908 | | 022 0010 0129 00129 0074 11356 11356 1147 1225 12454 1225 12454 12554 125 | | 0 5 7 0 0 5 7 0 0 5 7 0 0 5 7 0 0 5 0 2 2 2 1 0 5 6 1 1 1 6 1 3 2 2 3 5 2 2 2 2 2 2 2 2 2 2 2 2 4 7 5 2 2 2 2 4 7 5 | | 122 110 110 110 1120 1120 1121 1221 122 | | 256 256 256 256 256 256 256 256 256 256 | | .24439 .224499 .12587 .231301 .2587 .3357 .448733 .448733 .448733 .448733 .4137 .332043 .332043 | | |
| | M = (| 0.98 | $\alpha = 3.9^{\circ}$ | | Т | | | | 417 | Γ | .058 | | .042 | | . 1 2 7 |
| UPPER SURFACE | 50.00 | | . 196 .209 .228 .268 .122 .058 .463 1.249 .700 .466 .384 | - | 2976 .71846 .72986 .22664 .226 | _ | .011 .059 .156 .152 .165 .174 .596 .603 | | .417 .8848 .846 .628 .628 .628 .628 .628 .628 .628 .62 | | 07930772650166656663695391 0793052650167695391 0793052650167695391 0793052650167695391 0793052650167695391 0793052650167695391 0793052650167695391 0793052650167695391 | - | 83565229255543005557857295837 | ı – | 9649915212626102399664917211101239966491721110123211111111232121111111111111111 |
| 1 8 | 1.2.50 70.00 10.00 | 000000000000000000000000000000000000000 | 323 2235 2235 2353 1334 1334 10746 0046 0077 1021 11228 1138 | | .110 .119 .129 .138 .138 .145 .126 | | .047 .074 .100 .130 .153 .160 .159 .122 | | .096 .122 .163 .164 .143 .154 .200 .203 | - | .261 .112 .0016 .0016 .0072 .008 .108 .108 .108 .108 .108 .108 .108 | = | .26257374 .11077.11074.0027086 .10027086 .121672.0027150000000000000000000000000000000000 | : | 224 1474 1077 1017 12597 13317 13317 13317 13317 13317 |

TABLE II

| ſ | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | | |
|---------------|---|--|---|---|---|---|--|--|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | | | |
| | M = 0.98 | $\alpha = 5.9^{\circ}$ | | | | | | | | | |
| UPPER SURFACE | 1.25 2.50 7.50 10.000 20.000 30.000 40.000 450.000 450.000 650.000 750.000 85.000 85.000 | .319 .023 -0623 -174 -210 -210 -2250 -2250 -2250 -2265 -2270 -285 -2271 -015 -1015 - | .09496565545545536735535335335935406209611812901110329450679 | 010 - 1.071 - 1.004 9234 9254 3991 4042 095 0449 0710 1261 1261 6234 6334 6333 | . 326 - 1.0502 - 1.037984386734411899104062706291974961507509 | 251 - 1.070 - 1.103899607953167916790017035113711440443504437 | - 1.0427 - 1.0427 - 1.10336 9952 5888 23058 23058 23058 2004649 1004649 1004649 1004649 | 217 - 1.1106 - 1.0009946496718631538147221452258225823572312 | | | |
| LOWER SURFACE | 1.25 2.500 70.500 10.000 25.000 35.000 45.000 45.000 555.000 665.000 750.000 850.000 95.000 | . 420 . 427 . 397 . 3516 . 2731 . 203 . 1142 . 1088 . 023 . 0024 . 0024 . 0044 . 00710 . 0085 . 0085 . 0085 . 0085 | . 465 .3996 .3199 .2242 .1739 .1069 .0312 .0051 .0051 .0063 .00762 .00762 | .445 .3843 .2843 .1945 .0521 .0521 .0057 .0057 .0057 .00926 .0093 .1013 .1133 .1133 .1214 | . 403 .308 .278 .235 .114 .070 .032 .0123 .023 .048 .097 .1123 .1123 .1123 .1185 .2034 .277 | .416 .346 .258 .199 .1618 .065 .027 .034 .034 .0711 .1150 .156 .176 .185 | . 430 .359 .1995 .1095 .0048 .0035 0796 1168 3168 328 3188 35869 | .394 .38379 .10094 .0094 .1257 .33347 .334 | | | |
| | M = 0. | 98 α = 7.9 ⁰ | | | 1 | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 15.00 25.00 30.00 35.00 45.00 45.00 60.00 65.00 75.00 80.00 90.00 | i | 114 - 1.053 - 1.056976567148442643420150014 - 1.0279449708 | 175 - 1.180 - 1.126 - 1.061 - 1.0696724871701029026036048073658654657 | .203 - 1.164 - 1.200 - 1.1459 - 1.073 - 1.0279035822444217080012083176182526526526 | 452 - 1.195 - 1.212 - 1.157 - 1.166 979 747 563 339 242 203 178 178 153 153 153 154 514 514 | 529 - 1.0810 - 1.088963963707247954619409237543484448844884488 | - 1.054 - 1.0554 - 1.0554 - 1.0056 - 964 7022 6284 5307 4208 3827 4082 3315 3156 3156 3156 3156 3156 3156 3156 3278 | | | |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 45.000 60.000 65.000 75.000 80.000 895.000 | . 465 . 444 . 401 . 353 . 287 . 272 . 145 . 110 . 079 . 053 . 025 . 007 . 023 . 029 . 044 | .543 .476 .402 .349 .317 .251 .208 .165 .126 .095 .070 .033 .009 004 022 031 | .505 .436 .359 .301 .269 .215 .160 .121 .087 .055 .024 .000 045 .049 .049 .049 .049 .049 .049 .049 .049 | .255 .187 .144 .102 .072 .037 .008 028 048 087 104 152 | .407 .325 .269 .228 .177 .132 .081 .042 .010 025 062 142 154 154 154 204 224 | . 407 . 3150 . 2559 . 1051 . 0039 0807 1275 2263 2263 2853 2853 3384 | .362 .278 .203 .144 .044 | | | |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| Ī | DUDGEN | | | PRESSUE | RE COEFFICIENT, | P, AT: | | |
|---------------|--|--|---|--|---|---|--|---|
| | PERCENT - CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85 _Þ /2 | 0.95b/2 |
| | M = 0.9 | $\alpha = 11.4^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.500 7.500 15.000 25.000 25.000 45.000 65.000 65.000 65.000 65.000 90.000 | . 291 - 185 - 327 - 448 - 498 - 498 - 496 - 463 - 441 - 424 - 424 - 435 - 382 - 283 - 283 - 283 - 270 - 473 - 473 - 473 | 453 - 1.112 - 1.096 - 1.121 - 1.093 939 820 716 582 427 427 047 047 047 085 1330 980 - | - 496 - 1.1251 - 1.0045 9276 8247 6573 4388 4525 5881 64460 7556 6143 5883 | 690 637 571 498 389 3587 554 554 509 | 7 5 9 3 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 7 5 9 9 9 7 5 9 7 5 9 9 9 7 5 9 7 5 9 9 9 7 5 9 9 9 9 | 1 | 7 4 2 7 1 8 6991 6891 6692 66402 5865 5424 5865 5424 5424 4400 44508 4400 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 45.000 65.000 75.000 85.000 85.000 90.000 | 451 .569 .716 .605 .518 .407 .3296 .273 .235 .138 .120 .047 .056 .056 .0024 | .634 .591 .4448 .379 .3332 .2478 .2185 .1185 .1170 .0066 .0035 .001667 | .573 .542 .481 .433 .399 .342 .284 .311 .178 .147 .088 .065 .050 .029 033 061 1159 | .593 .508 .471 .417 .367 .305 .212 .182 .147 .109 .074 .043 .016 -009 -1100 -1100 -1100 -1161 -1209 | .510 .420 .3724 .277 .2227 .177 .1035 .0055 -0042 -0630 -1148 -1640 -251 | . 494 . 475 . 402 . 346 . 336 . 252 . 1193 . 146 . 101 . 055 . 0041 - 083 - 1216 - 1216 - 223 - 2724 - 354 | .4667 .3615 .223389 .00482 .1009 .00482 .17089 .2268817 .2293112 .2393111 .3324010 |
| | M = 0 | .98 $\alpha = 15.8^{\circ}$ | | | | | , | · · · · · · · · · · · · · · · · · · · |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 25.00 35.00 45.00 45.00 55.00 65.00 75.00 85.00 85.00 | 584 5543 545 545 488 391 062 0620 620 523 | 846 | 983 970 - 1.014 - 1.091 - 1.149 - 1.086 992 968 988 854 | 901 8818 893 928 927 - 1.0353 - 1.053 - 1.053 - 1.053 - 1.053 - 1.053 - 1.7974 9073 7711 7733 7755 691 | 678 66731 66395 66827 64873 71193 71193 71087 70876 662907 66375 | 631 630 621 614 610 602 595 589 589 588 581 588 588 588 | 607 5881 5774 5774 55774 55774 55774 55775 55776 55776 55775 55575 55775 55575 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 25.00 30.00 40.00 55.00 60.00 65.00 60.00 60.00 90.00 95.00 | .585 .774 .776 .583 .583 .493 .493 .493 .263 .212 .212 .145 .150 | .34 5 .34 4 .31 5 .27 4 4 .22 4 .20 0 .18 0 .15 5 .03 4 | .581 .518 .46 .411 .366 .333 .291 .26 .23 .17 .14 .11 | 57 565 520 481 424 4 386 384 386 386 386 386 386 386 386 386 386 386 | .5475 .4759 .4399 .2396 .2216 .2169 | . 489 . 443 . 436 . 361 . 307 . 255 . 213 . 164 . 119 . 068 . 018 . 018 . 029 1069 143 | . 4769 .3787 .3243 .3243 .16603 007203 11592 12596 227965 231484 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| ſ | | *** | | PRESSURE | COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|--|--|--|--|--|
| ļ | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.0 | $0 \alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 125000000000000000000000000000000000000 | . 410 .291 .226 .159 .0952 .0952 .0025 .0030 .00 | 583 - 0006 - 0006 - 0018 - 0045 - 0045 - 0059 - 0078 - 1064 - 1064 - 1064 - 1078 - 1098 - 109 | . 518 - 0529 - 0245 - 0015 - 0015 - 0104 - 029 - 0587 - 102 - 1206 - 2828 - 3324 - 5881 - 5899 - 587 | .821 .028 .028 .035 .045 .046 .020 .031 .050 .072 .113 .175 .255 .346 .464 .471 .482 .487 .483 | .511 .138 .056 .048 .022 .000 .002 .0002 .0009 .022 .044 .072 .114 .260 .312 .312 .3125 .463 .483 .483 .483 .483 .483 | . 494 .162 .038 .067 .031 .011 .0017 .001 .008 .021 .033 .107 .237 .2379 .2379 .2378 .3778 .3778 .3782 .3782 .3582 | . 438 .1282 .00411 0070 1560 12961 00654 12961 00654 2379 2276680 2277 2266807 2277 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 25.00 35.00 40.00 40.00 50.00 65.00 65.00 75.00 85.00 95.00 | 269 2273 1120 0059 0059 0059 0059 10089 1114 11669 11797 11997 22120 2225 | .00 2 .00 20 .00 20 .00 20 .00 18 .00 45 .00 45 .10 8 .10 14 .11 14 .11 17 1 .12 19 9 .22 17 .22 17 .22 17 .22 17 .22 17 .22 17 .22 17 .22 17 .22 18 .22 19 4 .12 2 | 032016026048057084111136181206233258262262248219247 | 092 083 031 069 145 169 198 211 239 263 289 319 332 331 305 264 280 | 236 163 166 182 195 224 255 272 367 367 369 386 386 386 386 312 312 312 336 | 303 227 238 230 180 2413 293 3311 3331 3361 4162 4462 4837 3107 3107 3107 3107 3107 | |
| ļ | M = 1. | 00 α = 3.9 ⁰ | | | | - | | T |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 20.00 25.00 35.00 45.00 65.00 65.00 65.00 75.00 85.00 | - 481 | 802 | . 2 2 0 8554 7824 5712 3112 3111 3033 0404 00315 . 1754 . 1691 5914 6619 6610 | . 4 4 6 . 8 3 5 . 8 3 5 . 7 9 5 . 3 5 3 7 . 11 1 3 3 . 00 4 6 4 . 00 4 7 5 . 12 1 9 0 6 1 . 14 9 9 9 . 50 1 6 . 5 1 4 | .077 7522 4698 237803 22636 107236 1 | .082 797 817 331 2747 192 148 071 029 .020 .027 .1126 .117 381 362 3328 .331 | .165 8502 64669 27371 23310 11039 009333 123538 25538 23377 244 |
| LOWER SURFACE | 55.00 | .348 .2517 .1534 .097 .0530 .0230 .0036 .0036 .0036 .00816 .1016 .1112 | .379 .3117 .247 .203 .17125 .0955 .0255 0267 0267 1133 1134 129 129 | . 364 .286 .215 .168 .089 .041 .003 .028 .054 .054 .107 .149 .1149 .1159 .1159 .1111 .1169 .1158 .1197 | . 331 . 254 . 206 . 116 . 047 . 010 . 028 . 051 . 081 . 107 . 1137 . 1163 . 163 . 163 . 172 . 177 . 179 . 179 . 223 . 263 | .340 .269 .180 .088 .050 - 004 - 071 - 095 - 1138 - 1459 - 189 - 189 - 226 - 231 - 242 - 213 - 242 | . 256 . 152 . 103 . 114 . 032 . 050 . 080 . 1149 . 1149 . 1246 . 288 . 321 . 343 270 290 348 351 | . 234 . 158 . 091 . 0455 1245 2669 2907 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| | | PRESSURE COEFFICIENT, P, AT: | | | | | | 7 |
|---------------|--|---|--|--|---|--|--|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40ь/2 | 0.55b/2 | 0.70ъ/2 | 0.85ъ/2 | 0.95b/2 |
| | M = 1.0 | $0 \alpha = 5.9^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.500 7.500 15.000 20.000 30.000 40.000 50.000 50.000 65.000 85.000 95.000 | . 365 . 051 . 051 . 145 . 179 . 203 . 121 . 234 . 234 . 234 . 234 . 243 . 244 . 243 . 244 . 244 | .135 927 891 626 410 3162 3316 317 319 325 325 317 126 117 126 119 034 119 034 119 034 119 034 119 034 119 034 119 034 119 034 119 034 119 034 119 034 119 - | . 032 - 1 .004 976 939 858 806 406 354 268 208 073 034 .094 .094 .145 .138 .163 138 .163 615 633 637 612 | . 344 - 1979 - 1.9615 9615 3669 1152 1153 0837 1552 0837 2105 2105 2105 55229 55225 | 191 9929 9625 9625 9625 1415 1419 1429 1 | - 249 - 9063 - 19762 - 19762 - 19637 - 2084 - 1843 - 1443 - 1043 - 1057 - 11053 - 13765 - 13765 - 13765 | - 1039 - 1.0356 - 1.9320 9920 8611 2149 1228 1228 11723 12355 2776 2272 22762 22764 22764 |
| LOWER SURFACE | 1 . 25 | . 4 33 . 4 44 . 3 98 . 3 771 . 3 368 . 2 2 2 9 . 2 1 6 2 . 1 2 0 . 1 0 7 4 . 0 0 4 0 . 0 0 1 1 . 0 0 3 4 . 0 0 6 5 . 0 0 6 6 . 0 0 7 6 . 0 0 6 6 . 0 0 6 6 . 0 0 7 6 . 0 0 6 6 . 0 0 6 6 . 0 0 7 6 . 0 1 1 1 | . 486 . 410 . 340 . 288 . 2860 . 1999 . 1187 . 0053 . 0295 - 0333 - 0460 - 0766 - 0776 - 0761 - 0761 | .460 .385 .304 .252 .2166 .074 .016 .074 .018 .077 .077 .077 .077 .078 .079 .098 .140 | . 423 .329 .302 .255 .132 .053 .053 .005 .005 .005 .005 .006 .007 .008 .008 .008 .008 .008 .008 .008 | . 420 .351 .265 .207 .169 .124 .075 .010 .022 .051 .113 .113 .158 .158 .158 .158 .208 .223 .203 .291 | . 4359 . 3555 . 2000 . 1114 . 0012 - 0062 - 0143 - 12955 - 2796 - 3000 - 3178 - 377 | |
| | M = 1 | $.00 \qquad \alpha = 7.8^{\circ}$ | | · | , | · · · · · · · · · · · · · · · · · · · | | |
| UPPER SURFACE | 1.25 2.50 7.50 10.00 20.00 30.00 30.00 40.00 55.00 60.00 60.00 60.00 75.00 80.00 90.00 | | 053 -1.018 -1.002931696571467422397383395177041123098107025975985985666 | 125 - 1.116 - 1.068 - 1.001987921659467355147069021031070098134638647648 | . 255 - 1.092 - 1.131 - 1.074 - 1.033 - 1.004 967 581 1748 1748 147 207 5151 51519 51519 | - 1.131 - 1.152 - 1.101 - 1.059 - 1.039 - 1.9372 772 551 280 280 120 120 120 080 537 537 537 5537 | 390 361 342 299 299 246 483 456 433 456 | 433 - 1.088 - 1.086 - 1.019 985 7583 486 443 424 4409 409 409 3751 3349 3344 |
| LOWER SURFACE | 1 33.30 | 509 488 4425 .3815 .2915 .2306 .168 .1417 .0077 .0073 .0028 .0015 .00127 | .272 .226 .191 .155 .120 .094 .060 .027 001 003 | 026 034 052 064 087 130 | . 159 . 089 . 028 . 028 - 029 - 047 - 060 - 083 124 136 | .415 .333 .277 .233 .189 .058 .058 .058 .007 .069 .1042 .131 .133 | . 412 .3158 .2588 .2589 .1169 .0221 .0025 .0253 .0253 .0253 .0253 .2369 .2534 .2534 .3167 | 120 242 2805 328 341 337 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO ĞAP)

| ſ | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|---|--|---|--|--|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.0 | 0 α = 11.4° | | | | | | |
| UPPER SURFACE | 12.50 12.50 10 | 303 -158 -299 -420 -469 -469 -470 -438 -419 -406 -402 -4106 -402 -439 -3618 -218 -218 -218 -3218 | - 421 - 1.085 - 1.0899 - 1.0899 - 1.0899 - 1.0899 98529 88529 6864 4196 00771 10400 955775 98775 7876 | 453 - 1.0831 - 1.9916 9916 8957 7256 6551 4824 4317 55669 55213 6669 66608 5351 | - 008 - 1.217 - 1.120 - 1.155 - 1.120 - 0763 - 0763 - 0763 - 0861 - 0813 - 0873 - 0884 - 0873 - 0884 - 0873 - 0884 - 0 | - 1.2037 - 1.2037 - 1.1745 - 1.1745 - 1.1745 - 1.199362 7646 7258 76933 6668 66308 66308 66308 66308 66308 66308 66308 | | 7 3 2 3 2 3 2 3 2 3 2 3 2 3 2 3 2 3 2 3 |
| LOWER SURFACE | 1.25 2.500 7.500 15.000 25.000 25.000 45.000 45.000 65.000 670.000 80.000 90.000 | 4582 .58817 .6188 .54437 .328411 .1301 .1180 .00648 .00648 | 647 6543 7457 73842 7457 73842 72961 7223 7156 71045 71045 715 715 715 715 715 715 715 71 | .581 .488 .436 .3498 .2559 .1863 .1276 .079 .0642 -0048 -00492 -139 | . 597 . 517 . 478 . 478 . 377 . 227 . 1291 . 155 . 028 . 0028 . 0024 . 085 . 0024 . 1043 . 1044 . 1043 . 10 | . 518 . 496 . 3733 . 28333 . 1848 . 1075 . 0028 . 0028 . 0058 . 1073 . 1545 . 1346 | .503 .484 .412 .3545 .2603 .1539 .0640 .0228 1253 1265 1265 1850 2740 351 | . 477 . 4477 . 3776 . 3250 . 1049 1049 1198 12366 12993 3308 3125 3403 3378 |
| | M = 1 | $.03 \alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 20.00 35.00 35.00 40.00 40.00 55.00 60.00 65.00 65.00 80.00 85.00 | .679 .245 .191 .125 .063 .063 .01200440610861015 .408157438324 | .576020009008048059074110110118059 .290 .388 .290 .3889398668600602 | .513 -065 -0450 -0553 -0890 -089 -089 -089 -1489 -23368 -3368 -3565 -528 -534 -534 | .832 .037 .045 .084 .0756 .066 .071 .096 .1155 .2154 .369 .376 .422 -421 -448 -448 | . 55 1 . 18 1 5 7 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 | . 52 9 . 2 4 2 . 0 8 4 . 10 9 . 0 7 4 . 0 2 5 . 0 0 4 8 . 0 6 0 3 . 0 9 7 . 1 4 4 5 . 2 6 7 . 2 4 4 3 . 2 4 4 3 . 2 4 7 . 3 3 3 5 . 3 1 6 | . 468 . 1699 . 10919 . 00427 00108 10186 00495 00414 019905 224459 22433 22433 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 25.00 25.00 35.00 40.00 55.00 60.00 60.00 70.00 75.00 85.00 | .009 -062 -067 -091 -114 -165 -184 -183 -201 -129 -214 | 013 .005 .005 .005 014 030 056 092 106 133 145 174 205 211 220 217 223 223 223 223 211 223 - | 033 021 034 049 085 112 131 153 166 192 218 248 249 249 249 224 224 249 249 249 | 045 0045 0038 062 114 137 166 206 235 260 282 298 298 296 214 214 214 2134 235 | - 203 - 1299 - 1457 - 1667 - 1216 - 2353 - 2708 - 3550 - 3550 - 3550 - 2670 - 2682 - 2293 | 27319519412462047257229635735574473278294278296296 | - 368 - 2631 - 22477 - 225907 - 331292 - 33689 - 33884 - 33515 - 27773 - 22565 |

TABLE II

WING WITH UPPER SURFACE SPOILER (NO GAP)

| T | | ,- | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|---|--|--|---|--|---|
| _ } | PERCENT | 0.135ь/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.0 | $3 \alpha = 3.9^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.50 10.00 15.00 25.00 25.00 35.00 45.00 45.00 65.00 65.00 65.00 65.00 90.00 90.00 | .547 .114 .029 056 098 125 1562 162 162 162 168 | . 353 614 254 2249 2249 2247 2247 2314 2451 2451 2451 2451 2135 1114 1114 1114 1114 1114 1114 1114 1114 11 | .246 7843 634 2997 2997 2992 2992 2992 2992 2992 2992 2992 2992 2992 2993 2993 2995 - | 482 | .123 792 715 616 310 084 0912 059 025 .028 .078 .1444 .239 416 414 414 415 414 | .106 .722 .732 .537 .296 .246 .217 .165 .118 .082 .008 .056 .107 .149 .160 .160 .150 .350 .3530 .3210 .301 | . 2 02 . 782 . 734 . 603 . 4440 . 241 . 194 . 186 . 100 . 055 . 049 . 086 . 196 . 227 . 227 . 227 . 217 . 216 . 216 |
| LOWER SURFACE | 1.25 25.500 70.500 105.000 205.000 405.000 405.000 505.000 605.000 805.000 905.000 | .324 .33253 .2237 .2206 .1149 .0071 .0053 0073 0073 0073 11006 | .366 .3041 .198 .1167 .1125 .089 .057 .029 .000 -018 -078 -091 -098 -113 -112 -098 -112 -097 -018 | .360 .290 .217 .170 .142 .098 .018 - 018 - 041 - 089 - 1124 - 1213 123 077 111 148 | . 346 . 269 . 223 . 174 . 133 . 0685 . 0004 - 0024 - 0056 - 1055 - 1125 - 1130 - 1125 - 1145 - 1465 - 209 | . 353 .285 .201 .148 .111 .080 .0089 .0083 0633 108 122 1454 154 178 178 175 175 | . 374 . 299 . 195 . 147 . 151 . 072 0036 075 1107 148 243 279 259 259 259 290 306 311 | .341 .2600 .1000 .1000 .1000 .1000 .1000 .1000 .1000 .22557 .22557 .226619 .22557 .22659 .22557 .22659 .22559 |
| | M = 1. | $03 \alpha = 5.8^{\circ}$ | | | | | | T |
| UPPER SURFACE | .00 1.25 2:50 5.000 10.00 15.000 25.000 25.000 40.000 50.000 60.000 60.000 70.000 85.000 85.000 | .508 .061 -:032 -:127 -:175 -:1950 -:2019 -:228 -:238 | .182879833766381296293291297314033152191070931840744 | . 0710 9708 87762 87762 28762 28556 00606 150606 150606 15972 558747 558747 | .382 908 962 898 852 583 1573 079 040 .008 .122 252 .240 .252 477 485 492 485 492 | | 190923984895883757221137094004300430051331331364352338832913461 | - 123 - 9962 - 88741 - 88537 - 2236 - 11206 - 11006 - 11028 - 12443 - 22448 - 22444 - 2249 - 432 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 20.00 30.00 30.00 40.00 40.00 60.00 75.00 60.00 80.00 85.00 | .123 :0966 .041 .0122 002 0047 0436 | .170 .139 .107 .058 .002 .025 .024 .038 .038 | .395 .3167 .2355 .184 .1361 .070 .0122 036 0477 0477 0477 0355 | .344 .3708 .2788 .2288 .1599 .1160 .050 0023 0040 0555 0655 1107 1360 | .383 .298 .243 .203 .111 .075 .047 020 009 073 115 161 175 175 | .390 .288 .235 .235 .148 .093 .044 .014 025 061 151 192 232 259 269 305 331 | .360 .287 .211 .159 .0070 .121 .187 .284 .298 .299 .270 .256 .256 .256 .257 .277 |

TABLE II

| | PERCENT | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|--|--|---|--|---|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.0 | $3 \qquad \alpha = 7.8^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 25.50 7.50 10.00 120.00 230.00 330.00 340.00 450.00 450.00 670.00 670.00 750.00 850.00 850.00 950.00 | . 470 . 0093 . 0193 . 2456 . 2253 . 22669 . 22667 . 22667 . 22667 . 23664 . 1510 . 1 0985 . 3882 . 337 | 009 - 1.00297493582760340593473453451601481037898910825609 | 082 - 1.0439 9999 99321 97221 4732 4739 1545 0182 1441 5801 5982 5982 | 297 1 025 1 007 2 968 2 940 2 9856 2 405 2 305 2 197 2 051 2 077 2 160 2 249 2 254 2 270 2 463 2 461 | 339 - 1.046 - 1.066 - 1.066 - 1.0809809588977305114245146125110110485511505 | 411 - 1.000 - 1.062 967 965 920 763 427 370 352 3316 303 279 283 225 4437 419 373 | 377 - 1.038 - 1.038 - 1.9689689408766242038037737763776377637663566326432643266 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 10.00 20.00 30.00 40.00 40.00 40.00 600.00 700.00 700.00 800.00 800.00 | 418 499 4891 486 447 402 3334 2265 1196 1367 10785 0050 0010 00207 0118 | .575 .5041 .388 .360 .296 .2515 .1747 .1237 .0629 .0319 .0119 .0118 | .537 .4694 .348 .3418 .2610 .21705 .1308 .0783 .0018 .00169 .00169 .00169 .00169 | .515 .428 .392 .350 .233 .193 .152 .123 .090 .063 .007 -008 -008 -004 -008 -1143 -1186 | .506 .448 .368 .313 .226 .174 .226 .1067 .0067 .0078 -0078 -0078 -1134 -1152 -1134 -1152 | .501 .443 .349 .293 .293 .207 .150 .015 .019 -017 .060 -108 -151 -189 -221 -231 -247 -2847 -2847 -3844 -3841 | .473 .4134 .2638 .2017 0077 12006 2316 2306 33005 33002 2995 2995 29961 2305 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| Ī | | | | PRESSU | RE COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|--|--|---|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.6 | $\alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 7.50 7.50 20.00 15.00 30.00 30.00 40.00 50.00 60.00 65.00 85.00 85.00 95.00 | . 2 2 0 . 0 5 9 . 0 4 2 . 0 2 8 . 0 0 5 . 0 136 . 0 724 . 0 646 . 0 704 . 0 500 . 0 94 . 1 312 . 5 5 9 9 . 3 2 4 7 . 1 5 8 | 48 48 412 46 202 437 68 68 017 32 2 | . 439 . 002 - 009 - 0033 - 033 - 023 - 021 - 021 - 021 - 025 - 021 - 023 - 021 - 023 - 033 - 033 - 033 - 033 - 033 - 033 - 033 - 035 - 035 | . 692 . 074 . 051 . 027 . 023 . 004 . 010 . 023 . 004 . 073 . 108 . 166 . 243 . 334 . 334 . 334 . 3437 . 284 . 288 . 293 . 299 | .441 .1893 .075 .034 .0326 .0341 .056 .0931 .1252 .3232 .3230 .2332 .2334 .238 | . 4 2 4 .276 .131 .102 .070 .036 .036 .036 .036 .036 .044 .109 .271 .244 .1454 .1454 | .315 .172 .117 .058 .020 .026 .026 .034 .034 .034 .034 .039 .039 .031 .050 .050 .050 .050 .050 .050 .050 .05 |
| LOWER SURFACE | 1.25 2.500 7.500 10.00 15.00 25.000 35.000 45.000 45.000 50.000 55.000 655.000 70.000 75.000 80.000 90.000 | . 12697 - 00312 - 00951 - 113497 - 12574 - 12574 - 12979 - 22791 - 22791 - 00882 | | 2231581459145916819221022122223223372422134916661753 | 205 207 217 223 234 234 233 234 152 154 168 168 | - 4116 - 2831 - 2231 - 2212 - 22126 - 2236 - 2338 - 2441 - 2330 - 1557 - 11597 - 11597 - 1262 | - 471 - 307 - 363 - 230 - 1992 - 2219 - 2216 - 2218 - 2218 - 2213 - 201 - 178 - 160 - 1122 - 114 - 177 | 491 2565 22339 221488 1788 1798 1662 1550 1121 1669 09935 0095 |
| | M = 0 | | | · · · · · · · · · · · · · · · · · · · | _ | | | |
| UPPER SURFACE | .00 1.25 5.00 7.50 10.00 20.00 30.00 40.00 45.00 60.00 60.00 60.00 70.00 75.00 90.00 | - 183 - 164 - 164 - 061 - 096 - 218 - 651 - 467 - 294 - 188 | 9137 3939 3336588 23126588 2312658 211261 103222 1611 118128 669427 669427 659427 | | | | | . 001 - 624 - 2242 - 2242 - 2332 - 1291 - 0059 - 0049 - 00 |
| LOWER SURFACE | 1.25 2.50 5:00 10:00 20:00 25:00 30:00 35:00 45:00 65:00 70:00 85:00 95:00 GAP | . 12: .06: .02: .00: .00: .00: .00: .00: .00: .10: .10 | 256 -184 -197 -1005 -1005 -1005 -1005 -1006 -1006 -110 | | | | | .245 .1692 .0029 .0065 .0089 .0089 .1027 .1115 .1117 .0083 .0069 .00633 .0023 |

TABLE III

| | PERCENT | | | PRESSUR | E COEFFICIENT, | P, AT: | | ·- · |
|---------------|---|--|--|--|--|--|--|--|
| | CHORD | 0.135b/2 | 0.25ь/2 | . 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.6 | $\alpha = 6.0^{\circ}$ | | - | | | | |
| UPPER SURFACE | 12.5.50000000000000000000000000000000000 | 158 - 583 - 498 - 498 - 3349 - 3349 - 328 - 328 - 328 - 1128 - 228 - 1128 - 228 - 128 - 228 - 128 - 333 - 328 - 328 | 815 - 1.280 922 691 570 496 403 350 266 266 266 163 088 103 103 103 103 707 711 6705 509 | 874 - 1.165 918 753 509 400 310 249 194 086 020 020 114 086 020 114 164 151 164 347 347 345 347 353 | 290 759 629 629 543 374 296 230 175 073 015 037 .087 .087 .087 .087 .087 .087 .087 .087 .087 .087 .087 .087 .088 | 896 9154 6133 6133 4494 257 0465 0465 0465 1288 1457 22355 2333 | 949 - 1.1066895184503903371751380800340057113112150143127125 | 738 - 1.0866865048326165121165073053042073054205370568068069 |
| LOWER SURFACE | 1 . 25 | .363 .342 .3062 .2239 .2250 .1355 .0552 .0037 0037 1001 1001 0046 | .442 .3644 .2934 .23097 .1370 .0602 .00266 .00266 .1265 .10660 .10660 .793 | . 455 .385 .296 .2386 .197 .146 .052 .017 0067 007 007 1043 1043 1043 105 | . 423 .357 .240 .194 .125 .083 .047 014 037 084 084 084 084 084 116 134 | . 442 .366 .275 .219 .174 .118 .037 .004 046 072 .081 114 136 078 078 078 078 078 078 078 | .4362 .257 .198 .170 .052 .016 051 051 093 120 120 129 084 087 087 087 087 | .391 .305 .307 .128 .076 .0032 0579 0114 1066 1064 081 083 0637 0637 045 |
| İ | M = 0.6 | $\alpha = 8.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 25.00 35.00 35.00 45.00 65.00 65.00 75.00 85.00 95.00 | .105 6516 5812 5504 425 425 4380 3380 3380 2965 1659 1659 1659 7474 5378 | - 1.171 - 1.080 - 1.0149263812269457147739732727660052014019770026055485 | - 1.398 - 1.18106 - 1.0019108184428637991811340840024073534553455 | 441 783 712 666 541 467 366 3049 180 124 0713 026 3114 | - 1.0869309138918872861808729585386234107036124196254263263231 | 71455204974834163714398417198198171154132083282934543826187 | 798640357555575557383928661381381101110112411261261261093413 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 25.00 25.00 35.00 45.00 50.00 50.00 60.00 70.00 80.00 90.00 | .389 .3877 .3770 .3258 .2297 .2162 .1173 .0703 .0144 .0156 0164 .0164 .0163 | .480 .437 .369 .314 .282 .2142 .131 .094 .059 .031 .0052 .081 .032 .081 .030 .031 | . 471 . 4740 . 362 . 315 . 272 . 2161 . 114 . 077 . 0049 - 0033 - 0755 - 118 . 0555 - 118 - 10867 - 10967 - 10 | . 493 . 381 . 381 . 277 . 205 . 1079 . 0048 - 00103 - 00647 - 105 - 0042 - 00453 - 1086 - 1086 - 1086 - 1086 - 1086 | . 476 . 430 . 356 . 2295 . 246 . 184 . 0949 . 0349 . 0056 0756 0756 0756 0766 0762 084 084 - 084 - 084 | . 454 . 412 . 322 . 266 . 238 . 164 . 109 . 0057 031 055 077 101 101 055 051 071 068 068 068 | .4155 .264 .196 .1414 .0099 .0144 0677 0667 0669 0569 0568 0377 0280 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| ſ | | | | PRESSURE | COEFFICIENT, | P, AT: | | |
|---------------|--|--|--|---|--|---|--|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.6 | 0 α = 11.3° | | | | | | |
| UPPER SURFACE | 1250 2500 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.00000 10. | .009 -1.041 -1.110 -1.95898489946556455644113588245108810881088108810881088 | - 1.708 - 1.344 - 1.356 - 1.368 - 1.404 - 1.467 - 1.403 - 1.217925624314216120012058021010681666609490 | - 1.506 - 1.072 - 1.060 - 1.060 - 1.030 - 1.019 - 930 - 886 - 776 - 629 - 555 - 476 - 3575 - 406 - 3571 - 401 - 4483 - 476 - 372 | - 988 - 904 - 893 - 8869 - 8857 - 8869 - 857 - 834 - 764 - 7764 - 7764 - 764 - 401 - 451 - 461 - 412 - 466 - 451 - 412 - 497 - 497 - 497 - 399 | 645 5772 5772 55677 55677 55677 5531 551242 5023 4470 4470 44718 34871 34871 34871 34871 34871 34871 34871 34871 34871 34871 | - 454 - 489 - 375 - 3752 - 3541 - 3322 - 3341 - 3322 - 3311 - 3322 - 2647 - 2236 - 2246 - 224 | 3994 2994 29774 226617 2236617 223693 22159 221093 220093 118825 1176995 |
| LOWER SURFACE | 1.25 2.500 7.500 10.00 25.00 25.00 35.00 40.00 50.00 60.00 65.00 75.00 65.00 75.00 65.00 75.00 | .390 .377 .560 .498 .463 .432 .361 .324 .228 .192 .153 .097 .053 .020 -063 .020 -063 .030 | . 497 . 524 . 490 . 4402 . 407 . 3388 . 236 . 159 . 1084 . 0523 . 0059 . 0064 . 0064 . 0061 . 779 | . 468 . 506 . 462 . 417 . 381 . 260 . 214 . 177 . 139 . 103 . 076 . 041 . 015 . 009 . 038 . 038 | .504 .497 .417 .417 .370 .297 .1157 .1157 .0185 .022 .008 033 .0073 .0015 068 073 | .490 .4833 .3873 .3275 .220 .1714 .0992 .0321 0520 0336 090 0499 | . 482 . 485 . 392 . 3338 . 3137 . 175 . 1086 . 0049 . 0072 . 0093 . 0093 . 0064 . 1266 . 1266 . 1266 . 1266 | .413 .3790 .218090 .218095 .01167 .00157 .00167 .00881 .00991 .00991 |
| | M = 0 | | | | 0.7.0 | 1 - 483 | 423 | 548 |
| UPPER SURFACE | 1.00 1.25 2.50 5.00 10.00 25.00 25.00 35.00 35.00 45.00 45.00 65.00 65.00 70.00 85.00 95.00 | 571 446 348 | - 1.951 - 1.8887 - 1.99517 - 1.99793 - 1.95517 - 1.94535 - 1.66768 - 1.407749 - 1.37343 - 1.6768 - 1.4087 - 1.537313 - 1.6868 - 1.6868 - 1.6876 - 1 | - 1.140 - 1.101 - 1.097 - 1.096 - 1.097 - 1.075 - 1 | - 879 - 6988 - 6882 - 6677 - 6475 - 6444 - 6333 - 6081 - 55030 - 4708 - 5221 - 4991 - 486 | - 483 - 4799 - 4886 - 4887 - 4991 - 4887 - 4991 - 4882 - 4467 - 4582 - 44582 - 4437 - 44397 - 43975 - 3975 - 3975 - 3975 - 3975 | 420 411 406 413 413 413 407 407 407 397 377 3816 3726 3726 3727 327 317 | - 483 - 477 - 4767 - 4769 - 4775 - 4773 - 4737 - 4594 - 4459 - 4435 - 4435 - 4214 - 4214 - 4214 - 4359 - 3376 - 3376 - 3310 |
| LOWER SURFACE | 55.00 | 311 512 610 595 570 570 456 407 351 306 277 212 211 158 1158 1158 1158 1158 1158 1158 1158 1158 1158 1158 1178 11 | .436 .553 .580 .587 .521 .453 .453 .304 .260 .231 .185 .144 .116 .074 .008 .247 .120 .085 .038 .038 | . 415 . 532 . 533 . 506 . 475 . 420 . 361 . 226 . 122 . 154 . 1154 . 1154 . 1154 . 120 . 020 . 027 . 027 | . 480 . 536 . 532 . 494 . 461 . 388 . 334 . 285 . 249 . 201 . 166 . 124 . 086 . 047 . 005 . 005 | | . 483 . 436 . 374 . 299 . 234 . 181 . 137 . 097 . 0047 . 0047 . 0061 . 0077 . 105 . 105 . 153 . 153 . 336 | .3123 .200 .10552 0559 10752 17753 17753 |

TABLE III

| Ţ | | | - | PRESSU | RE COEFFICIENT | , P, AT: | | |
|---------------|--|--|---|---|---|---|--|---|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85ъ/2 | 0.95b/2 |
| | M = 0.6 | 0 α = 19.5° | | | | | 1 | |
| UPPER SURFACE | 1250 250 250 250 250 250 250 250 250 250 | | 743 781 753 722 680 | 1.014354 .994354 .994354 .9994 .9944 .9944 .9944 .9944 .9944 .9944 .9944 .9944 .9944 .9944 .9944 .9944 .9944 | 646 550 557 548 531 515 | | 537 527 518 519 519 519 5499 490 4871 475 475 5249 5249 5365 3864 387 | |
| LOWER SURFACE | 1.25 2.500 7.500 10.500 15.000 225.000 30.000 40.000 50.000 65.000 75.000 85.000 95.000 | . 563 . 700 . 673 . 646 . 5536 . 481 . 482 . 348 . 279 | . 427 .576 .614 .599 .5067 .445 .327 .247 .191 .1022 .247 .1034 -0593 -1872 | .355 .559 .547 .508 .464 .3551 .267 .282 .138 .096 .0038 .0038 .005 .005 .005 .005 .005 .005 .005 .00 | .545 .514 .487 .375 .326 .287 | . 383 . 494 . 508 . 486 . 455 . 402 . 334 . 295 . 203 . 161 . 115 . 072 . 039 017 038 109 207 | .488 .476 .426 .357 .2243 .1933 .0953 .0124 | . 3 14 . 397 . 344 . 297 . 208 . 078 . 0035 . 0045 . 0055 . 0055 . 0085 . 1085 . 1018 . 128 . 1417 . 163 |
| | M = | 0.60 $\alpha = 25.9$ | | | 829 | 706 | 607 | 505 |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 20.00 30.00 35.00 40.00 40.00 60.00 60.00 70.00 80.00 95.00 | 8648 8648 86713 86713 8733 8753 8753 8753 8753 8753 8753 7793 77983 | 87802094369436943699228655351333990 | | - 798 - 799 - 799 - 798 - 789 - 789 - 785 - 765 - 7765 - 778 - 805 - 7747 - 633 | 702 6998 6998 69867 66867 66755 6673 6731 7195 7195 75196 55482 5248 | 59885378 558878 55878 5557468453 55468453 55468453 55468453 55468453 55468453 54683 4453 4433 4 | 494 484 486 487 477 487 483 481 481 461 475 452 453 453 394 3375 3375 |
| LOWER SURFACE | 55.00 | .15823 .7243 .7441 .6641 .5944 .496 .3165 .3165 .3165 .3175 .3175 .3175 | .5705 .6656 .6657 .6514 .5790 .4819 .3945 .2854 .2914 .1901 .3038 .2419 .3138 .34819 .34916 | 475 574 574 574 574 574 574 416 7327 223 111 | 4 417 | 450 534 530 514 476 436 436 436 436 436 436 436 43 | . 453 . 493 . 493 . 488 . 327 . 225 . 225 . 225 . 1723 . 001 . 001 | .383 .4158 .3548 .272 .209 .1087 .0087 .0023 .0023 .0069 .1011 .118 .118 .118 .129 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | | | | PRESSU | RE COEFFICIENT | P, AT: | | |
|---------------|---|--|---|---|--|--|---|--|
| _ | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 85 α = 0.0° | * · · · · · · · · · · · · · · · · · · · | | | L | | 1 |
| UPPER SURFACE | 0 5 5 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 | 2 51 . 118 . 0864 . 0 632 . 0 207 . 0 479 . 0 774 . 0 556 . 1256 . 2337 . 68625 . 4154 . 240 | .51805704460570700780760830762083010174330023116611525 | . 477 036 0249 0245 0386 0386 0141 0012 0604 1786 3569 3787 3885 | .764 .019 .011 -0014 .017 .015 .021 .025 .025 .028 .128 .128 .135 .256 .337 .298 -298 -298 -307 | .4788 .1688 .0885 .00392 .0231 .0231 .0245 .0912 .1189 .37312 .3251 -3251 -3251 -3251 -3251 | . 465 . 275 . 120 . 0657 . 0313 . 035 . 035 . 035 . 0368 . 1051 . 2252 . 2373 . 1874 - 11646 - 1152 | .3899 .11555 .005577 .005772 .005772 .005772 .005773 .005703 .005703 .005703 .005703 .005703 .005703 .005703 .005703 .005703 .005703 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.000 25.000 35.000 45.000 45.000 60.000 70.000 80.000 90.000 90.000 | . 183 . 1068 . 00347 . 00747 . 00747 . 1348 12099 2368 2368 2368 2368 2467 2467 2467 2467 2467 | 102076068087096117155173196214243285327336001147756 | 169123134169214236227281265265163163201201 | | 407 3048 23441 23540 22601 23781 23909 23884 2088 2088 21899 21899 21979 | | |
| | M = 0.8 | | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 25.00 35.00 35.00 45.00 45.00 65.00 65.00 65.00 85.00 95.00 | 2 47 2514 1877 287 | 647 595 519 | 001 - 1.009 825 437 374 341 273 172 172 087 087 085 085 176 176 176 176 176 176 176 176 176 176 178 180 380 380 384 384 | .27793071740933327714610005700905110915421222929930023002 | 106841724358326291208172131095048006 .064120 .158173182241241241 | 081884668325292212156119081040 .009 .063 .114 .145 .151168158158151 | .06679462731922621681391075006100450045004900530056 |
| LOWER SURFACE | 2.50 5.00 7.50 10.00 20.00 25.00 35.00 40.00 45.00 55.00 60.00 65.00 75.00 85.00 90.00 85.00 90.00 85.00 90.00 85.00 | .336 .374 .217 .178 .143 .087 .085 .0017 .017 .0193 .158 .223 .061 .158 .061 .158 .061 | .33 2 .19 5 .19 5 .11 40 5 .00 31 1 .00 29 9 .00 8 2 .11 46 .16 9 .11 46 .16 9 .23 8 .00 48 2 .00 76 6 .00 76 6 .00 75 5 | .345 .269 .183 .191 .051 .028 .028 .026 .116 .1156 .156 .156 .156 .162 .172 .172 .198 .099 .105 .131 .161 .162 | .330 .255 .204 .142 .105 .027 .006 .008 .065 .093 .119 .163 .177 .188 .207 .099 .138 .164 .164 .164 .164 | .323 .144 .155 .100 .064 .023 .055 .086 .113 .137 .163 .1137 .191 .199 .1154 .158 .163 .158 | .320 .243 .137 .086 .071 .0044 081 1111 149 171 189 204 209 209 209 191 124 126 126 | . 255 . 1090 . 0117 . 1235 1146 1166 1166 1166 1163 1136 1136 |

TABLE III

| | | | | PRESSUE | RE COEFFICIENT, | P, AT: | | |
|---------------|---|--|--|---|--|--|---|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | $85 \alpha = 6.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 10000000000000000000000000000000000000 | 2 18026 2 1802 | - 1.041 - 1.041 - 1.0204 - 1.0204 | - 2999 - 1.2035 9791 6799 4652 22161 0442 0442 0442 0442 0428 0428 3744 38813 3886 | . 128 731 680 523 5716 443 2887 1230 0676 0397 078 108 317 317 | 474 807 7629 6856 573 478 3651 1551 0047 1147 1561 2559 2660 2660 | 37862055344351361526421365136608850690069020101978154 | 304776644655144533453183310830708077308920975 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 25.000 25.000 45.000 45.000 65.000 70.000 80.000 90.000 90.000 90.000 | 390 390 3352 3352 32752 11603 10750 005278 005278 005278 1005279 1005278 10052 | . 443 .3693 .2940 .1484 .1484 .0687 .00257 .00257 .11487 .1187 .0215 .1187 .0215 .1187 | . 448 .3786 .3866 .3337 .1452 .0517 .0141 .0041 .0045 .0957 .1266 .0972 .1266 .0722 .0722 .0723 .0723 .0723 .0735 .0736 | . 447 .357 .305 .249 .228 .086 .045 .019 .019 .019 .119 .1162 .096 .096 .098 .1345 .145 | . 434 .362 .269 .209 .108 .0060 .021 014 043 104 124 124 157 157 157 | . 424 .344 .244 .1869 .0978 .0058 .0043 .0141 .1141 .1160 .1 | . 377 . 291 . 204 . 1362 0428 1131 1131 1136 1136 1098 0955 0075 00766 0044 |
| | M = 0. | | | - | | | | |
| UPPER SURFACE | 105.000 12.500 105.000 | 267466906658796128199980889833 0 | - 1.147 - 1.147 - 1.147 - 1.1055 - 1.0553 73917 32788 32788 32788 3278 3278 3278 32788 32788 32788 32788 32788 | 522 9056 8375 7232 5556 4428 3675 4284 3555 4284 3555 4284 3555 4284 3555 31495 314999 3381 | 039 7417 6531 5887 5887 5321 3552 13633 13633 37427 37427 37427 37427 | | 634 5743 4989 4767 4401 4110 4115 4121 4211 | 544 6518 5997 5997 5592 4873 2689 188 209 189 189 1992 1992 1160 1437 |
| LOWER SURFACE | 1.25 5.50 10.000 15.000 25.000 25.000 45.000 45.000 45.000 65.000 80.000 80.000 90.000 | 4 10 4 38 4 38 3 350 3 2 255 2 2 255 1 3 3 9 0 0 7 6 0 0 1 4 - 0 0 5 9 - 1 1 3 2 - 0 0 2 9 - 0 2 9 - 0 2 9 | .506 .445 .376 .376 .292 .214 .170 .131 .094 .0630 0045 1053 1060 1050 1060 1060 1060 | . 493 . 493 . 347 . 298 . 261 . 203 . 105 . 0067 . 009 . 018 . 0053 . 0070 . 090 . 130 . 047 . 047 . 047 . 118 . 118 . 118 . 118 . 118 | . 497 . 412 . 368 . 3168 . 189 . 142 . 098 . 032 . 030 - 060 - 060 - 062 - 135 - 063 - 129 - 129 - 129 | .475 .421 .342 .282 .240 .178 .082 .087 .016 019 073 135 020 1135 078 078 078 078 | . 473 . 414 . 323 . 266 . 243 . 168 . 107 . 059 . 026 - 026 - 121 - 137 - 147 - 147 - 018 - 059 - 069 - 069 - 069 | .421 .269 .188 .133 .029 069 069 111 120 117 110 097 087 067 060 |

TABLE III

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | · · · · · · · · · · · · · · · · · · · | |
|---------------|--|---|--|--|--|---|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.8 | $\alpha = 11.3^{\circ}$ | | | | | | |
| UPPER SURFACE | 12.50 57.500 105.000 1 | 177 - 4522 - 7744 - 7756 - 6731 - 6625 - 731 - 6621 - 5928 - 593 - 3459 - 012 - 230 - 998 - 568 - 568 - 568 - 568 - 568 - 568 | 912 1.476 1.458 1.480 1.480 1.452 1.377 1.341 1.240 1.452 0.438 0.411 0.040 0.042 0.048 0.076 0.076 0.076 0.076 | 983 - 1.135 - 1.084 - 1.092 - 1.061 - 1.033 983 883 995 907 864 914 748 655 487 447 475 439 | - 433 - 812 - 796 - 796 - 798 - 785 - 7780 - 7879 - 7879 - 769 - 741 - 741 - 7551 - 5513 - 5513 - 5523 - 5549 | 622 5338 5326 5226 55254 55051 4484 4474 4451 4451 4414 3999 3566 33866 | 466 396 379 368 376 376 376 363 357 345 345 316 316 329 272 287 287 287 | 500394395406401399374367347347326288428842662622659259 |
| LOWER SURFACE | 1 . 25 2 . 50 5 . 000 10 . 000 15 . 000 25 . 000 35 . 000 45 . 000 55 . 000 65 . 000 70 . 000 80 . 000 90 . 000 90 . 000 GAP | 4 22 4 69 4 530 5 39 4 52 3 74 3 40 2 39 - 172 - 111 - 00 45 - 077 - 0749 - 00 44 | .567 .544 .489 .437 .406 .334 .240 .2200 .161 .1088 .047 .018 .028 .047 .074 .074 .074 .075 .074 | .516 .506 .447 .404 .365 .305 .250 .206 .1166 .1315 .0638 .0219 078 .018 .018 .018 .018 .018 .018 .018 .01 | .532 .504 .462 .412 .362 .2942 .1956 .117 .0044 .0024 .0059 .0035 .0035 .0035 .0044 .0035 .0044 .0035 | .508 .487 .375 .375 .216 .170 .131 .094 .018 012 047 072 120 120 120 1203 203 203 | . 489 . 468 . 389 . 340 . 318 . 173 . 121 . 0076 . 0026 - 0158 - 1058 - 1154 - 1154 - 1152 - 1186 - | .445 .402 .3267 .1995 .0995 0995 1249 1587 167 174 174 172 174 174 174 174 174 |
| | M = 0. | | 1 000 | 1 042 | 939 | 543 | 482 | 532 |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 25.00 35.00 35.00 45.00 55.00 65.00 65.00 75.00 885.00 | 0 8 9 2 6 8 5 6 5 6 5 6 5 6 5 6 5 6 5 6 5 6 5 6 | - 1.282 - 1.441 - 1.391 - 1.398 - 1.2699 - 1.1846 - 1.1846 | - 1.042 978 - 1.0011 - 1.0008 972 9313 8940 847 847 7737 6653 6653 628 628 | - 829 - 6624 - 6443 - 6624 - 6169 - 6114 - 6093 - 5988 - 55731 - 5528 - 55328 - 55328 - 55328 - 55328 - 55328 - 55321 | 5435 5400 542 5425 545 545 545 5429 537 5329 5211 5023 4905 4545 4546 4546 4450 4466 | 482 471 471 473 473 473 463 464 459 459 459 459 459 459 459 459 459 459 473 473 464 459 398 398 398 373 463 | |
| LOWER SURFACE | 1.25 2.50 10.00 15.00 20.00 25.00 30.00 45.00 55.00 60.00 80.00 80.00 80.00 90.00 | 3 547 6 6952 6 6019 5 4765 4 268 3 3 3 3 6 3 2 3 2 5 1 2 3 2 6 0 3 2 9 0 1 6 5 1 1 3 0 0 4 3 | .584 .619 .598 .5528 .461 .414 .363 .321 .242 .199 .1519 .072 .012 .012 | .505 .505 .505 .506 .477 .362 .311 .229 .154 .154 .046 .045 .046 017 093 162 751 | .577 .535 .459 .459 .458 .2838 .2838 .195 .106 .028 .019 .079 .079 .095 .0010 .1210 .2110 .2110 | .507 .477 .3997 .288 .2393 .150 .111 .0611 | . 492 . 445 . 401 . 370 . 240 . 185 . 132 . 080 . 018 . 0168 . 1143 . 161 . 083 . 1243 . 1243 . 2462 | . 427 .380 .310 .263 .1045 .0048 .1151 .1786 .2007 .217 .2217 .2217 .2217 .2249 .249 .249 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| Ī | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|---|--|--|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.8 | 5 α = 19.6° | | | | | | |
| UPPER SURFACE | 25.000 1.25.000 70.000 15.000 25.0000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.0000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.0000 25.000 25.00 | 028683842 - 1.092 - 1.075 - 1.065 - 1.0199619678497716996554557 | 1.0052228866336 1.0052228866336 1.0021288666336 1.002128866636 1.002128866636 1.002128866636 1.002128866636 1.002128866636 1.002128866636 1.002128866636 1.002128866636 1.002128866636 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.002128866 1.0021288666 1.0021288666 1.0021288666 1.0021888666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.0021288666 1.0021288666 1.0021288666 1.0021288666 1.002188666 1.002188666 1.002188666 1.002188666 1.002188666 1.0021886666 1.002188666 1.002188666 1.002188666 1.002188666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.0021866666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.00218666 1.0021866 1.00218666 1.00218666 1.0021866 1.0021866 1.00218666 1.0021866 1.0021866 1.002186 | 953 867 834 846 837 827 827 819 794 749 674 668 670 6670 6670 6670 6670 6670 | 821 | 650 6463 640 640 637 6325 6233 6233 6233 6004 5998 5999 5317 5316 | 578 - 5770 - 5571 - 5571 - 5571 - 5574 - 5554 - 5525 - 5529 - 5539 - 56867 - 6899 - 4436 - 4448 | 5686311525531525531525515899188760115899188760118044884651180 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 15.000 25.000 30.000 30.000 40.000 50.000 60.000 70.000 80.000 90.000 95.000 | 2 4 5 9 1 4 4 5 9 1 4 4 5 9 1 4 4 5 9 1 4 7 7 8 4 5 0 7 7 4 0 0 1 5 7 7 2 3 5 7 6 7 6 7 6 7 6 7 6 7 6 7 6 7 6 7 6 7 | .569 .6652 .6615 .5401 .5407 .4507 .2827 .2290 .11418 .068 .14477 .0118 .7132 | .470 .585 .589 .4895 .3997 .3060 .22193 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .1131 .009 .009 .009 .009 .009 .009 .009 .00 | . 5567 . 5573 . 5573 . 533 . 534 . 4464 . 3472 . 2562 . 1169 . 1071 . 0051 . 0171 . 01 | .426 .507 .5125 .4852 .4952 .3055 .2116 .1181 .0028 -0195 .00586 -1881 -1881 -1881 -1881 -1881 -1881 -1881 -1881 | . 4097 . 4828 . 4448 . 3666 . 35066 . 1256 . 1092 . 0019 - 0019 - 1140 - 1177 - 1177 - 2716 - 292 - 414 | .387 .4169 .3183 .2117 .0045 0045 1147 1147 12150 22554 2254 2254 2254 |
| | M = 0. | | 903 | 825 | 773 | 731 | 869 | 624 |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 55.00 65.00 65.00 65.00 75.00 85.00 95.00 | 774 758 747 726 | - 8002 - 8007 - 80066 - 880632 - 88128 - 88128 - 88314 - 88314 - 879812 - 777794 - 777700 | 8099 89982 800779931 79939 77999 77779 77794 77533397 77222 | 754 7551 7551 7557 7443 7443 7443 7510 7804 7315 6884 6872 68864 6792 | 731 729 726 726 726 726 719 719 718 733 745 727 724 629 629 623 630 | 658 6427 6333 6341 6149 615 615 615 615 632 6532 716 7147 734 5549 5421 5421 | 6097 60951 65991 55981 55991 55994 55984 55884 55884 55884 588487 55884 58882 |
| LOWER SURFACE | 50.00 | . 733 .815 .815 .7504 .606 .562 .523 .443 .3346 .230 .230 .230 .237 | .51 2 .6777 .7362 .6756 .5906 .59463 .4159 .4159 .3668 .2682 .192 .2682 .193 .2758 .1348 .00750 | .511 .470 .429 .384 .340 .294 .294 | .556 .619 .590 .582 .543 | . 479 . 5443 . 5522 . 448 . 496 . 318 . 275 . 1773 | | .404 .442 .3895 .198 .130 .059 051 051 124 160 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|--|--|--|---|---|
| <u></u> | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 90 α = 0.0° | | <u> </u> | | | | |
| UPPER SURFACE | 1.250 5.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.00000 10.0000 10.0000 10.0000 10.0000 10.00000 10.00000 10.00000 10.00000 10.00000 10.00000 10.00000 10.00000 10.00000 10.00000 10.000000 10.000000 10.00000000 | .2 67 .2 049 .1214 .0 223 0 143 0 679 0 743 0 743 0 743 0 124 .2 149 5 5836 4 378 4 378 3 72 | . 52 5 5 2 5 6 5 2 2 5 6 5 2 2 5 6 5 2 2 5 6 5 2 2 6 6 5 2 2 6 6 6 5 2 6 6 6 6 | - 481 - 0642 - 0379 - 0379 - 0503 - 0122 - 0124 - 0244 - 060 - 1108 - 1108 - 12867 - 2863 - 3887 - 493 - 493 - 4408 | .783 .011 .003 -007 .007 .004 -010 .003 .020 .042 .071 .118 .180 .265 .354 .340 .316 -316 | .484 .148 .065 .047 .034 .006 .009 .021 .033 .056 .083 .126 .187 .272 .320 .313 .242 .273 .272 .273 | . 469 .357 .1084 .0054 .0042 .0018 .0118 .0123 .036 .091 .1212 .222 .222 .222 .2198 .1198 .1198 | .345 .164 .1042 .0042 .00317 00824 00759 00463 00678 00775 00777 00877 |
| LOWER SURFACE | 1.250 5.500 10.000 15.000 25.000 35.000 45.000 45.000 50.000 70.000 80.000 90.000 90.000 | .206 .150 .150 .0024 -0036 -0056 -0046 -131 -131 -131 -128 -282 -449 -282 -449 -282 -183 -155 | 068047066077101122144162187204238263263402107105113156770 | 18 0033 1132 1132 1587 2134 2258 2258 2368 2258 | 212 169 107 144 165 2343 265 293 293 2897 2897 2890 238 227 | 362 2747 2513 2534 2534 2534 2534 2534 279 295 3128 3335 3284 2310 2205 | 436 328 2681 2689 2629 3133 3447 3166 2504 2507 1665 1547 1447 1447 1447 1447 | 533 413 313 3367 373 265 245 231 2330 2330 2330 231 235 245 - |
| | M = 0.9 | | | | | . ,,,,, | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 20.00 20.00 30.00 40.00 40.00 45.00 55.00 65.00 75.00 85.00 95.00 | 265 - 073 - 092 - 1141 - 1793 - 2237 - 2444 - 237 - 2444 - 237 - 2444 - 237 - 2464 - 247 - 198 - 060 - 141 - 7013 - 497 - 457 - 364 | .166842766401356342318305304274181086 .021 .174 .199 .176 .052723723672672 | .054 -1.004 -893 -514 -423 -380 -279 -2116 -168 -131 -090 -039 -026 -094 -1164 -169 -179 -399 -401 -395 | .349907784459356304260165078027036092 .141200 .190322321322 | 052 856 773 424 3148 236 191 1065 055 0023 .114 .1568 .171 .181 258 258 258 | 042 904 819 3305 281 275 178 058 058 058 058 058 146 141 146 141 172 181 1766 166 | .09182473733733333312014410840657066406670667072080089 |
| LOWER SURFACE | 1.25 2.550 5.000 10.000 15.000 25.000 35.000 45.000 50.000 60.000 60.000 60.000 80.000 95.000 95.000 95.000 | .347 .327 .3296 .233 .191 .156 .100 .094 .0018 .0018 .0031 .095 .1119 .1149 .1287 .085 .085 .084 .113 | .338 .264 .197 .151 .037 .0037 .0026 -0059 -1114 -1149 -1171 -2355 -066 -080 -1066 -155 -769 | .351 .268 .185 .138 .106 .0097 0304 092 123 1143 1170 1182 1170 1182 1179 1179 1179 1196 | . 334 .2503 .144 .107 .022 .0012 .0046 .1017 .127 .1152 .1152 .1187 .129 .165 .166 .198 .319 | .3249 .2449 .158 .101 .0063 .0226 .0063 .1221 .1577 .2177 .216 .2178 .1916 .1926 .1936 .1936 .1936 | .3289 .2449 .1442 .0978 .0077 090 123 1677 2247 2244 2335 1455 1455 1455 1455 1457 1457 | .267 .178 .1022 .0024 .148 .162 .191 .188 .183 .1733 .1135 .1135 .1136 .1169 |

TABLE III

| Ī | | | PRESSURE COEFFICIENT, P, AT: | | | | | | | |
|---------------|--|--|---|--|--|--|--|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | | |
| | M = 0.9 | $\alpha = 6.0^{\circ}$ | | | | | | | | |
| UPPER SURFACE | 1.5.00 7.500 1.5.000 25.000 25.000 25.000 45.000 45.000 65.000 85.000 85.000 | 329 | 093972962281271749244127388155301187397626 | 195 - 1.352 - 1.19017245455720641173026306230623081341704146399 | . 191 - 1.187 - 1.175 - 1.0868978411271270169125017017181321321321 | 4 2 6 8 2 9 3 8 7 6 0 1 7 6 2 9 2 9 4 2 9 3 6 1 9 1 6 0 4 6 0 6 4 5 1 0 4 6 0 6 4 5 1 0 4 6 2 6 6 9 2 6 6 9 2 6 5 9 | 318636446441233742243241819271247127127127127127127127127127127127127127127 | 266873724598561526434301126109104113109104113109107101094083 | | |
| LOWER SURFACE | 1.25 25.500 7.500 10.000 25.000 25.000 35.000 40.000 55.000 65.000 75.000 85.000 95.000 95.000 | . 4 01 . 3 978 . 3 2 79 . 2 2 4 4 2 . 1 6 9 . 1 6 9 . 1 7 6 . 0 2 5 7 . 0 2 | . 47 .3799 .2481 .1540 .0742 .0046 .0055 .0915 .1711 .1200 .00115 .1711 .1021 .1021 .1021 .1046 | .440 .36766 .219134 .01934 .00403 .00403 .005799 .11059 .11059 .11059 .11059 .11059 .11059 .11059 .11059 .11059 .11059 .11059 | . 431 .337 .239 .200 .1200 .078 .037 .005 - 062 - 062 - 1136 - 1156 - 1156 - 1156 - 1157 - 11 | . 432 .359 .270 .211 .168 .114 .059 .020 017 082 114 137 163 163 162 153 153 177 293 | . 425 .344 .186 .244 .186 .098 .035 .013 .052 097 136 218 218 219 1194 | .371 .284 .196 .116 .052 .052 .176 .177 .1765 .1751 .1171 .1174 .1084 .074 .0045 | | |
| | M = 0. | | | | 056 | - 494 | 577 | 507 | | |
| UPPER SURFACE | 1.25 2.5,0 5.00 7.50 10.00 20.00 30.00 40.00 40.00 40.00 60.00 60.00 70.00 75.00 90.00 | 445 424 415 421 413 403 402 461 290 120 127 667 557 471 444 | 408 081 121 105 002 721 721 666 658 | - 1.308 - 1.218 - 1.1179777728047682926181013201320132013201320132014484477448 | - 1.301 - 1.2290 - 1.223 - 1.422 999 774 656 557 404 279 201 036 036 036 036 036 3300 3304 3304 3304 | 267 | - 577 - 528 - 506 - 506 - 490 - 469 - 4450 - 431 - 408 - 380 - 380 - 3618 - 307 - 3618 - 307 - 327 - 327 | 700 661 631 636 645 547 425 235 209 209 208 214 190 175 167 | | |
| LOWER SURFACE | 1 . 25 2 . 50 7 . 50 10 . 00 25 . 00 35 . 00 40 . 00 45 . 00 55 . 00 60 . 00 75 . 00 85 . 00 95 . 00 | .4443 .419 .371 .2456 .2466 .1945 .122 .091 .029 .033 .000 041 089 .1609 01609 | .456 .3388 .332 .2315 .186 .110 .073 .044 .007 033 .044 .106 147 .136 | . 428 .350 .298 .2623 .148 .068 .0035 .0027 027 057 0769 138 | . 475 . 398 . 357 . 301 . 260 . 185 . 093 . 060 . 021 043 074 097 154 089 1357 089 137 304 | . 420 .3355 .279 .238 .177 .121 .004 030 017 117 1156 056 111 127 127 127 | .410 .314 .260 .238 .159 .094 .004 081 125 125 160 182 191 174 036 075 075 | . 3517 .1887 .1017 - 0407 - 11452 1552 1562 1142 127 1144 1053 0985 0862 | | |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| Ī | | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | |
|---------------|--|--|---|--|---|---|--|---|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40ኪ/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | | | |
| | M = 0 | $.90 \qquad \alpha = 11.3^{\circ}$ | | | | | | | | | |
| UPPER SURFACE | 00 1 25 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 | 2 2 3 | 715 - 1.326 - 1.3463 - 1.3463 - 1.3485 - 1.148 - 1.063365217029056040056748374837665665 | 71 - 1.147 - 1.060 - 1.051498998157686586586425914488517517508 | 971 9413 869 791 577 431 375 4473 473 454 | 767 678 672 6653 6553 6410 6249 592 592 5491 492 493 4470 4846 404 | 495 4419 4225 4225 4225 4235 33899 3365 3389 3365 3324 3215 2953 3102 297 | 445 3665 3665 35605 3551 3540 35226 35226 3 | | | |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 20.00 25.00 35.00 40.00 55.00 60.00 75.00 75.00 85.00 90.00 | 4 3 2 5 3 2 5 3 5 5 5 5 5 5 5 5 5 5 5 5 5 | .589 .5507 .4418 .346 .227 .253 .210 .1740 .1006 .024 024 027 .057 .063 063 063 063 | .535 .515 .451 .404 .368 .312 .207 .168 .028 .005 .008 .001 .001 .003 .009 .009 .009 .009 .009 .009 .009 | . 496 . 453 . 358 . 284 . 188 . 152 . 117 . 004 . 077 . 041 . 025 025 105 | .503 .487 .376 .333 .275 .216 .172 .131 .093 .057 .016 017 048 078 129 .046 051 135 135 130 | .500 480 480 487 354 .332 .251 .188 .135 .041 -008 -054 -096 -129 -157 -100 -127 -127 -217 -220 .274 | . 462 .348 .278 .2000 - 0045 - 1139 - 1167 - 1179 - 1189 - 1186 - 1186 - 1186 - 1186 | | | |
| | M = 0 | | | 1 075 | 951 | - 533 | T = 488 | - 535 | | | |
| UPPER SURFACE | .00 1.25 5.00 10.00 20.00 35.00 40.00 40.00 60.00 60.00 80.00 90.00 95.00 | 7734 697 621 546 462 367 267 055 - 1.160 821 691 559 | 528 659 929 940 746 536 | 956 951 951 955 953 920 921 829 879 883 755 680 6641 631 | - 741 - 730 - 724 - 721 - 695 - 687 - 6539 - 6539 - 6211 - 5312 - 55923 - 5569 - 5559 | 527 5322 548 5560 5560 5666 5549 5497 5494 4945 4945 487 | 470 472 473 477 426 442 | 466 461 456 449 437 443 | | | |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 20.400 25.00 30.00 45.00 55.00 60.00 70.00 80.00 90.00 95.00 | .5066 .7059 .66126 .5126 | .415 .364 .323 .283 .284 .300 .156 .117 .0070 .009 | .570 .541 .541 .361 .271 .271 .271 .271 .271 .271 .271 .27 | 5556 5379 487 4487 449 379 388 399 1159 1074 1140 1074 1140 | .512 .479 .440 .401 .349 .290 .2243 .197 .154 .0267 .0163 0163 0163 0164 | . 494 . 446 . 404 . 384 . 190 . 138 . 083 . 034 022 076 130 186 111 145 303 325 | .4392 .3929 .21645 .00645 .007315 .013904 .21341 .22667 .22667 .22667 .226809 | | | |

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| Γ | | | | | | F | RESSURE | COEF | FICIENT, | P, A | T: | | | | | |
|----------------|--|---|--|--|---|------|---|------|---|-----------------------|--|---------------------------|--------|---|------|---|
| | PERCENT CHORD | 0.1 | 35b/2 | 0.25 | b/2 | 0.40 | 0/2 | 0. | 55b/2 | 0. | 70b/2 | (|).85b, | /2 | 0.95 | 5b/2 |
| 1 | M = 0.9 | 10 0 | y = 19.8° | | | | | | ·T | | | T | | 588 | | .634 |
| | 1.25 2.500 70.000 15.000 25.000 35.000 35.000 40.000 35.000 70.000 80.000 70.000 80.000 90.000 | | 019 6024 1.130 | - 1 - 1 - 1 - 1 - 1 | .129 .1106 .094 .0086 .0083 .0 | | 942 8844 8854 8871 8771 8 | | 806 .705 .698 .698 .698 .698 .672 .666 .672 .674 .674 .675 .675 .675 .675 .675 .675 .675 .675 | | .647 .6440 .6339 .6336 .6336 .6233 .6223 .6223 .5396 | - | | 582 571 577 577 580 576 580 576 586 586 586 586 661 478 488 488 488 488 488 488 488 488 488 | - | .6001 .6001 .6001 .6001 .5001 |
| LOWER SURFACE | 1.25 2.500 5.500 7.500 10.000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.50000 2.50000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.5000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.500000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.50000 2.500000 2.500000 2.500000 2.50000000000 | | 2 6799 7 7315 7 727 6 4 0 1 0 2 4 7 0 2 4 4 5 0 6 4 1 5 2 1 4 9 2 6 2 4 7 0 2 4 8 2 7 6 1 2 6 2 6 2 6 2 6 2 6 2 6 2 6 2 6 2 6 | | 60 4 66 8 2 66 8 2 66 6 2 65 7 5 5 7 8 64 7 8 64 7 7 8 6 | - | .498 .596 .596 .583 .512 .457 .3729 .289 .207 .214 .207 .214 .208 .207 .044 .114 .232 .014 .014 .014 .014 .014 .014 .014 .014 | | . 578 . 585 . 548 . 518 . 418 . 418 | | 4 4 2 | 533552 | | .416 .502 .490 .459 .459 .379 .2652 .106 .053 .053 .053 .053 .139 .139 .139 .139 .139 .139 .139 .13 | | 4 4 3 7 . 4 6 1 3 . 4 6 1 3 . 4 6 1 3 . 6 1 6 1 . 6 1 8 1 8 1 8 1 8 1 8 1 8 1 8 1 8 1 8 1 |
| ! | GAP M = | | $\alpha = 26.2^{\circ}$ | _ - _ | .701_ | | | | | | 74 | 3 | _ | .961 | | . 628 |
| UPPER SURFACE | M = | | . 173 . 465 . 845 . 8325 . 811 . 7990 . 7997 . 7997 . 7997 . 7997 . 7997 | | .798 .775 .779 .772 .769 .762 | | 8 25 8 086 8 086 8 086 8 096 8 096 7 799 6 799 6 799 7 777 7 764 7 743 7 | 1 | 750 747 7747 7747 7746 7746 7747 7746 7747 7701 7701 7701 7701 7701 7701 | 35455228 | - 73 - 73 - 73 - 73 - 73 - 73 - 73 - 74 - 74 - 74 - 74 - 74 - 74 - 75 - 75 - 76 - 76 - 76 - 76 - 76 - 76 - 76 - 76 | 81333326246666969979941 | | 6819636363666366666666666666666666666666 | | 60033 60033 6003 6003 6003 6003 6003 60 |
| 1 OWER SURFACE | 1.85 5.05 10.00 15.00 20.00 35.00 45.00 | 500000000000000000000000000000000000000 | . 088 3802 8866 8866 8866 8866 8866 8866 8866 8 | 57.55.55.55.55.55.55.55.55.55.55.55.55.5 | .548 .7655 .7651 .7655 .6650 .5782 .4408 .35074 .41767 .21767 | | .419 .664 .676 .626 .5843 .5026 .4626 .4195 .3302 .2333 .157 | | . 48 .637 .610 .577 .526 .48 .440 .315 .216 .09 .210 .044 .042 .57 | 883807672846933 60429 | 1 | 2330633107561197 85379 | : | 291 4826 522 470 427 427 427 427 427 427 427 427 427 427 | | . 42 . 46 . 44 . 34 . 17 . 105 . 004 . 114 . 121 . 24 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | PERCENT | | | PRESSURI | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|--|---|---|---|---|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40ь/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0 | .94 $\alpha = -0.3^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 20.00 30.00 35.00 45.00 55.00 65.00 65.00 75.00 85.00 90.00 | 282 122 1102 083 0014 0018 0018 0019 0059 0072 0059 1074 1059 1059 1059 1059 1059 1059 1059 1059 | .539012012021042072085074085074401820536736779027902601 | - 4890230170160230080280280240250771272304286433425418 | .790 .048 .034 .025 .004 .013 .025 .004 .013 .024 .069 .169 .359 .359 .347 .318 .318 | . 482 .155 .073 .053 .027 .004 .009 .009 .022 .035 .057 .088 .131 .193 .274 .312 .309 .324 .277 .277 | . 462 .284 .128 .098 .061 .022 .030 .030 .030 .039 .0568 .1439 .2316 .2037 1914 184 | . 321 . 201 . 1370 . 0351 084 144 093 0752 098 098 080 080 128 128 128 133 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 25.00 35.00 40.00 45.00 50.00 50.00 70.00 85.00 90.00 85.00 75.00 | 7809952442233553158547611 2151090054922335531566 | 092 043 0660 1100 11410 1830 2357 2254 2354 2351 2311 2311 2311 2307 | 127 096 100 1170 1170 1280 1292 2242 2242 2317 449 2132 2152 2586 512 | | 371 257 259 263 267 269 328 346 3746 4478 4478 459 3506 296 247 240 247 240 297 | 484 3820 3065 3155 3776 4450 4450 4450 4506 311 1204 1204 1237 1237 | 633 510 343 3782 4014 416 400 3604 180 164 1144 1140 117 1094 105 |
| 1 1 | M = 0 | | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 15.00 25.00 35.00 45.00 50.00 50.00 75.00 85.00 85.00 95.00 | . 282 140 133 133 136 176 187 2213 2241 256 266 291 291 291 5085 | 243 7981 3403 32992 32992 22902 - | . 130 979 881 528 394 378 098 063 029 .029 .029 .140 .154 .148 148 453 459 441 427 | . 397 8650 770 545 346 211 1617 076 026 026 138 198 193 335 335 335 338 338 | .034872809477352318249217177127078020 .041 .096 .132 .148 .143 .170278278276276276 | .01991784945153282483161816180690321361301207197187178 | .11790763542932312981089006600770069008910990 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 20.00 25.00 30.00 35.00 45.00 65.00 65.00 65.00 85.00 85.00 85.00 85.00 85.00 | . 358 .289 .2893 .2055 .169 .1055 .0200 00284 0777 1364 1364 1364 1364 1364 1364 | | .341 .260 .184 .137 .058 .008 .032 .038 .038 .139 .1284 .174 .1289 .1444 .1842 .1842 .1842 .1842 .1842 | . 334 . 246 . 202 . 149 . 114 . 022 . 016 - 054 - 074 - 093 - 135 - 151 - 188 - 222 - 193 - 187 - 234 | . 315 . 240 . 150 . 096 . 030 . 035 072 125 125 125 200 232 227 251 251 251 261 311 | . 32 5 .2 4 2 .13 9 .00 6 .00 7 6 .00 5 4 .00 9 3 1 .17 12 2 .2 2 5 6 .2 3 3 4 0 .2 4 8 3 .1 6 8 8 .1 6 8 8 .1 4 4 1 .2 1 5 | . 266 .110 .034 .1407 .2904 .275 .265 .246 .214 .176 .1176 .1176 .1176 .1176 .1176 .1176 .1176 .1176 |

CONFIDENTIAL

TABLE III

| | PERCENT | | | PRESSU | RE COEFFICIENT, | , P, AT: | | |
|---------------|---|---|---|--|---|---|--|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.405/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0 | .94 $\alpha = 5.8^{\circ}$ | | | | <u>. </u> | · | <u> </u> |
| UPPER SURFACE | 1.25 2.50 5.50 105.000 25.0000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.0000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.000 25.0000 25.000 25.00 | - 4423 - 3242 - 227778 - 227778 - 22729 - 2272 | .011942961558758741963885358530583 | 083 - 1.187 - 1.144 - 1.092 - 1.00634784131131000470490851041201201478484464 | .273 -1.150 -1.201 -1.126 -1.070 -1.016652244181085044085044181085044181085044181085044181085044181085044181085044181085044343343 | - 1.204 - 1.204 - 1.244 - 1.0715 4155 234 136 0826 0286 0286 142 142 2877 2777 2777 | 398 - 1.063 - 1.001893800669494396180180180058004206707521092072109 | 228937855568024866376119571124112111201227112711095 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 45.00 60.00 60.00 70.00 80.00 90.00 90.00 | .419 .3775 .2955 .1779 .0883 .00883 .00883 .00883 .00883 .00883 .00883 .00883 .00883 .00883 .00883 .00883 | .449 .3713 .2489 .1587 .117 .0747 .0149 .0149 .1087 .1087 .1087 .1087 .1089 .1093 .1 | . 437 . 378 . 321 . 189 . 135 . 080 . 037 . 001 - 036 - 074 - 1127 - 151 - 177 . 151 - 177 . 181 - 119 - 175 - 119 - 175 - 181 - 194 - 194 | . 407 . 321 . 285 . 231 . 194 . 108 . 005 . 0027 . 001 . 0095 . 128 . 148 . 147 . 219 . 152 . 164 . 164 . 164 . 206 | .418 .349 .309 .3061 .1070 .01050 .0025 069 147 147 147 175 129 | . 431 .3247 .1913 .0920 .0020 .0058 .1041 .1499 .2479 .3290 .330 .330 .330 .330 .330 .330 .330 .3 | 3703786 3213786 32137788 3010308788866 301087888668 301087888668 301088 3010888 3010888 301088 301088 301088 301088 301088 3010888 3010888 301 |
| | M = 0 | $0.94 \alpha = 7.8^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.25 2.50 2.50 10.000 125.000 125.000 225.000 35.000 45.000 45.000 705.000 805.000 995.000 | .2 88 . 5392 . 4498 . 3774 . 37695 . 37695 | 207 - 1.074 - 1.063 - 1.051903643643438437437453457181093059059774736687619 | 261 - 1.273 - 1.215 - 1.305 - 1.1077842607482345237172096058058018524486477459 | . 148 - 1.248 - 1.288 - 1.322 - 1.182 - 1.126 - 1.0718473783600550062136138160375375375375 | 581 - 1.290 - 1.242 - 1.196 - 1.135 6555 6555 473 403 403 403 251 1708 057 007 | | 415 573 559 5533 5179 4893 4489 4425 425 3969 3528 31065 2271 2422 2422 |
| LOWER SURFACE | 1.25 2.55 5.00 10.00 13.00 25.00 35.00 45.00 45.00 65.00 65.00 65.00 80.00 90.00 90.00 90.00 90.00 | 4 4 3 3 4 5 1 4 2 7 3 4 5 1 4 2 7 7 3 4 0 1 2 2 4 9 1 1 2 3 | . 523 . 4454 . 3303 . 223 . 1407 . 0074 . 0046 0036 1355 . 1382 . 1477 . 0772 . 0046 0046 0046 01315 01317 0146 014 | . 492 . 427 . 345 . 292 . 262 . 198 . 059 . 024 - 009 - 036 - 066 - 088 - 112 . 142 . 050 - 169 - 169 - 205 - 205 - 685 | . 464 .383 .340 .283 .247 .170 .081 .081 .095 .026 063 026 063 026 130 130 130 130 130 130 130 130 130 | .454 .394 .310 .256 .213 .155 .099 .054 -017 -026 -089 -1164 -1667 -167 -167 -168 -168 -168 -168 | . 463 .403 .208 .249 .230 .152 .088 .035 053 057 159 264 264 264 193 118 118 | . 42 99 .2867 .1930 .0053 1148 1249 2258 2258 2258 11715 1119 1104 1000 0086 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | PERCENT | | | PRESSURI | COEFFICIENT, | P, AT: | | |
|---------------|--|--|---|---|--|---|---|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 94 α = 11.4 ⁰ | | | | | <u></u> <u>-</u> | |
| UPPER SURFACE | 1.25 2.50 5.50 10.000 20.000 25.000 25.000 45.000 45.000 655.000 655.000 805.000 805.000 | 534 577 571 537 519 509 494 | 063 063 856 849 761 | 628 - 1.219 - 1.1050985995994822733658543585554356754161364359485548 | 466 435 401 | 815 8003 7907 800997 800997 80099 776340 773440 773440 551687 55049 | 50B 5115 4957 4696 4601 4519 4601 4519 428 4319 3980 3980 3813 362 | 52944844514556445143564437437742041043889388435533542 |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 15.00 25.00 30.00 35.00 40.00 45.00 55.00 60.00 85.00 75.00 85.00 95.00 GAP | .6367 .574 .4083 .3689 .2628 .2240 .11395 .0955 | 4435 4435 3052 4435 3052 4435 4435 4435 4435 4435 4435 4435 44 | .545 .515 .452 .404 .367 .308 .224 .164 .128 .091 .025 .025 .037 -038 .025 -037 -115 -073 -141 -176 | . 538 . 487 . 444 . 3951 . 276 . 183 . 1146 . 1062 . 0024 . 0050 . 081 . 146 . 089 . 0897 . 1373 . 1455 | .500 .415 .320 .2622 .1604 .0759 0041 140 140 140 1651 1651 1651 | . 491 . 496 . 393 . 342 . 2375 . 1222 . 0774 . 0222 . 0772 . 1164 . 2231 . 231 . 131 . 134 . 1399 . 2686 . 2825 | . 4837 4837 4837 4837 4837 4837 4837 4837 |
| ļ | M = 0 | $.94 \alpha = 15.7^{\circ}$ | | | | | • | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 25.00 35.00 35.00 40.00 45.00 65.00 65.00 75.00 85.00 95.00 | . 156 - 414 - 629 - 799 - 841 - 831 - 818 - 757 - 709 - 671 - 6634 - 6535 - 526 | - 1.314 - 1.313 - 1.298 - 1.281 - 1.279 - 1.130 629 117 037 051 270 188 935 821 727 | - 1.1080 - 1.1080 - 1.1086 - 1.0066 - 1 | 599 587 589 | 5767 56677 56677 5771 5771 5768 55617 57549 55549 553723 551177 551170 5500 | 5215205120521522152215127511515511515497492492492495495485 | 557 5557 5553 5554 5556 5556 5551 5551 5547 55488 |
| LOWER SURFACE | 1 . 25 2 2 . 50 0 5 . 50 0 10 . 50 0 10 . 50 0 0 25 . 50 0 0 15 . 50 0 0 6 5 . 50 0 0 0 6 5 . 50 0 0 0 6 5 . 50 0 0 0 6 5 . 50 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 | .534 .629 .6285 .6288 .5409 .4497 .3348 .3488 .3152 .2522 .246 .198 .1064 .0181 .1282 .0048 | .633 .620 .575 .547 .478 .429 .380 .339 .297 .262 .215 .171 .134 .085 .027 .272 .134 .0066 .0029 .0079 .739 | .551 .585 .552 .519 .488 .430 .375 .328 .250 .210 .173 .096 .006 -006 -006 -005 -061 -005 -061 -0772 | . 584 . 571 . 546 . 497 . 458 . 390 . 344 . 293 . 207 . 1170 . 1086 . 044 . 0041 . 065 . 045 | .522 .4952 .4119 .3593 .256 .2155 .1729 .0844 .0003 094 .070 094 | . 400 . 508 . 462 . 421 . 398 . 324 . 265 . 210 . 110 . 059 . 009 . 114 . 107 . 201 . 107 . 125 . 220 . 305 . 355 . 355 . 355 | - 44639 - 44639 - 44639 - 319756 |

TABLE III

| | | PRESSURE COEFFICIENT, P, AT: | | | | | | | |
|---------------|---|--|--|---|--|--|--|---|--|
| | PERCENT CHORD | 0.135ь/2 | 0.25b/2 | 0.40ъ/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 | |
| | M = 0.9 | $\alpha = 20.0^{\circ}$ | | | | | | | |
| UPPER SURFACE | 1.00 1.25 2.55 5.50 10.00 15.00 25.00 25.00 35.00 45.00 65.00 65.00 65.00 85.00 85.00 85.00 | .046 .522 .793 -1.047 -1.047 -1.060 -1.924 807 727 694 742 6890 470 337 -1.045 337 -1.695 470 | 1 . 198 1 . 286 1 . 286 1 . 265 1 . 245 1 . 138 1 . 144 1 . 1425 1 . 1425 1 . 1080 1 . 0419 1 . 9745 1 . | - 977 - 9642 - 9768 - 9768 - 9760 - 9932 - 9914 - 8814 - 776 - 784 - 7742 - 7442 - 7748 - 7733 | - 847 - 757 - 757 - 757 - 7745 - 7745 - 7764 - 705 - 706 - 699 - 698 - 688 - 6805 - 6848 - 6717 - 6738 - 668 | | 6329 - 6286 - 6286 - 6628 - 6628 - 6618 - 6619 - 6619 | 65 9 66 4 2 66 4 12 66 3 1 66 3 1 - | |
| LOWER SURFACE | 1.25 2.500 7.500 10.00 15.000 20.00 25.000 35.000 45.000 50.000 65.000 75.000 85.000 95.000 95.000 | 25737243600559745554637642764276427642764477662764776447764 | 627 77006 66794 65946 65946 4997 4116 737314 2848 113 2848 113 2142 1071 100 | .518 .6117 .5973 .5729 .47350 .3750 | 5796 5902 5635 5635 4836 5336 43952 736217 7 | 45335 53150 53150 53160 53160 53173 53 | . 4215 . 515 . 5104 . 475 . 464 . 3940 . 2839 . 1863 . 10792 . 11863 . 10792 . 1196 . 1197 . 1197 | . 4197 . 4470 . 4470 . 4378 . 2158 . 2158 . 2080 . 008675 . 11988 . 2158 . 2158 | |
| | W = 0 | | | | 0.7.0 | 7.60 | - 991 | 749 | |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.00 25.00 35.00 45.00 45.00 66.00 67.00 70.00 75.00 85.00 95.00 | 768 776 716 | - 944 - 931 - 924 - 9327 - 9333 - 9233 - 928 - 928 - 928 - 928 - 928 - 928 - 928 - 927 - 884 - 884 - 885 - 887 - 887 - 887 - 887 | 942 8687 8886 8886 8878 8600 8437 82391 77855 7787 7787 7787 | 734 734 728 | 760 7561 7551 7551 751 746 747 747 743 743 743 738 718 798 719 798 719 799 698 713 | 891 7531 6890 6992 6992 6986 6986 6991 6991 7139 7139 7444 6335 6488 6324 6324 | 731 724 704 7111 705 706 706 7113 7113 706 706 706 706 706 706 706 | |
| LOWER SURFACE | 1 . 25 2 . 50 7 . 50 10 . 00 20 . 00 30 . 00 35 . 00 45 . 00 55 . 00 60 . 00 77 . 00 88 . 00 90 . 00 95 . 00 | 306 | .596 .725 .765 .752 .737 .688 .601 .557 .515 .475 .432 .384 .345 .292 .224 | . 465 6618 .6657 .6644 .6681 .5198 .4377 .3568 .3128 .2699 .145 .3169 .145 .3169 .1653 .014 | . 590 . 637 . 608 . 592 . 556 . 506 . 461 . 339 . 293 . 248 . 300 . 144 . 077 . 252 . 096 . 039 - 039 | .389 .525 .567 .559 .541 .458 .414 .373 .330 .286 .242 .117 .101 .029 .191 .058 .025 .107 .107 .107 .108 | .503 .531 .520 .514 .459 .410 .310 .260 .208 .152 .098 .042 014 053 .076 162 219 293 | | |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | PERCENT | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|--|--|---|--|--|
| | CHORD | 0.135ь/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 98 α = 0.0° | | | | | | |
| UPPER SURFACE | .00 1.25 2.500 7.500 10.000 120.000 25.000 35.000 405.000 555.000 65.000 85.000 85.000 95.000 | .303 .073 .089 .081 .0073 .032 .003 003 0049 052 .068 068 .146 .2373 624 581 581 554 554 | .562 028 0117 0337 0651 0937 0731 .0538 .2408 .3763 3999 86795 7795 602 | .503 070 036 026 021 017 .0025 .0039 .087 .2081 .303 .286 523 524 514 5507 | .833 .019 .018 .005 .024 .005 .013 .013 .025 .013 .025 .038 .064 .164 .164 .328 .328 .328 .328 .342 .424 .423 .423 .400 | .507 .1119 .0390 .0074 0064 0014 0016 .0018 .0018 .0105 .1708 .3116 .3538 3538 3538 3538 3538 | . 478 . 201 . 049 . 019 . 019 . 010 023 010 0062 . 0056 . 109 . 189 . 210 . 189 . 285 274 274 310 | .414 .122 .075 .039 .081 165 164 1083 083 087 1196 1196 1196 1197 1198 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 15.000 25.000 30.000 35.000 40.000 45.000 55.000 66.000 77.000 85.000 95.000 95.000 | 2585 2157 21079 20463 2000 2000 2000 2000 2000 2000 2000 20 | 007 .007 .0023 .00252 .00552 .00941 .11349 .11499 .2005 .2387 .07799 .11745 .11745 .11745 .11745 .11745 .11745 .11745 .11745 | 0345 0353 0053 0051 1916 1462 1462 1216 2268 2268 2368 237 3331 1829 1829 1829 1829 2030 2030 2030 | 105 085 0328 085 1543 2065 2361 2361 3471 3952 187 2247 2250 2502 | | 284 225 229 185 2402 312 312 343 476 4612 352 476 4612 291 352 294 219 | 392 2355 2757 3156 33164 3293 3771 3402 3771 3122 2363 1874 160 |
| | M = 0 | | | | | | | |
| UPPER SURFACE | 100 125 2.50 5.00 10.000 15.000 25.000 35.000 45.000 45.000 75.000 85.000 70.000 80.000 90.000 | .314 .0722 0443 0843 1244 1679 1794 12163 | . 29 2 . 7036 . 2763 . 2267 . 2267 . 2254 . 2254 . 2254 . 2259 . 2269 . 2269 | . 178 838 763 576 341 3319 226 0217 063 024 017 063 150 157 180 531 547 540 531 | - 449 - 800 - 818 - 729 - 570 - 150 - 118 - 101 - 081 - 0081 - 0016 - 044 - 104 - 118 - 141 - 188 - 178 - 434 - 434 - 434 - 434 | . 048 736 682 485 292 237 202 116 0619 019 019 141 172 188 353 353 353 354 345 345 | . 057 746 576 303 2637 190 147 1073 073 1079 1079 105 282 282 282 282 282 282 282 | .152 .7713 .5220 .2918 .22537 .1616 .0990 .11048 .1173 .11815 .1172 .11815 .1172 .11891 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 55.00 65.00 75.00 85.00 95.00 95.00 | 3864 3370 3377 23044 11490 00526 000526 000526 10192 111325 11125 11125 | | . 354 . 276 . 201 . 154 . 126 . 074 . 028 0043 014 1153 246 286 286 266 266 267 277 145 145 267 145 267 145 267 145 267 - | . 3355 . 2500 . 2138 . 1244 . 0292 0500 1368 217 189 1817 216 216 240 240 2440 2440 | . 315 . 2442 . 1493 . 093 . 0058 - 0068 - 0688 1166 1869 2266 2266 2298 2298 2998 | . 323 . 248 . 144 . 098 . 088 . 014 . 032 067 110 154 177 208 278 314 356 320 321 320 321 320 | 279856000011388665150138861500000000000000000000000000000000000 |

TABLE III

| Ī | | ···· | PRESSURE COEFFICIENT, P, AT: | | | | | | | |
|---------------|---|--|--|--|--|---|---|---|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85ъ/2 | 0,95b/2 | | |
| | M = 0.9 | $98 \alpha = 5.9^{\circ}$ | | | | | | | | |
| UPPER SURFACE | .00 12.50 5.00 10.00 15.00 15.00 25.00 25.00 45.00 55.00 65. | 2317 2213352 2213352 223422 | 449 376 362 3538 3758 3758 3758 3758 3750 1043 088 1018 8341 798 | 941 849 459 410 269 1134 0049 086 1007 5687 547 543 | 994 - 1 . 971 971 908 836 787 193 141 094 021 126 126 167 471 469 461 | - 1.0140 - 8912 - 8622 - 7510 - 1855 - 1451 - 0006 - 0047 - 1337 - 1469 - 3866 - 3830 - 375 | - 1.049 897 8679 3260 2159 1048 010 059 3127 3127 3286 3276 | 873 4570 2763 1837 1229 1445 1485 1942 21138 22141 22141 | | |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 15.00 35.00 35.00 45.00 45.00 55.00 65.00 75.00 85.00 95.00 95.00 | .430 .360 .315 .2815 .2911 .1520 .085 -004 -0045 -0067 -0867 -2980 | .180 .1403 .070 .036 .012 025 062 | - '036 | .203 .1110 .066 .027 .0027 .0063 086 111 | .3434 .3434 .1999 .1160 .00153 00285 0160 11735 1295 295 1099 2492 | .195 .184 .101 .040 | .395 .3234 .1500 0113 2067 3034 35539 38539 2891 2992 2270 2478 | | |
| | M = 0 | | 1 1 1 1 | 104 | 226 | - 484 | 561 | 458 | | |
| UPPER SURFACE | 35.00 40.00 45.00 50.00 60.00 65.00 75.00 80.00 95.00 | 419 386 347 318 327 321 321 321 321 321 321 321 321 324 349 349 349 554 554 519 | - 41y - 428 - 222 - 015 - 081 - 162 - 058 - 834 - 785 - 772 - 637 | - 1.036 - 1.010 - 920 - 608 - 4493 - 214 - 119 - 056 - 0012 - 055 - 064 - 075 - 065 - 075 - 085 - 0 | - 1.063 - 1.016 - 988 - 913 - 607 - 408 324 183 037 053 112 189 189 189 189 189 | - 1.071 - 1.047 972 701 353 4152 238 179 084 062 439 439 439 439 439 | 9889 670 560 504 481 455 435 375 334 375 334 395 395 395 395 389 | | | |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 25.00 25.00 35.00 45.00 65.00 70.00 85.00 85.00 95.00 | .449 .446 .362 .292 .273 .219 .171 .144 .118 .056 .025 .025 .046 .046 .046 .058 .058 .058 .058 .058 .058 .058 .058 | .540 .473 .404 .350 .250 .250 .250 .164 .128 .091 .024 .038 .031 .209 .164 .209 .164 .785 | .445 .361 .310 .278 .220 .165 .165 .079 .044 .005 -020 -047 -057 -136 | . 392 . 3555 . 2999 . 266 . 184 . 140 . 067 . 031 003 102 165 165 | .407 .3269 .2312 .115 .0777 .0037 .0099 1187 187 187 187 187 27313 | . 408 . 311 . 257 . 242 . 159 . 099 . 007 - 046 079 120 179 243 264 331 254 347 351 | . 36857 . 28878 . 10582 . 10552 . 10552 . 126661 . 22626 . 12626 . 126 | | |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | DUDGENM | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|---|---|--|---|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 98 α = 11.4 ⁰ | | | | | | |
| UPPER SURFACE | 12.5000000000000000000000000000000000000 | .295 1723 4413 483 486 445 442 442 442 442 442 420 419 460 385 235 | 467 -1.106 -1.087 -1.120 -1.080 -1.080 -1.030 -1 | 490 - 1.0938 - 1.9846868475725913448751853335138666166616681 | 013 - 1.065 - 1.059 - 1.045 - 1.030 999 941 871 834 767 767 736 698 576 421 418 385 503 5743 5743 | 7 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 | 763 6166 6674 6674 6670 6611 66313 66313 6112 5540 5540 540 540 540 646 674 674 674 674 674 674 674 654 540 | 7 1 7 - 6435 - 66234 - 66137 - 58617 - 55857 - 5557 - 5541 - 5330 - 5222 - 5110 - 5047 - 4946 - 483 |
| LOWER SURFACE | 1.25 2.500 5.500 10.500 20.000 25.000 45.000 45.000 55.000 605.000 705.000 85.000 95.000 95.000 | . 467 .568 .6171 .5708 .6171 .5720 .403 .3055 .3065 .2636 .1873 .173 .173 .173 .173 .173 .173 .028 .028 .028 .028 .028 .039 .039 | 638 5540 487 458 384 3913 2216 2216 2216 2060 0001 242 0060 0001 242 0067 0067 | .575 .5477 .4394 .3392 .2378 .15237 .15237 .0953 .0953 .0953 .0953 .0040 .0024 | . 569 . 509 . 468 . 418 . 377 . 307 . 258 . 211 . 177 . 138 . 097 . 0012 . 0034 . 121 . 135 . 0407 . 126 . 176 . 544 | . 5 25 . 4 35 . 4 35 . 38 43 . 28 46 . 18 42 . 10 63 . 0 62 5 . 0 61 1 . 10 1 . 0 5 3 . 0 66 6 . 11 6 7 . 12 15 . 73 1 | 508 483 408 3357 2558 11430 00557 00501 11518 274 118 274 274 275 275 275 275 275 275 275 275 275 275 | 365 3259 1338 0335 1338 0375 12531 2680 3329 3379 3379 3379 3379 34132 4447 468 |
| | M = 0 | | 840 | 001 | 706 | 5.6.7 | - 630 | . 650 |
| UPPER SURFACE | 25.00 10 | 7436 7366 627 5934 5557 5557 5793 4990 990 709 6019 524 | 8457 - 1.25603 - 1.26603 - 1.26603 - 1.1875 - 1.1875 - 1.17587 - 1.17587 | 744 637 578 | 7 0 4 9 | 563 5554 5630 5970 6679 67118 7029 6785 6648 66306 6648 6648 6648 | - 630 - 6628 - 6616 - 628 - 6216 - 628 - 631 - 6226 - 6619 - 6619 - 6600 - 589 - 587 - 567 - 587 - | |
| LOWER SURFACE | 1.25 25.20 25.20 20.20 10.00 2 | . 4 85 . 4 89 . 3 85 . 3 53 . 2 96 . 2 40 . 2 03 . 1 05 . 2 16 . 1 59 . 1 103 | .669 .683 .652 .608 .577 .5113 .416 .375 .333 .254 .278 .172 .125 .070 .173 .1043 .043 .043 .043 .043 .043 | .589 .5678 .578 .5410 .4574 .3575 .2789 .206 .1631 .0934 .0939 .0936 .0368 .0368 .0368 .0368 .0368 .0388 .03 | . 593 . 586 . 565 . 520 . 480 . 418 . 372 . 320 . 284 . 240 . 157 . 115 . 074 . 031 . 034 . 210 . 040 . 040 | .524 .542 .509 .471 .434 .3818 .2839 .2001 .1601 .0417 .043 .0147 .0263 .1289 .1289 .1289 .1289 | . 494 . 526 . 483 . 440 . 421 . 349 . 239 . 194 . 1048 . 0078 . 0069 . 1216 . 1276 . 1187 . 262 . 312 . 312 | . 4489 .4489 .4488 .385 .339 .125 .0512 097 128 1284 2140 2246 2246 2313 3354 3354 394 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| Ì | | | | PRESSUR | E COEFFICIENT, | P, AT: | <u>_</u> | |
|---------------|---|---|---|---|---|--|---|--|
| | PERCENT CHORD | 0.135ь/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 0. | 98 α = 20.2° | | | | ١, | | |
| UPPER SURFACE | 12.50000 12.5000 12.5000 12.5000 12.5000 12.5000 12.5000 12.5000 12.50 | 0 5 3 3 4 6 9 8 8 9 4 1 8 9 8 1 9 9 5 1 1 1 9 9 5 5 1 1 1 9 9 5 5 1 1 1 9 7 1 6 6 6 6 7 0 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 | - 1.079 - 1.3083 - 1.3006 - 1.3006 - 1.3006 - 1.3006 - 1.2875 - 1.2892 - 1.28969 - 1.38969 - 1. | - 1 . 144 - 1 . 171 - 1 . 111 - 1 . 116 - 1 . 117 - 1 . 110 - 1 . 074 - 1 . 055 - 1 . 066 - 1 . 044 - 1 . 010 984 884 | - 736 - 673 - 6873 - 6890 - 6990 - 723 - 7623 - 796 - 807 - 807 - 807 - 795 - 795 - 7784 - 777 - 759 - 757 - 759 - 751 - 757 - 757 - 758 - 771 - 757 - 745 - 723 | 717 - 717 - 720 - 720 - 725 - 727 - 736 - 736 - 736 - 736 - 736 - 736 - 737 - 736 - 721 - 7147 - 697 - 688 - 688 - 688 - 688 | 682 6839 6735 6887 68852 66852 | 698 688 6886 6886 6886 6884 6881 6681 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 10.00 25.00 35.00 40.00 40.00 65.00 75.00 65.00 75.00 85.00 90.00 90.00 95.00 | 300 545 8873 6777 6652 6652 6549 5741 4410 3312 222 3303 2172 122 | 67322 67322 770592 67225 67225 68438 4083 4083 4083 4182 4217 1847 1847 1807 1919 | 54448395555669466928 06654566928 0766288 0766288 0766288 0766388 0766388 0766388 0766388 0766388 0766388 076638 07660000000000000000000000000000000000 | 564 622 5964 5107 4201 53365 2253 2170 1227 2996 0096 0096 0097 0097 0097 | 478 5560 55398 4687 5732 28488 24062 1118 0809 1829 1929 1929 1929 1939 1939 1939 1939 19 | . 447 .5330 .5030 .492 .427 .374 .276 .224 .1724 .0008 .0087 .0087 .0087 .0098 | . 432 4995 4492 . 4402 . 31029 . 1123 . 11248 11395 11395 1217137 128077 1334894 |
| | M = 1 | .00 α = -0.1° | | | | | 493 | .401 |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 25.00 35.00 35.00 45.00 65.00 65.00 65.00 75.00 85.00 | 311 1288 1118 1091 00515 00200 - 0020 - 0033 - 0544 - 0735 2328 3288 - 5468 - 64023 - 64023 - 3393 | .573 .0011 .0011 .0053 013 028 047 082 027 .090 .167 .257 .356 .404 .391 .326 887 720 601 | 1 | 830 049 049 049 034 034 034 036 047 084 185 269 365 1458 - 468 - 468 - 107 | 510 .0363 .0369 .0029 .0018 .00154 .0014 .0049 .1178 .2651 .3229 422 422 428 428 | . 483 . 285 . 087 . 096 . 008 . 009 . 0014 . 0114 . 029 . 037 . 067 . 1079 . 228 . 2166 342 | . 1478 . 0734 . 0754 . 0174 1500 16318 0778 0778 0781 25566 22644 26725 26725 |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 20.00 25.00 45.00 55.00 60.00 65.00 70.00 85.00 85.00 65.00 | .217 .172 .0959 .0059 .0059 .0050 0054 0053 1054 1134 1 | 002 042 060 083 1024 124 143 170 193 225 256 | 018 027 048 087 113 1155 181 202 281 329 329 141 191 191 191 191 191 192 | 083 030 068 181 196 196 225 250 250 355 376 376 355 221 221 221 221 | 163 158 178 187 2186 262 262 3149 3749 414 424 226 226 3297 2297 | 233 223 218 182 236 268 303 311 336 | - 309 - 2644 - 2599 - 3094 - 3094 - 3094 - 3094 - 3098 - 3459 - 3275 - 2275 - 2230 - 2344 |

CONFIDENTIAL

TABLE III

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|--|---|--|---|---|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | , M = 1. | $\alpha = 3.8^{\circ}$ | | | | | | |
| UPPER SURFACE | .005 1.250 5.000 10.000 15.000 25.000 30.000 45.000 45.000 555.000 655.000 85.000 85.000 90.000 | 2 32 107 .268 .3557 554 686 | .099 .211 .238 .174 099 .070 802 849 763 566 | . 2 2 3 - 78 6 - 7 2 0 - 50 0 - 30 0 - 30 0 - 29 6 - 29 6 - 03 3 - 04 4 - 03 3 - 13 2 - 17 8 3 - 18 3 - 25 5 7 - 55 2 9 | . 487 - 755 - 777 - 686 - 524 - 321 - 145 - 108 - 056 - 051 - 065 - 120 - 163 - 209 - 186 - 471 - 471 - 471 - 471 | .083 .677 .627 .446 .327 .265 .207 .183 .150 .108 .069 .059 .150 .110 .181 .189 .283 .414 .418 .418 | .018 711 698 547 281 250 175 1061 061 068 .083 .120 128 .121 108 .335 335 330 3323 3323 | .20671966648137125522015610840840961301880253254235222229232 |
| LOWER SURFACE | 1 .25 2 .50 5 .00 10 .00 20 .00 25 .00 40 .00 45 .00 55 .00 65 .00 75 .00 85 .00 95 .00 GAP | .392 .3755 .22466 .2155 .1488 .10644 .0044 .0044 .0054 .0054 .1096 .1104 .11083 .11267 | .051 -008 -0085 -097 -1500 -236 -2075 -1000 -129 | | 328 -249 -211 -151 -166 -039 -040 -040 -156 -221 -221 -221 -221 -221 -221 -221 -22 | 318 -146 -100 -061 -020 -086 -118 -1350 -179 -215 -240 -105 -1707 -2474 -474 | . 301 . 209 . 081 . 078 . 011 . 084 . 121 . 160 . 175 . 252 . 315 . 345 . 349 . 317 . 326 . 348 | . 266 . 189 . 124 . 0052 . 0198 2491 |
| | M = 1 | | | | | | | |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 20.00 20.00 25.00 40.00 45.00 55.00 60.00 75.00 85.00 85.00 90.00 | .3 27 401 279 2301 1894 2217 22317 22319 2259 2259 2359 2359 2559 2559 25737 | 81982257944993953133133103281301241241248188755677582 | - 9665 - 8784 - 7887 - 3963 - 3766 - 3276 - | .3789349749178507867383541670190390390493493493494494494 | 17994094388580573741570780031170031170031170031441441441445435 | 245 966 976 8835 8303 5746 1288 1288 1288 0236 1285 0316 0316 0316 0316 0316 0316 0316 0316 0316 | 115974955825243664352153311431021143235525622572455248 |
| LOWER SURFACE | 1.25 2.50 5.50 10.00 20.00 25.00 35.00 45.00 45.00 65.00 65.00 65.00 85.00 95.00 | . 446 .370 .327 .293 .237 .164 .099 .011 .0021 .004 .004 .005 .005 | .401 .332 .282 .259 .192 .153 .083 .026 011 043 120 215 .068 068 068 | .299 .216 .164 .113 .069 .033 .0001 049 143 164 176 | | | | |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | | | | PRESSU | RE COEFFICIENT, | P, AT: | | |
|---------------|--|--|---|---|--|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1. | $.00 \alpha = 7.8^{\circ}$ | | I | | | | l |
| UPPER SURFACE | .00 1.850 5.500 10.5000 25.000 25.000 45.000 45.000 65.000 65.000 65.000 85.000 85.000 | .331 5106 4149 2978 2882 2982 2982 2982 2982 2982 2972 2972 2079 6627 5673 5673 5673 | 069 991 965 890 689 446 371 378 382 382 383 184 1064 1412 8299 767 698 608 | 126 - 1.071 - 1.030 - 1.0169529826455045501880260870671971975818556 | .269 -1.046 -1.079 -1.033 -983 -9931 -913 -362 -385 -146 -0075 -1314 -196 -211 -196 -211 -196 -211 -196 -211 -196 -211 -516 | 404 - 1.064 - 1.072 - 1.029 976 9032 4768 296 358 296 194 194 10533 0533 0533 469 469 461 | 483 - 1.069 - 1.0794 995 895 6490 426 383 318 318 328 328 435 435 435 435 435 435 435 435 435 435 435 435 435 435 338 | 383 984 975 8621 646 547 505 434 408 378 378 366 349 378 378 366 349 378 37 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 25.000 25.000 30.000 40.000 50.000 60.000 70.000 85.000 90.000 90.000 90.000 90.000 | 4 7 0 8 6 8 8 4 4 5 7 0 8 2 8 4 4 5 7 0 8 2 8 4 4 5 7 0 8 2 8 4 4 1 1 3 7 0 2 9 9 1 0 0 2 9 9 1 0 0 0 1 0 0 0 1 0 0 0 0 1 0 0 0 0 | .546 .478 .410 .3559 .2592 .21767 .136 .0738 .0018 .0018 .0194 .181 .0193 .019 | .517 .4564 .3152 .2241 .1372 .0048 .0043 0043 0043 00807 00807 00807 00807 00807 00807 00807 00807 00807 | . 480 .3910 .2958 .1839 .1011 .0033 .0043 0709 1193 .054 1052 1052 1052 | .471 .3924 .2728 .1762 .1262 .1061 .0040 .0040 .0040 .0112 .1149 | . 469 .3812 .2557 .1611 .1059 .0129 .0129 .0252 157 2359 2317 2378 2378 2378 | .4367 .23686 .2162 .10536 .00386 22660 2295 33383 23544 2993 2993 31149 |
| | M = 1. | | ,,,,,, | .030 | - ,468 | 508 | 423 | |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 45.00 60.00 65.00 75.00 85.00 95.00 | 3 0 2 - 15 3 2 - 4 16 3 - 4 6 4 - 4 6 8 - 4 2 6 - 4 10 4 - 4 10 2 - 4 10 2 - 4 10 2 - 4 10 2 - 5 2 2 - 5 2 3 - 9 6 4 1 - 5 2 3 - 9 6 4 1 - 5 1 3 7 - 7 1 3 7 - | 441 - 1.080 - 1.063 - 1.094 - 1.0494 - 1.0494 - 1.04968678699551417406002113108110018830831 | 459 - 1.057 - 1.018 977 8948 8039 6684 529 456 529 529 456 644 6 | .008 -1.004978964589928842813785078294459446041261858555521 | 692787787778778778781778784780772762749736722749736684684684689694694 | 750 699 6755 6666 6636 6636 6643 6440 6315 5975 5503 5203 512 512 | 608 546 525 525 525 5210 497 486 456 443 443 443 443 4414 414 414 414 415 403 398 389 |
| LOWER SURFACE | 1.25 2.50 5.50 10.00 15.00 25.00 35.00 45.00 45.00 60.00 70.00 80.00 80.00 95.00 95.00 | 472 .672 .672 .620 .578 .528 .446 .412 .362 .310 .243 .190 .181 .138 .108 .034 .125 .050 .033 | .647 .606 .546 .494 .466 .395 .395 .391 .262 .225 .190 .151 .108 .070 .004 .024 .253 .123 .072 .017 .017 | .588 .5554 .489 .445 .406 .3493 .248 .2209 .175 .138 .107 .074 .019 -023 .048 .019 -023 .047 .047 .057 | . 579 . 521 . 481 . 488 . 388 . 375 . 270 . 221 . 185 . 146 . 109 . 002 . 002 . 002 . 002 . 006 . 105 . 152 . 154 . 154 | .531 .438 .348 .348 .235 .188 .149 .110 .074 .036 .006 049 107 099 .071 .099 | .517 .493 .493 .367 .368 .267 .210 .158 .116 .071 032 097 169 268 305 481 | . 481 . 442 . 374 . 309 . 255 . 151 . 034 102 153 188 214 255 247 255 258 258 258 365 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| ſ | | | | PRESSUF | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|---|---|--|---|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55ь/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| T | M = 1.00 | 0 α = 15.9° | | | | | · · · · · · · · · · · · · · · · · · · | |
| UPPER SURFACE | 1.25 2.500 7.500 10.000 15.000 25.000 25.000 35.000 45.000 45.000 670.000 670.000 805.000 905.000 | . 2 07 - 2 70 - 482 - 646 - 696 - 673 - 5747 - 5522 - 5511 - 5536 - 4562 - 2431 - 959 - 660 - 660 | 786 - 1.163 - 1.159 - 1.156 - 1.107 - 1.0976 - 1.0976 - 1.0978 - 1.0978 - 1.08478837550641 | 865 - 1.104 - 1.042 - 1.067 - 1.067 - 1.069 - 1.023987965961974 - 1.049 - 1.049 - 1.049 - 1.049 - 1.049 - 1.045965961974 - 1.045886960832947768664 | 822 691 739 725 7725 841 877 842 | 1.1495548546854685483669769956993689867736646 | 712 - 7184 - 6448 - 66497 - 66317 - 66316 - 663361 - 663361 - 66312 - 55566 - 55564 - 55564 - 55564 | 27565576655344713086471430847140847180847180847140847140847140847 |
| LOWER SURFACE | 1.500 7.500 1.500 | . 418 .5777 .7636 .6800 .5554 .4405 .3108 .2227 .1127 .2376 .118 | .692 .703 .671 .627 .596 .531 .4437 .395 .3539 .275 .275 .147 .095 .330 .147 .007 .007 .007 .007 | . 598 . 624 . 587 . 553 . 520 . 468 . 414 . 371 . 330 . 293 . 2257 . 221 . 182 . 1146 . 115 . 105 . 051 . 050 . 013 . 067 . 709 | . 594 . 596 . 576 . 530 . 494 . 428 . 332 . 299 . 253 . 213 . 173 . 173 . 092 . 047 . 012 . 230 . 059 . 045 . 104 . 104 | .538 .558 .527 .487 .451 .3994 .297 .2256 .213 .175 .136 .053 .022 -015 .149 .045 -100 -153 -153 -153 -153 | .509 .539 .496 .441 .367 .260 .212 .168 .072 .039 .118 .072 .039 .1145 .145 .169 .201 .201 .201 .201 | 4 76 4 48 2 4 44 4 3 90 3 4 3 7 2 4 6 1 4 8 0 6 5 - 0 6 0 9 - 0 6 2 - 1 1 8 7 - 2 1 1 - 1 2 0 3 - 2 6 9 - 2 7 1 - 3 0 0 6 - 3 3 8 - 3 7 7 |
| | <u>M = 1</u> | T | 673 | 507 | .846 | .544 | . 525 | . 449 |
| UPPER SURFACE | .00 1.25 2.50 7.50 10.000 25.000 35.000 40.000 50.000 50.000 60.000 70.000 80.000 80.000 90.000 | 2 52 0 74 0 74 0 74 0 74 0 74 0 0 68 0 0 42 0 12 0 0 31 1 0 0 55 1 0 0 55 2 0 12 2 0 12 2 0 12 2 0 12 1 0 1 | . 573 001 003 028 062 096 096 099 | 482 473 467 | . 054 . 059 . 059 . 074 . 062 . 076 . 078 . 124 . 166 . 223 . 370 . 371 . 386 . 371 . 341 . 413 413 413 426 | .180 .113 .113 .075 .056 .057 .051 .058 .073 .087 .122 .163 .220 .306 .361 .372 .394 .374 .374 .380 .380 | .304 .130 .116 .084 .078 .042 .059 .063 .079 .087 .111 .1235 .2802 .272 .262 .272 .262 .272 .262 | .189 .11420 .1022 -0025 -00158 -10131 -0028 -0039 -0161 -0233 -02132 -22132 -22132 -22132 -22132 -22132 -22132 -22132 -22132 -22132 |
| LOWER SURFACE | 1.25 250 5.500 70.000 20.000 25.000 35.000 35.000 45.000 65.000 65.000 65.000 90.000 90.000 90.000 | 092 125 144 184 360 167 178 | .0056 0364 0544 075 0990 1112 1189 189 209 223 | 014 044 054 1084 1084 1452 1252 2081 2452 2811 1664 1655 1656 15 | 051 000 037 053 114 131 163 163 1215 245 278 300 310 174 - | 121 114 129 141 152 207 218 256 255 317 352 317 202 212 212 2212 232 232 | 185 174 166 130 214 238 2576 298 333 4097 363 291 291 226 | 260 194 214 210 2402 284 284 284 333 3799 334 3165 1765 189 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | DEDGENT | | | PRESSU | RE COEFFICIENT | , P, AT: | | |
|---------------|---|--|---|---|---|--|---|--|
| | PERCENT CHORD | 0.135ь/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| UPPER SURFACE | M = 1 .000 1.25 2.500 7.500 10.000 15.000 25.000 25.000 40.000 45.000 50.000 65.000 65.000 | .2591140880850880961121145152151166221160217 | .35763757392392319214214214224224246246246246246246 | . 254 728 677 5551 282 287 289 284 057 016 057 016 057 016 057 016 057 016 058 097 141 189 186 | . 529 727 754 663 559 348 278 067 032 003 049 108 160 196 240 | .110685681669516304082082073046011039046011039046011039046011039046011039046 | . 150 656 644 469 268 222 191 101 067 022 019 1161 174 165 | .239 - 673 - 626 - 472 - 342 - 218 - 1181 - 1181 - 1056 - 040 - 030 - 044 - 082 - 146 - 1217 |
| | 70.00 75.00 80.00 95.00 95.00 | .337 522 532 501 448 | .099 737 759 683 606 510 | .217 485 501 497 482 469 | .232 425 425 424 431 433 | .241 381 404 390 392 393 | .157 299 294 288 281 | 213 197 195 201 199 |
| LOWER SURFACE | 2.50 7.50 10.00 15.00 20.00 30.00 30.00 45.00 45.00 55.00 60.00 60.00 70.00 | .333 .304 .266 .227 .201 .149 .146 .101 .063 .047 .027 .023 .045 .076 .066 | 290 235 189 177 118 086 056 027 - 000 - 0149 - 077 - 122 - 1198 | . 289 . 215 . 171 . 144 . 097 . 053 . 022 - 012 - 036 - 094 - 139 - 177 - 216 2267 | . 260 . 2225 . 172 . 135 . 065 . 029 007 039 179 159 159 1213 213 219 | .277 .191 .136 .099 .061 .008039107131135162187223 | .278 .126 .1178 .126 .119 .055 .005 .0063 .063 .091 .123 .123 .259 .2252 | 2349 -1032 -00447 -1052 -2669 -2669 -2884 -2884 -2884 -2831 |
| | 80.00 85.00 90.00 95.00 GAP | $\begin{array}{cccccccccccccccccccccccccccccccccccc$ | 056 080 090 095 771 | 041 073 099 135 615 | 092 103 140 174 395 | 117 155 192 219 443 | 241 265 269 278 313 | 241 251 253 243 |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 20.00 25.00 35.00 45.00 45.00 55.00 65.00 65.00 75.00 85.00 90.00 | 278 - 3044 - 2082 - 1187 - 1287 - 1287 - 2004 - 2082 - 1287 - 2004 - 2004 - 2004 - 2004 - 2009 | .171844795675388315284284283290304117147147741744675522 | . 076904830751721452332347290061058154153153153508508508508508 | . 423 879 913 850 742 717 542 176 086 040 . 070 . 135 . 185 226 . 237 . 226 449 455 452 | 14089191086207806884871620735035051814817918444013417415 | 1979059188317636172762311630960480054094118117116322318310301291 | .24166264662466231491811059039003440442149212211957197 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 25.00 25.00 35.00 40.00 45.00 55.00 60.00 75.00 85.00 95.00 95.00 | 390 4209 3842 3842 3848 2370 1148 2297 1148 12087 00455 00201 - 00202 - 00202 - 00202 - 0035 | .485 .414 .347 .299 .277 .213 .175 .141 .110 .080 .059 .024 012 073 163 | . 478 . 404 . 326 . 277 . 246 . 196 . 108 . 073 . 039 . 002 . 040 . 093 . 132 . 157 . 139 . 029 . 024 . 029 . 024 . 029 . 024 | .445 .357 .327 .274 .231 .155 .015 .029 -015 -047 -035 -048 -113 -132 | .439 .369 .281 .227 .192 .099 .062 .023 011 043 134 134 136 184 136 184 | . 456 .3886 .28316 .2316 .147 .042 .0045 0461 1662 2549 22549 22563 22563 22563 22563 22563 22563 | 3370 -1040 -00456 -01993 -2264 -2268 -2266 -2286 -2282 |

TABLE III

WING WITH UPPER SURFACE SPOILER (WITH GAP)

| | | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|--|--|--|---|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.0 | $\alpha = 7.8^{\circ}$ | | | | · · · · · · · · · · · · · · · · · · · | | |
| UPPER SURFACE | . 00 12.50 5.50 10.50 10.00 10 | 2 8 4 | | 085 - 1.008 967 964 903 881 807 640 439 385 095 001 - | . 308 - 1980 - 1980 - 1965 - 8865 - 8812 - 578 - 368 - 315 - 024 - 1267 - 274 - 467 - 457 | 352 -1.0012 975 935 913 644 428 329 283 224 197 1867 142 425 448 425 448 425 448 425 448 425 448 425 448 425 448 425 448 428 448 428 448 | - 423 - 969 - 8969 - 8862 - 7888 - 587 - 464 - 3161 - 3369 - 361 - 324 - 3172 - 2947 - 430 - 435 - 437 - 437 | - 0 8 3 - 9 20 0 - 78 7 - 77 8 4 - 75 8 1 - 22 9 - 11112 - 108 8 - 15 9 8 - 16 9 8 - 23 3 1 - 16 9 8 - 23 3 1 - 23 3 3 - 23 3 1 - 23 3 3 - 23 3 3 3 - 23 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 25.000 25.000 45.000 45.000 45.000 65.000 75.000 80.000 90.000 95.000 | . 4 2 2 . 4 9 9 . 4 9 4 . 4 8 5 . 4 20 1 . 3 20 5 . 2 1 16 6 . 1 10 7 . 1 2 7 . 2 3 1 3 1 3 1 1 1 1 1 1 1 1 1 1 1 1 1 1 | .571 .502 .435 .384 .2846 .2074 .143 .080 .047 .047 .047 .041 .140 .0641 .0641 .0641 .0641 .722 | .541 .476 .394 .349 .349 .2608 .166 .1090 .0042 .0092 .0092 .0077 .025 .025 .0503 .0 | . 513 . 430 . 387 . 3393 . 2233 . 180 . 144 . 0817 . 028 . 074 . 071 . 091 . 096 . 094 . 0 | .504 .444 .365 .3073 .220 .1188 .00181 .0018 .0018 .0018 .0018 .0019 .0079 .0079 .136 .0070 .1394 .472 | 2 4 4 2 2 3 3 2 2 9 4 2 2 9 1 2 2 0 4 4 2 2 0 4 4 2 0 6 6 6 2 2 6 1 6 5 7 1 6 6 6 7 2 6 6 1 1 8 7 7 1 2 8 0 6 6 6 1 2 2 6 1 1 8 7 7 1 2 8 0 6 6 6 1 4 1 9 | . 420 .3469 .1946 .0425 1331 22726 22726 227310 - |
| | M = 1 | | - 300 | . 396 | 0.54 | 630 | 695 | 679 |
| UPPER SURFACE | .00 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 65.00 65.00 65.00 75.00 85.00 85.00 | . 3 05 101 241 413 4121 3872 3541 3541 3548 378 378 3744 3548 3744 344 | 382 - 1.004 988 - 1.014 972 925 804 743 511 347 050 .161 077 024 722 660 564 | | . 054 - 930 - 918 - 9018 - 8616 - 7528 - 676 - 647 - 618 - 5416 - 424 - 581 - 581 - 580 - 5301 - 5301 | 620 736 725 723 723 725 725 725 725 726 725 - | 695 6647 6234 6335 6335 6300 6200 6201 6201 5649 5649 5495 518 518 518 539 | 6324 60003 5977 5646 55326 55266 55266 55266 4988 4835 4670 4630 |
| LOWER SURFACE | 1.25 5.00 7.50 10.00 25.00 30.00 35.00 40.00 45.00 55.00 65.00 70.00 85.00 95.00 | . 4 69 . 594 . 735 . 6509 . 563 . 444 . 3304 . 279 . 226 . 218 . 1745 . 068 . 1613 . 075 . 056 | 674 635 578 526 498 427 377 335 229 226 185 142 108 039 018 158 158 165 165 165 165 | .616 .581 .517 .472 .435 .381 .325 .278 .237 .206 .167 .139 .107 .084 .058 .013 .088 .013 | . 5945 . 507 . 458 . 418 . 342 . 296 . 253 . 217 . 181 . 140 . 104 . 067 . 033 . 016 . 059 . 029 . 020 . 071 . 122 . 521 | .526 .462 .414 .376 .317 .263 .219 .181 .141 .104 .067 067 063 071 | | . 427 .358 .2390 .238 .018 .057 183 229 229 2314 268 2914 3514 375 |

TABLE IV

| | | | | PRESSU | RE COEFFICIENT, | , P, AT: | | |
|---------------|---|--|--|--|--|---|--|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.6 | $\alpha = 0.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 1.250 2.500 1.250 2.500 1.500 1.500 2.500 | .017 .106 .039 -013 -051 -082 -117 -128 -154 -154 -154 -154 -1210 -2183 -2219 -239 -282 -219 -088 -089 | .4651601231131136133153166182190202217232247254269750003057090115 | . 4342421851758176319419821372352442362494901033181 | . 647 300 236 164 188 213 2216 225 224 2341 249 250 247 247 178 178 178 | . 406 . 3784 28315 2219 2219 22131 | . 399 496 317 256 239 227 226 227 226 227 226 227 226 227 226 227 226 227 228 219 203 180 150 128 129 125 117 | .271456338261265245210206188187176186176176176176141128118108092084074 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 15.000 20.000 20.000 40.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.000 60.0000 60.000 60.000 60.000 60.00000 60.0000 60.0000 60.0000 60.0000 60.0000 60.00000 60.00000 60.00000 60.00000 60.00000 60.00000 60.00000 60.00000 60.00000 60.00000 60.00000 60.0000000 60.00000000 | .159 .0930 .0099 .0037 067 067 067 067 067 051 .0996 .2996 .2996 .3281 281 | 040 0366 0666 0666 0680 070 0388 0388 0388 2918 7354 7354 7354 6233 5129 | .008 .0015 0012 0040 0033 0013 0013 .0014 .0046 .1415 .3092 35617 3617 3844 3745 | .0365 0299 0311 0020 .0010 .0019 .0019 .0039 .1527 .3128 .3008 295 3044 3144 3220 | 210853495097626310 955962 2108532233558275222 2333343 2108532233558275222 2333343 | . 221 . 098 . 103 . 0655 . 0228 . 031 . 031 . 041 . 068 . 133 . 196 . 253 . 253 . 2157 1147 1364 1372 | . 173 . 122 . 052 . 017 006 023 034 033 037 037 037 037 037 038 039 - 039 - |
| | M = 0. | | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 20.00 25.00 30.00 40.00 40.00 50.00 50.00 60.00 70.00 85.00 95.00 | . 0 21 - 157 - 2542 - 2552 - 2752 - 2752 - 2752 - 2871 - 2871 - 2879 - 2 | 200 - 1.0545813853854342332363215315315326320315327326320315327327327327327378 | 483 - 1.383 734 547 484 443 3682 3652 345 337 320 320 320 3152 170 180 170 | 224 - 1.27487841053345345333563356349336339266252186180180179 | 454 9956 7573 66110 457 4289 33645 3315 24482 2402 1982 1751 | 354 931 803 742 674 583 467 427 329 295 295 285 206 177 160 1160 1142 132 | 5775 9725 6696 66141 4058 32758 2259 2 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 35.00 45.00 45.00 65.00 65.00 80.00 90.00 GAP | .3093 .2845 .2044 .1778 .1377 .00797 .0057 .0054 .00695 .1669 .15669 -1.6699 - | . 37 3 . 29 4 . 21 6 . 16 9 . 10 8 5 . 06 7 . 08 7 . 08 7 . 08 8 . 12 17 . 12 6 7 . 32 9 6 . 67 18 . 67 18 . 67 18 . 58 6 . 58 6 | . 408 .3255 .2356 .1746 .1126 .1124 .153 .192 .266 .311 | . 465 . 360 . 213 . 2063 . 1149 . 136 . 127 . 133 . 144 . 265 . 332 . 255 . 322 . 259 . 286 . 330 | .466 .3166 .3166 .1266 .1543 .1447 .1711 .22618 .3311 094 .2211 .2221 | . 479 . 3555 . 311 . 2254 . 1264 . 1455 . 127 . 1107 . 123 . 1252 . 200 . 230 . 209 143 128 128 129 . 195 | .398 .341 .239 .168 .104 .007 .007 .007 .008 .007 .005 .005 .005 .006 .007 .006 .007 .006 .007 .006 .007 .007 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| Ī | | | · · · · · · · · · · · · · · · · · · · | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|--|--|--|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55ს/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.6 | $\alpha = 6.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 12.500 12.500 10.5000 | 0 8 7 8 7 8 7 8 7 8 7 8 7 8 7 8 7 8 7 8 | - 1.0443 - 1.0443 - 1.0443 7529 4450 4450 4410 | 784 99147 - 8827763 - 862763 - 665517224 - 441668 - 352062 - 35550 - 441683 - 1665362 - 1665362 | - 431 - 9947 - 9947 - 8842 - 6365 - 5265 - 5 | 830 981 985 9265 897 744 6638 547 4514 3741 3741 3269 2303 2313 2098 2313 | 741 - 1.9624 9545 9915 8244 7723 6458 3449 2228 2285 2177 1646 1129 | 79221897771221897712218977122189771221897797979797979797979797979797979797979 |
| LOWER SURFACE | 1.25.00 7.05.00 1.05.0 | 3453835835858585858585858585858585858585 | 124 1158 1258 1285 1344 1324 1458 1568 15684 16684 16527 | 4425199 8678 16668 924 4227 382 2297 11168 1833 22907 383 3333 3333 3333 3333 3333 3333 333 | \$128 \$128 \$120 \$120 \$120 \$120 \$120 \$120 \$120 \$120 | .4850 .4402 .3442 .3258 .2584 .2037 .11847 .11899 .2263 .333 .3339 .20118 .20118 .20118 .20118 .20118 | 4906 4390 34391 350354 31776 31144 3114 31144 3114 3114 3114 3114 3114 3114 3114 3114 3114 3114 3114 314 3 | *4195 *3957 *3957 *2299 *1179 *00180 *2001280 *200499 ********************************** |
| | M = 0 | | 1 - 990 | _ 1 143 | - 910 | - 1 024 | - 602 | - 549 |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 20.00 20.00 20.00 30.00 40.00 45.00 50.00 60.00 70.00 80.00 90.00 | - 0 46 - 0 46 - 0 66 - 0 6895 - 0 6895 - 0 514 - 0 488 - 0 428 - 0 428 - 0 4292 - 0 4092 - 0 1891 - 0 18 | - 1.063 981 9182 755 633 487 4453 4433 3867 3067 1521 121 050 | 177 | - 1.306 - 1.300 - 1.284 - 1.284 - 1.300 - 1.274 - 1.316 - 1.274 - 1.316 - 261 - 232 - 213 - 206 - 230 - 1618 - 125 - 126 - 129 | 258 | 692802771785774743666640570495049504092375628632863 | 5479 55479 5547 5535 4765 4291 3317 3317 3317 32873 2273 2251 |
| LOWER SURFACE | 1 . 25 2 2 . 50 0 7 . 50 0 0 15 . 00 0 25 . 00 0 35 . 00 0 65 . 00 0 7 5 . 00 0 85 . 00 0 90 . 0 | 349454 4040438860 38860 22379 118657 11484 22326 114733 11484 1148 | .4699 .3953 .2624 .2960 .11776 .11776 .12007 .1595 .7044 | . 471 . 460 . 4075 . 2393 . 2307 . 2004 . 2004 . 2020 . 20 | .520 .482 .417 .373 .352 .246 .227 .219 .2217 .228 .299 .372 .367 .247 .253 .268 .268 | . 490 . 489 . 452 . 399 . 362 . 2242 . 2212 . 2212 . 2213 . 2817 . 3144 2556 2766 2766 2766 | 178 294 174 203 203 203 166 203 203 1679 200 224 211 | 0197 0 |

CONFIDENTIAL

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| ſ | | - | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|--|--|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85ь/2 | 0.95b/2 |
| | M = 0. | 60 α = 12.1° | <u> </u> | · · | | | | |
| UPPER SURFACE | 1.25 2.500 7.500 15.000 25.000 25.000 35.000 45.000 45.000 65.000 65.000 65.000 90.000 | 004 - 1.253 - 1.341 - 1.053 8252 6944 577 538 516 500 4751 4437 4437 226 1149 114 | - 1.854 - 1.714 - 1.759 - 1.842 - 2.1125 - 2.1215 347 3407 3403 4018 362 1257 0846 0846 | - 1.556 - 1.179 - 1.2218 - 1.2304 - 1.3794 - 1.3501 - 1.3501 - 1.3161 - 1.311698489654542705168316831683 | - 1 . 0 4 6 9 11 9 17 9 07 9 07 9 08 8 6 6 8 5 8 8 5 8 8 5 7 8 0 3 7 4 7 7 12 6 13 8 5 5 6 5 5 9 5 1 8 | 77562246611669257665541055465541751175117549248854889 | - 48 8 - 4420 - 4329 - 427 - 415 - 404 - 3882 - 3761 - 362 - 3539 - 348 - 3339 - 3101 - 2867 - 2878 | 411 - 3549 - 3349 - 33422 - 334322 - 335120 - 335120 - 335120 - 325512 - 32 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 10.00 15.00 25.00 35.00 40.00 55.00 65.00 75.00 65.00 75.00 85.00 95.00 | 2875 .5749 .5559 .4812 .5539 .44672 .22747 .2248 .346970 .2458 | . 467 .547 .515 .478 .386 .307 .268 .268 .272 .291 .4112 .7873 .7873 .4095 .359 | . 4519 .5102 .5137 .3843 .3091 .2667 .2667 .2762 .3773 3992 3992 3944 | .503 .509 .419 .366 .327 .295 .246 .230 .231 .239 .270 .351 | .473 .504 .435 .3425 .2292 .2269 .205 .191 .187 .1230 .2318 .654 .694 .737 .7318 | 471 444 429 384 384 389 201 1161 1127 1125 1127 1125 1127 1127 1127 112 | .408 .408 .3374 .2238 .0044 .0018 .009247 11446 115527 1177 |
| | M = 0 | .85 $\alpha = 0.0^{\circ}$ | 500 | .480 | .715 | .460 | . 453 | .307 |
| UPPER SURFACE | 1.00 1.25 2.50 5.00 10.00 25.00 35.00 35.00 45.00 45.00 65.00 75.00 80.00 95.00 | | 152 | 14711911311421672172292452262927622549254910661765208 | - 223 - 1866 - 1170 - 2167 - 2377 - 2377 - 2447 - 2772 | 3114 2137 22137 22137 22137 22137 2279 2279 2279 22788 22788 22788 22788 22788 22788 22788 22788 22788 - | 426 299 251 244 238 238 271 271 276 275 276 275 276 - | 458 2844 3008 2548 2218 22103 22103 21102 2103 2 |
| LOWER SURFACE | 30.00 35.00 40.00 45.00 | .130 .045 .027 045 067 067 076 076 051 051 051 051 533 533 533 533 533 | 684 705 674 617 | 017 005 022 054 101 168 254 264 364 368 388 | 011043043023010002021010021070176255315298301305 | 064 0630 0112 0114 024 0419 0132 02705 1322 12705 1322 12705 1322 12705 1322 12705 1322 1499 12449 | .073 .097 .066 .051 .021 .028 .033 .063 .057 .150 .2252 .218 184 154 | .106 .00311 .002613 005637 004406 00399 00554 00554 00702 00702 00702 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| | PERCENT | | | PRESSUR | E COEFFICIENT, | P, AT: | - | |
|---------------|--|---|---|--|---|---|---|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85ь/2 | 0.95b/2 |
| | M = 0 | .85 $\alpha = 4.0^{\circ}$ | | | | | | |
| UPPER SURFACE | .00 1.250 5.00 10.000 15.000 25.000 35.000 45.000 65.000 70.000 88.000 95.000 | 0 27 - 0 15667 - 0 20 03 - 0 2 343 - | . 123 | 017 - 1.124 980 628 537 467 462 445 447 446 460 460 384 327 305 382 170 202 219 219 | . 231 - 1.284 - 1.126 803 686 527 5127 479 451 479 451 398 381 306 270 270 258 226 226 216 | 117 - 1.245 - 1.192 - 1.029899805643558468447432418388362324227249225239217161 | 129 - 1.326 - 1.216 - 1.216 - 1.163 - 1.073877706582490426389354321296272240218188188188188188 | 381 - 1.253 - 1.202 - 1.048981902697569455358304282282285025502550241241242 |
| LOWER SURFACE | 1.25 2.50 5.50 10.000 15.000 25.000 35.000 45.000 45.000 555.000 665.000 70.000 80.000 90.000 90.000 | .341 .327 .282 .245 .163 .098 .0973 .0055 .0047 .0911 .2420 .6500 .5075 .2010 | . 375 . 289 . 225 . 1748 . 114 . 089 . 0771 . 081 . 154 . 2313 . 3231 . 3281 676 5652 . 238 | . 404 .331 .258 .2287 .190 .159 .137 .125 .117 .127 .1243 .171 .222 .295 .331 .326 | . 470 . 345 . 261 . 205 . 212 . 179 . 160 . 146 . 142 . 147 . 185 . 235 . 231 . 355 . 263 . 263 . 263 . 263 . 263 . 264 . 264 . 284 . 347 | . 478 . 389 . 320 . 267 . 233 . 199 . 171 . 156 . 150 . 149 . 162 . 187 . 228 . 237 . 320 . 238 . 231 . 241 . 241 . 327 | . 505 .371 .329 .275 .243 .186 .165 .144 .117 .117 .130 .161 .212 .233 .211 | .411 .358 .261 .192 .073 -003 -0027 -038 -047 -069 -089 -089 -085 -100 -115 -120 -1121 |
| | M = 0. | $85 \alpha = 6.0^{\circ}$ | | | | | | |
| UPPER SURFACE | 12.5000000000000000000000000000000000000 | .013 184 308 351 338 360 391 406 411 388 381 409 427 446 461 469 469 370 - | 114 - 1.148 - 1.114911921570465448455464475498519545590692141118087 | 237 - 1.304 - 1.201 951 821 658 573 562 562 569 573 573 573 573 218 218 218 216 | .015 -1.331 -1.221 -1.048968903824766641585513453453453347315277301226 | 378 - 1.380 - 1.380 - 1.389 - 1.389 - 1.245 - 1.95 945 746 637 531 456 388 341 309 260 227 209 203 196 | 324 -1.100 -1.020 -1.029 -1.029 -1.029 801 803 804 8 | 466 832 819 764 736 673 581 526 464 441 415 396 396 367 367 328 328 329 286 |
| LOWER SURFACE | 1.25 2.50 5.50 10.00 10.00 25.00 35.00 45.00 45.00 65.00 65.00 65.00 80.00 90.00 90.00 | 3 73 -3 75 -3 75 -3 75 -3 03 -2 44 -2 03 -164 -103 -123 -123 -123 -123 -123 -123 -123 -12 | .464 .395 .313 .262 .232 .185 .154 .131 .123 .128 .145 .183 .243 .326 .342 .300 | .471 .3351 .2603 .1262 .1762 .1674 .1991 .3109 .346 .3331 -3330 .3533 | 463 410 3345 2881 2881 2814 1182 1189 2114 1182 1189 210 246 303 307 307 2779 2884 364 | 517 44987 33023 2222 2298 33022 2299 2298 334 22555 33 22555 33 22555 33 | 524 4187 3398 22306 11855 11448 1145 12108 2230 174 115622 11792 | . 4 27 . 39 30 . 2 40 . 1 80 . 1 12 . 00 7 . 00 7 . 00 54 . 00 69 . 07 8 . 08 6 . 08 8 . 10 6 . 111 . 113 9 . 114 5 . 16 3 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| | DEDGENT | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | |
|---------------|--|--|--|---|--|--|---|--|--|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.556/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 | | | |
| | M = 0.8 | $\alpha = 8.0^{\circ}$ | | | | | | L | | | |
| UPPER SURFACE | 1000 1250 57.500 10.000 25.000 35.000 40.000 40.000 40.000 55.000 65.000 75.000 85.000 95.000 | 0 1 2 3 3 4 3 9 5 4 4 9 5 4 8 2 8 6 6 2 4 9 9 5 5 3 5 6 6 6 2 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 | - 1.24 - 1.3559 - 1.96125 - 1.99125 99125 55472 555472 55866 66886 68246 1066 024 | 433 - 1.324 - 1.204 - 1.082 985 860 8625 7719 612 5018 515 515 226 226 220 | - 1086 - 1.4623 - 1.314 - 1.234 - 1.234 - 1.027 027 0864 029 459 459 459 3997 394 3067 2216 2216 2216 2216 | 595 - 1.177 - 1.168 - 1.110 - 1.091 - 1.002 908 866 753 762 6651 500 531 459 432 395 432 3964 310 | 40 2 731 680 6997 690 6317 55847 5543 5500 4764 4464 4463 3349 3349 3349 | 434 544 537 510 522 516 493 464 438 438 437 375 375 340 375 340 330 302 302 302 302 302 | | | |
| LOWER SURFACE | 1.250 5.500 105.000 12 | .3916.44502.4450391.224093.1154.12602.2469.21763.1764.18676.24818.7958.11036 | 54794923 54794923 547949227 5479413 547949227 5479 5479 5479 5479 5479 5479 5479 547 | .513 .473 .407 .370 .288 .289 .209 .209 .207 .211 .230 .264 .370 .368 .370 .319 328 .373 | . 465 . 459 . 396 . 360 . 342 . 296 . 2244 . 222 . 223 . 223 . 235 . 367 . 387 275 292 . 308 317 . 297 | . 528 . 490 . 447 . 390 . 357 . 301 . 2270 . 245 . 225 . 226 . 226 . 252 . 301 . 348 . 349 . 349 . 349 . 353 . 353 . 353 . 353 . 355 . 355 | . 523 . 447 . 424 . 373 . 271 . 225 . 178 . 144 . 160 . 199 . 232 . 216 257 273 296 | . 439 . 420 . 332 . 279 . 213 . 143 . 013 - 017 - 081 108 117 117 129 148 157 183 189 197 | | | |
| | M. = 0. | $\alpha = 12.1^{\circ}$ | | | | | | | | | |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 25.00 35.00 45.00 50.00 60.00 70.00 85.00 95.00 | .002 -539 -730 -794 -828 -783 -752 -752 -644 -636 -635 -619 -561 -510 -497 -497 -444 -425 -336 -233 | - 1.461 - 1.4624 - 1.424 - 1.3995 - 1.3755 - 1.3558 - 1.3688 - 1.5275 - 1.3688 - 1.5275 - 1.5275 - 1.5275 - 1.5285 - 1.5285 | 838 - 1.165 - 1.118 - 1.103 - 1.083 - 1.083 - 1.083 - 1.029 - 1.029 - 1.029 - 969 - 945 - 945 - 8850 - 8850 - 8850 - 740 - 767 - 7709 - 675 - 625 - 625 - 470 | 515 845 835 826 835 825 825 814 774 767 745 728 710 665 630 620 563 563 563 | 72971570269568668166466276205875764554521521483487 | - 456 - 487 - 458 - 475 - 473 - 452 - 434 - 420 - 412 - 408 - 391 - 391 - 386 - 385 - 385 - 385 - 338 - 338 | 372 345 3154 3217 3103 2880 2880 2660 2660 2667 2663 2663 2663 2663 2679 270 | | | |
| LOWER SURFACE | 1.25 2.55 3.50 10.00 10.00 15.00 25.00 35.00 45.00 45.00 60.00 60.00 80.00 90.00 90.00 | 351 507 620 5575 498 434 379 313 2857 2257 2254 339 534 534 577 281 281 290 69 | . 57 4 . 57 4 . 57 23 3 . 4 4 3 8 . 3 4 3 8 . 3 4 3 8 . 3 2 8 8 . 2 2 7 7 5 . 2 8 2 9 . 3 2 8 9 . 3 3 6 8 . 3 3 5 8 . 3 5 8 . 3 5 8 . 3 6 8 9 . 3 7 6 . 3 7 6 . 3 7 8 9 . 3 7 8 9 . 3 7 8 9 . 3 8 9 . 3 8 9 . 3 8 9 . 3 7 8 9 . 3 8 | 526 543 5507 4432 383 340 3210 2254 2254 2259 2276 3778 | 495 499 471 429 417 364 2296 2242 233 249 379 379 - 679 - 679 - 6927 - 371 | .509 .512 .484 .435 .4401 .3497 .2657 .2214 .2021 .2021 .2638 .3242 .6335 .6235 .6235 .6235 .6235 .6342 | .504 4464 4453 4408 .371 .302 .266 .230 .192 .166 .148 .144 .242 .221 .390 .3868 .3553 .192 | .441 .4371 .314 .259 .190 .104 .056 .0022 -0036 -0077 -1162 -178 -1188 -1191 -188 -191 -187 -201 | | | |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| Ī | | PRESSURE COEFFICIENT, P, AT: | | | | | | | | | | | | | |
|---------------|---|---|--|-----|---|-----|--|----|--|----------|---|-----|---|----|---|
| | PERCENT CHORD | 0.13 | 5b/2 | 0.2 | 5b/2 | 0.4 | 0b/2 | 0. | 55b/2 | 0.7 | 70b/2 | 0.6 | 35b/2 | 0. | 95b/2 |
| UPPER SURFACE | M = 0.90 1.250 5.000 10.000 15.000 25.000 35.000 45.000 45.000 65.000 65.000 80.000 90.000 | , | 0.1° 0.345 0.195 0.345 0.195 0.364 0.172 0.795 0.1112 1.1345 1.1631 1.2266 1.22445 1.3766 1.2266 1.2266 | | .526 .075 .053 .054 .057 .087 .087 .1132 .1174 .1132 .1242 .2241 .2242 .2348 .328 .328 .328 .046 .046 .046 .046 .046 .046 .046 .046 | | .486 .128 .103 .102 .121 .131 .131 .135 .216 .235 .284 .335 .257 .244 .335 .257 .244 .355 .183 .255 .183 .255 .183 .255 | | .730 .202 .175 .1150 .150 .217 .2217 .260 .260 .260 .260 .277 .286 .277 .286 .277 .286 .277 .286 .286 .277 .286 .286 .286 .286 .286 .286 .286 .286 | | 4345245292258786287373736287822222222222222222222222 | | . 460 . 422 . 314 . 250 . 260 . 2654 . 2773 . 3123 . 3337 . 3069 . 2310 . 2539 . 2310 . 2539 . 2310 . 2310 | | .320 .489 .389 .291 .326 .3350 .3553 .2232 .2337 .215 .145 .145 .106 .1089 .090 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 25.00 35.00 35.00 45.00 45.00 65.00 70.00 85.00 95.00 95.00 | | 0253 02655 03552 0352 03 | | .046 .0337 .037 .0518 .052 .072 .078 .081 .089 .036 .099 .016 .099 .288 .271 .236 .688 .706 .628 .676 .628 .539 .193 | | .046 .034 .029 .038 .028 .021 .000 .057 .172 .257 .257 .287 .289 .377 .382 .391 .395 .312 | -, | .009 .012 .038 .042 .0208 .0019 .024 .074 .1261 .263 .337 .324 .304 .305 .305 .314 | | .1658 .0658 .0051 .0013 .0013 .0013 .0024 .0013 .0024 .0023 .003 .003 .003 .003 .003 .003 .00 | | 212 0790 0998 0049 0018 0025 0025 00378 00594 1418 22423 11944 11768 11640 | | 1170 1170 1079 100309 10060 10 |
| ļ | M = 0.9 | 00 α = | = 4.0° | I | 01.4 | | .090 | | .341 | <u> </u> | .004 | - | .022 | | .280 |
| UPPER SURFACE | .00 1.25 2.50 5.00 7.50 10.00 25.00 35.00 40.00 45.00 65.00 70.00 80.00 95.00 | | 277 2733 2298 3344 33753 4113 55786 66 | | .224 .150 .139 | | 981065 91665 47518 4418 4418 4424 45025 55175 2956 2847 2847 2847 | - | 1.151 1.121 .954 .4694 .481 .495 .490 .511 .5583 .583 .452 .272 .273 .2429 .2229 | - | 1.22976 1.22076 1.22076 1.22076 1.22076 1.22070 1.2207 | - | 1.2280790809751897037374 | _ | 1.2899 1.28626991 1.1626991 1.10891 1.0891 1 |
| LOWER SURFACE | 60.00 | | 3540 34946 22466 2270 1335 0756 0555 0459 0925 4333 6685 47959 145 | | .370 .224 .1817 .115 .072 .0782 .1125 .082 .1125 .239 .325 .325 .3263 .657 .695 .5763 .218 | | 388 316 243 213 1160 1147 1127 1117 1163 2287 3287 3287 3528 3528 3528 3538 3538 | | . 284 . 290 | - | . 244 | | . 495 . 325 . 2316 . 2316 . 1136 . 11316 . 11099 . 11619 . 12233 . 11629 . 116 | | .3482 .3188 .105343 .00789 .00992 .00992 .00992 .00992 .1127 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| ſ | | PRESSURE COEFFICIENT, P, AT: | | | | | | | |
|---------------|--|---|--|---|--|--|--|---|--|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40ъ 2 | 0.55b/2 | 0.70b/2 | 0.85ъ/2 | 0.95b/2 | |
| UPPER SURFACE | M = 0.90 1.25 2.50 5.00 7.50 10.00 25.00 25.00 35.00 45.00 45.00 65.00 65.00 75.00 85.00 | α = 6.00 .0211042322812813293183433573560339360339360360360360360370370360370 | 1.014 1.062 1.063 841 656 415 415 413 426 443 486 489 547 442 452 452 410 | 103 - 1.197 - 1.1973 953 788 683 593 534 531 538 550 5683 5683 528 528 532 532 533 5360 528 532 332 | 1 142 1 135 1 1 241 1 1 137 1 1 069 - 866 - 656 - 6575 - 6808 - 6806 - 675 - 6806 - 675 - 6806 - 357 - 391 - 391 - 391 - 221 | 214 - 1.391 - 1.385 - 1.294 - 1.218 - 1.1047 - 1.9557557557672667266841932527222522582309 | 230 -1.312 -1.231 -1.252 -1.201 -1.160 -1.104 -1.029 -971 -883 -717 -648 -589 -525 -445 -337 -292 -274 -243 -1191 -172 | 470 -1.060 -1.052 -985 -985 -965 -796 -726 -726 726 726 736 530 519 498 447 447 4403 3383 3363 3344 3318 | |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 20.000 30.000 40.000 450.000 450.000 450.000 450.000 450.000 905.000 905.000 905.000 | .3936 .4160 .3347 .32200 .2227 .1434 .11183 .11366 .462967 .4668 .4668 .1829 | .470 .399 .318 .238 .192 .1609 .1311 .136 .159 .261 .343 .290 -678 -678 -586 -433 .243 | .467 .404 .3293 .2537 .1891 .1567 .1690 .2323 .346 .3588 3669 3689 | . 471 . 391 . 318 . 267 . 2266 . 2026 . 184 . 1775 . 181 . 245 . 3757 - 2673 - 288 - 274 . 275 | .506 .4373 .3282 .2841 .2194 .1881 .1910 .2248 .33410 .2338 .2454 .2454 | .514 .402 .370 .284 .226 .176 .153 .141 .136 .141 .163 .2310 .2310 .158 .158 .1666 .177 | . 4 2 6 . 3 9 5 . 3 9 5 . 2 3 8 . 1 7 0 5 . 0 0 4 0 6 . 0 8 4 0 . 0 9 8 . 0 9 9 . 0 9 9 . 0 9 9 . 1 1 2 7 . 1 1 4 2 6 . 1 1 5 5 . 1 1 7 1 | |
| UPPER SURFACE | M = 0.2 1.25 2.50 5.00 10.00 15.00 20.00 30.00 40.00 45.00 55.00 65.00 65.00 65.00 85.00 90.00 | 90 | 177 - 1.121 - 1.097986914861790682569480468554534560566505511291 | 277 - 1.218 - 1.141999859859737714680667663657608522434319252 | 038 - 1.444 - 1.365 - 1.300 - 1.238 - 1.915 870 762 762 762 762 763 470 379 328 419 350 296 279 | 425 - 1.211 - 1.218 - 1.181 - 1.152 - 1.139 - 1.002 919 876 660 660 660 603 5024 510 436 436 436 436 | 352 864 791 819 812 782 7680 6641 5965 5487 5487 5053 471 387 3855 3557 | 446 639 628 591 604 631 523 523 521 497 481 431 431 431 340 340 3320 313 | |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 15.000 25.00 35.00 45.00 50.00 55.00 65.00 70.00 85.00 90.00 | . 4 06 . 4 68 . 4 75 . 4 34 . 4 38 . 2 3 38 . 2 2 45 . 1 172 . 1 152 . 1 152 . 1 152 . 1 2 82 . 1 2 82 . 1 3 79 . 2 79 . 1 106 | . 527 . 474 . 396 . 345 . 311 . 259 . 222 . 196 . 183 . 184 . 196 . 229 . 228 . 355 . 370 . 317 | .511 .464 .394 .318 .215 .242 .218 .200 .196 .218 .2218 .321 .363 .363 -384 -389 -398 -398 -398 -398 | . 476 . 441 . 376 . 336 . 325 . 227 . 227 . 226 . 211 . 205 . 206 . 223 . 256 . 313 . 383 . 370 346 351 321 321 321 | . 523 . 472 . 426 . 369 . 3314 . 249 . 212 . 203 . 206 . 218 . 247 . 298 . 344 . 354 . 3575 . 3755 . 347 | .521 .432 .409 .359 .3257 .228 .1798 .1750 .153 .1952 .208 2864 2844 2984 292 .186 | .435 .415 .331 .272 .2119 .045 045 062 077 119 1134 1134 1156 1264 | |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| | DEDCENT | | | PRESSUR | E COEFFICIENT, | | | |
|---------------|--|--|---|--|---|---|--|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.90 | $\alpha = 12.1^{\circ}$ | | | | | | L |
| UPPER SURFACE | 1.25 2.500 7.500 15.000 25.000 25.000 30.000 40.000 50.000 60.000 75.000 60.000 75.000 95.000 | 001 426 6619 6932 732 684 6839 5579 6620 6635 644 649 7668 7668 4908 322 | 578 - 1.354 - 1.352 - 1.318 - 1.287 - 1.252 - 1.241 - 1.130 - 1.2575975975985845847474103103 | 660 - 1.233 - 1.164 - 1.150 - 1.130 - 1.120 - 1.027 - 1.0047 - 1.041 993 9941 9941 995 732 732 732 732 732 732 695 | 4528658728818902853580927717557106852633963396000585 | 673 700 694 682 674 6654 6654 6413 622 590 570 550 5539 539 539 522 499 471 | 482 524 493 507 505 490 473 462 453 445 431 425 425 425 425 425 425 425 425 425 425 431 425 431 425 431 425 431 425 431 425 436 370 370 370 370 370 375 356 356 | 423 384 385 375 355 355 355 336 336 323 322 311 305 |
| LOWER SURFACE | 1.25 25.50000 70.50000 225.0000 225.0000 450.0000 450.0000 760.000 890.000 995 6AP | 378 534 6185 5044 3842 3842 3842 3842 3842 3842 3843 3842 3843 3843 | . 59 4 . 58 5 . 5 28 . 48 1 . 39 0 . 31 6 . 29 4 . 28 3 . 30 1 . 33 3 . 41 9 . 7 2 7 - 1 . 0 5 6 . 3 2 8 . 3 2 8 . 3 3 3 . 3 3 4 . 3 5 7 . 3 6 8 . 3 7 4 | . 546 . 549 . 505 . 477 . 435 . 3146 . 314 . 286 . 271 . 263 . 263 . 263 . 263 . 384 . 384 . 384 . 384 . 384 . 384 . 384 . 386 . 334 . 386 . 334 . 386 . 334 . 386 . 386 | . 498 . 497 . 467 . 428 . 293 . 293 . 248 . 235 . 235 . 255 . 377 693 7032 . 374 | .515 .485 .435 .436 .270 .2243 .2210 .2210 .2217 .227 .2313 .2313 .247 .250 .2635 .635 .635 .635 .344 | . 498 . 457 . 450 . 404 . 369 . 302 . 227 . 1991 . 167 . 148 . 141 . 150 . 226 . 227 448 . 2236 . 236 . 3361 . 3361 . 3361 | . 435 . 442 . 374 . 313 . 262 . 1997 . 045 - 027 - 1062 - 1163 - 212 - 212 - 224 - 224 - 225 - 235 - 235 |
| | M = 0. | | | | | · - · · · · · · · · · · · · · · · · · · | r | |
| UPPER SURFACE | 12.500 12.500 10.000 | .023 .2047 .0746 .028 .013 .0668 .01077 .1127 .1184 .2053 .22799 .2267 .2267 .22581 | . 533 0749 0547 084 111 1280 1711 1291 237 2366 279 3709 3709 3709 1868 1868 1868 1885 1836 | . 496 1083 085 1183 1147 1722 2202 2422 271 299 338 370 415 425 250 271 | 7 3 9 1 1815 1 01976 1 1675 1 1675 1 1675 1 2348 1 2368 1 3765 1 3715 1 4419 1 3237 1 3247 1 2446 | . 476 2933 2240 2258 2258 3375 3375 33928 4337 4337 4337 4337 2852 28513 | . 468 4328 2891 2891 3005 3305 3414 3777 3783 3783 401 2491 2491 1440 1443 | . 3352 |
| LOWER SURFACE | 1.25000000000000000000000000000000000000 | .240 .174 .101 .060 .039 025 074 075 084 084 021 .174 360 360 496 4461 368 | 040 033 053 053 064 099 098 044 106 209 2297 .2616 7759 7759 7728 6722 5999 .174 | 0537 0381 0311 0411 037 021 0100 .022 .096 .162 .244 .285 443 433 433 433 433 | . 015 006 028 024 012 017 090 .023 .097 .164 .329 329 329 330 334 | .129 .036 .031 .003016 .013 .003 .004 .124 .188 .271 .325283286283278283 | . 191 . 047 . 083 . 053 . 0036 . 003 . 0017 . 0016 . 0242 . 080 . 135 . 208 . 236 . 206 . 206 . 206 . 209 199 199 196 1985 | .160 .1113 .043 .0030 030 138 094 062 056 073 094 108 084 093 118 131 131 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| Ī | DDD GDVM | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|--|---|---|---|---|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70ь/2 | 0.85ъ/2 | 0.95b/2 |
| UPPER SURFACE | M = 0.900 1.25 2.50 5.00 10.00 15.00 25.00 30.00 35.00 45.00 45.00 65.00 65.00 80.00 90.00 | 4 α = 4.0° .024 .0430611231691832442242249273315315355355 | . 273 | .17189465234083723823723914014484474484544544544543583366356 | . 430 - 1 .013 997 8620 4351 4457 457 457 472 472 5538 578 578 578 578 578 5331 4451 431 | .115 -1.061 -1.059 936 889 8559 474 5509 5541 556 586 586 586 645 641 605 489 489 489 487 | .093 -1.102 -1.062 -1.062 -1.961929882790553580605605605673715280220255184171137 | - 151 - 1.133 - 1.123 - 1.00218 - 1. |
| LOWER SURFACE | 1.25 2.50 7.500 7.500 15.000 25.000 25.000 40.000 40.000 50.000 60.000 60.000 80.000 90.000 90.000 | .366 .350 .302 .258 .231 .179 .144 .106 .077 .069 .043 .043 .060 .098 .239 .445 .1057 .317 .396 | .361 .290 .221 .175 .145 .107 .083 .066 .065 .080 .116 .251 .314 .315 .250 763 817 755 673 554 | .377 .306 .237 .206 .176 .144 .125 .116 .105 .125 .125 .125 .125 .201 .271 .309 .309 .305 .438 .434 .434 .434 .438 | . 442 . 301 . 218 . 171 . 179 . 144 . 127 . 111 . 107 . 115 . 155 . 208 . 275 . 339 . 322 332 335 336 . 337 | .444 .338 .276 .224 .193 .160 .135 .124 .119 .125 .141 .171 .218 .280 .312 .305 278 276 278 278 278 .316 | .468 .323 .285 .237 .204 .148 .134 .116 .101 .094 .1094 .110 .142 .192 .208 .183 184 184 189 192 | .3999 .3236 .1660 .10482 00598 112362 114632 16732 11498 1147 |
| | M = 0. | | | 1 | 248 | 075 | 0.25 | 3.43 |
| UPPER SURFACE | 105 250 250 250 250 250 250 250 2 | 0193 | . 093 - 1.024 - 1.042 775 479 416 397 387 387 387 453 - | 006 - 1.176 - 1.147 - 1.076 971 556 4656 4654 4654 4544 5400 5560 5660 5660 5600 5380 3882 3882 | 2470 -1.2450 -1.23515 -1.09515 -1.09515 55167 5516 | - 1.2751 - 1.2877 - 1.19522 - 1.19522 - 1.0377 - 1.0377 - 1.66078 6637 66537 66537 6773 371357 371357 371357 | - 1.306 - 1.214 - 1.236 - 1.152 - 1.152 - 1.055 - 1.055 - 1.025 - 1.026 - 1.014 9766 7667 536 336 - | - 1.3436 - 1.3518 - 1.3188 - 1.2194 - 1.0976 - 1.0046 - 1.0935 - 1.0936 - 7766 7766 7664 5538 3418 3418 3418 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 45.00 65.00 65.00 60. | . 4 068 . 4 397 . 3 358 . 2 2 2 8 . 1 8 5 0 . 1 3 8 . 1 1 0 4 . 1 1 4 6 . 1 4 6 9 . 4 4 8 7 . 7 4 7 4 | . 470 . 379 . 3269 . 2366 . 1958 . 1368 . 1358 . 1368 . 1368 . 2074 . 3534 . 2774 . 3534 . 2774 . 3534 . 2780 . 27 | . 469 . 327 . 293 . 218 . 1190 . 1756 . 157 . 163 . 120 . 294 . 332 . 446 . 445 . 445 . 342 | . 493 . 379 . 351 . 251 . 187 . 167 . 155 . 154 . 186 . 291 . 344 . 344 . 344 . 344 . 344 . 352 . 352 | .490 .411 .346 .2958 .220 .187 .170 .160 .1712 .235 .2394 .331 .380 277 271 261 | .5070 .3529 .2667 .1844 .1142 .1288 .1352 .2080 .2080 .204 | . 424 .383 .293 .273 .1005 0057 1207 1207 1564 1564 1688 1668 1668 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| | PERCENT | | | PRESSUE | RE COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|--|--|--|---|--|
| | CHORD | 0.135b/2 | 0.,25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95ь/2 |
| | M = 0.9 | $\alpha = 8.0^{\circ}$ | | | | | L | L |
| UPPER SURFACE | 1.00 1.250 5.00 10.00 10.00 20.00 20.00 30.00 45.00 45.00 45.00 65.00 75.00 85.00 95.00 | 015 - 1379 - 1379 - 13348 - 13941 - 13945 - 14078 - 14078 - 14378 - 14436 - 14559 - 14556 - 14644 | 066 -1.086 -1.109 8769 87993 5677 4777 4298 493 493 4965 534 534 4965 4913 4 | - :164 - 1:1638 - 1:1638 - 1:0935 - :0935 - :0856 - :6760 - :6635 - :6635 - :6636 - :5998 - :6013 - :6618 - :6617 - :5892 - :5802 - :4164 - :387 - :3551 | 7079 -1.324 -1.249 -1.210 -1.175 -1.000 -1.175 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 -1.000 -1.1872 - | 270 - 1.1030 - 1.090 - 1.0645 - 1.0045 - 1.00589839847881178374337623545559655865844506 | - 2444 - 18148 - 77835 - 77835 - 77861 - 77741 - 77414 - 77414 | - 468 - 4725 - 47715 - 76675 - 766706 - 766675 - 766675 - 766625 - 766625 - 766655 - 766655 - 766655 - 7666625 - 7666625 - 766655 - 76665 - 76 |
| LOWER SURFACE | 1.250 5.050 10.000 25.000 45.000 45.000 65.000 65.000 65.000 90.000 90.000 90.000 | 42912344923449234425886666887444233622816857757570435570435570575704357043 | . 544 . 484 . 484 . 355 . 2275 . 2275 . 2297 . 1296 . 1294 . 2251 . 3789 . 3789 . 3799 . 3799 . 36860 . 270 | . 523 . 471 . 402 . 367 . 283 . 2850 . 2206 . 2200 . 2216 . 2316 . 3362 . 3467 . 467 . 467 . 472 . 368 | - 494 - 445 - 373 - 3730 - 2720 - 2720 - 2720 - 2720 - 2730 - 27300 - 273 | .533 .468 .420 .359 .327 .244 .216 .202 .194 .219 .242 .242 .344 .344 .344 .392 .403 .344 .343 .345 | .534 .427 .404 .317 .2523 .1193 .1146 .1151 .1151 .1225 .229 .2866 .2886 .2899 .186 | 4 2 3 4 7 1 3 9 6 4 9 2 1 4 2 3 4 7 1 3 9 6 4 9 2 1 1 2 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 |
| | M = | $0.98 \qquad \alpha = 0.0^{\rm O}$ | | | _ | | | |
| UPPER SURFACE | 1.25 2.55 5.00 10.000 15.000 25.000 35.000 45.000 45.000 60.000 70.000 85.000 80.000 90.000 | .048 .244 .187 .0793 .070 .0321 044 075 084 166 179 227 227 227 227 227 | .5610150020030230370683107150150150150150151 | .5110420280370570741271561731272202803483483483481831831831845247 | .767101091049084107153180218218248354354354397464170231240270 | . 494201157162201213223273273289315347418448448392424225280297331 | . 48729021522432443249334358419500506419320380381380381380381381381 | |
| LOWER SURFACE | 1.25 2.500 10.000 10.000 20.000 20.000 20.000 45.000 45.000 65.000 75.000 85.000 85.000 90.000 87.000 87.000 87.000 87.000 87.000 87.000 87.000 | 271 206 206 2071 2071 2071 2071 2075 2075 2075 2075 2075 2075 2075 2075 | 0 2 2 2 00 9 9 01 5 03 0 9 01 5 03 0 9 07 4 08 9 01 8 25 2 . 33 0 . 29 9 . 22 9 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 22 9 9 22 9 | 069045037029019008023057119262296538542539537533 | 009 021 031 023 010 007 002 .005 .015 .034 .059 .103 .165 .249 .327 .318 446 446 446 446 | .102 .019 .023 .022 .022 .021 .027 .027 .037 .037 .037 .313 .313 | . 132 004 .004 .015 .005 0013 005 001 .010 .024 .053 .107 .188 .227 .199 316 281 281 287 .197 | |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| Γ | | | | PRESSU | RE COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|---|--|---|--|--|
| ' | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| OFFER SORFACE | 25.00 30.00 40.00 45.00 55.00 65.00 70.00 | 8 \(\alpha = 4.0^\) 0 \(0.045 \) 0 \(0.097 \) 0 \(0.077 \) 0 \(0.077 \) 1 \(1.137 \) 1 \(1.137 \) 1 \(1.137 \) 1 \(1.137 \) 2 \(1.124 \) 2 \(2.044 \) 2 \(2.044 \) 2 \(2.249 \) 2 \(2.266 \) 2 \(2.977 \) 3 \(1.374 \) 3 \(1.450 \) 4 \(1.550 \) 3 \(1.450 \) 4 \(1.550 \) 3 \(1.450 \) 3 \(1.450 \) 3 \(1.450 \) 3 \(1.450 \) 3 \(1.450 \) 3 \(1.450 \) 4 \(1.550 \) 3 \(1.450 \) 4 \(1.550 \) 3 \(1.450 \) 4 \(1.550 \) 5 \(1.450 \) 5 \(1 | 314 339 345 364 390 847 264 300 287 256 | - 4848 - 3223 - 31117 - 33737 - 33707 - 37707 - 41718 - 4771 - 3797 - 3797 | - 465 - 492 - 511 - 516 - 514 - 430 - 416 - 3775 - 389 | 411 574 518 470 460 431 | | |
| LOWER SURFACE | 1.25 2.500 57.500 105.000 205.000 305.000 405.000 405.000 505.000 805. | 387 3726 3286 2890 2052 2158 1130 1000 091 076 065 272 272 477 1025 - 1025 - 1025 | . 166 . 130 . 104 . 087 . 107 . 1087 . 143 . 288 . 366 . 338 . 264 | . 200 . 165 . 139 . 133 . 147 . 2192 . 3217 | .3128 .128 .1992 .1157 .132 .1157 .127 .125 .2668 .3302 .3102 .4459 .4459 | .186 .149 .122 .110 .103 .105 .121 | .080 .075 .082 .1765 .1699 | 380 318 323 1157 0428 - 11145 - 1145 - 1145 - 1145 - 1255 - 2248 - 2248 - 2248 - 2248 - 2248 |
| Į | M = 0.9 | $\alpha = 5.9^{\circ}$ | | .079 | .324 | .020 | .031 | 210 |
| UPPER SURFACE | 1.250 5.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.00000 10.0000 10.0000 10.0000 10.0000 10.0000 10.0000 10.00000 10. | 267 2867 303 349 349 379 361 | 846 597 374 333 331 346 372 346 372 336 372 336 336 336 336 | | - 1.038 - 1.026 - 951 - 873 - 799 737 468 468 468 515 562 5762 | - 1.0564 - 1.0664 - 1.0864 - 1.9847 - 1.9849 - 1.8499 - 1.7949 - 1.55359 - 1.55359 - 1.56356 - 1.5636 - 1.5636 | - 1.071 - 1.024 - 1.023 980 961 923 8698 792 780 798 674 674 630 547 414 630 396 396 547 547 546 5547 5547 | - 1.0893 - 1.09853 - 1.09758 - 1.09758 - 1.09758 - 1.09758 - 1.09758 - 1.09767 - 1.097 |
| LOWER SURFACE | 1.25 25.500 70.000 15.000 25.000 35.000 45.000 45.000 65.000 65.000 70.000 80.000 90.000 90.000 90.000 | . 422 . 444 . 244 . 234 . 244 . 20 . 166 . 153 . 123 . 129 . 249 . 290 . 200 . 200 . 200 . 200 | 3 | . 4036 . 3063 . 22634 . 2276 . 2276 . 2276 . 2276 . 2276 . 3374 . 3376 . 3376 . 3776 . 3776 | .389 .268 .217 .1192 .1193 .1155 .1165 .284 .224 .224 .333 .334 .443 .4452 .4452 .4462 .4462 | . 401 .339 .285 .2509 .1769 .146 .145 .155 .177 .220 .281 .318 .314 .318 .314 .318 .314 .318 .314 .318 .319 .4418 .4427 .4430 .430 | . 370 . 382 . 249 . 192 . 1746 . 127 . 114 . 1110 . 138 . 139 . 1214 . 190 326 307 307 308 169 | 3285 2224 1173 - 0079 - 1103 - 1160 2760 2777 2884 3003 3008 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| • | PERCENT | | | PRESSU | RE COEFFICIENT, | , P, AT: | | |
|---------------|--|---|--|--|--|---|--|--|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 0.98 | $\alpha = 7.9^{\circ}$ | | | | I | | |
| UPPER SURFACE | .00 1.25 2.550 5.00 7.50 10.000 25.000 35.000 35.000 45.000 50.000 50.000 70.000 70.000 85.000 85.000 | 0293 02114 021148 021148 02140 | .015 -1.026 -1.058 -1.058 -1.058 -1.058 -1.058 -1.0616 | 079 - 1.120 - 1.109 - 1.068 - 1.0178604522487543553564553564573564573564393392 | 589 422 658 580 | 167 - 1.189 - 1.191 - 1.124 - 1.079 - 1.033 - 1.007 - 1.003934902884823739665587717610561561561564 | 155 - 1.178 - 1.037 - 1.133 - 1.100 - 1.076 - 1.03398496290286487777246856766416266537576541 | 398 929 9147 879 869 839 8007 759 7522 7722 7685 662991 55988 55644 |
| LOWER SURFACE | 1.25 2.50 5.00 10.00 15.00 25.00 35.00 45.00 45.00 65.00 70.00 85.00 90.00 90.00 | 4509 55081 4400 55081 4400 53080 280800 280800 280800 280800 280800 280800 280800 280800 2808 | .555 .418 .3735 .2846 .2214 .2177 .22773 .4377 .4377 .4377 .4377 .4378 .7865 .7865 .716 | 53 28 408 407 407 407 407 407 407 407 407 407 407 | 506 447 575 5775 5772 5772 5772 5772 5772 5 | .536 .465 .413 .3516 .267 .232 .211 .199 .1916 .1899 .244 .344 .344 .344 .427 .437 .437 .436 | .523 .418 .395 .344 .310 .244 .192 .164 .135 .135 .150 .221 .202 .424 -405 -425 -444 -468 .175 | . 4368 . 4370 . 4370 . 4499 . 00915049 . 009150796 . 1150796 . 223334546 . 22335456606 |
| | M = 1.0 | | | | | | | |
| UPPER SURFACE | 60.00 65.00 70.00 80.00 85.00 90.00 95.00 | 069 2808 1118 092 0049 0024 0044 0057 0094 0057 0094 1047 1047 1057 1048 1048 1048 1049 1049 1049 1049 1049 1049 1049 1049 | .017 .015 .007 .020 .050 .066 .089 .1134 .1134 .156 .217 .2217 .2217 .2217 .2217 .2217 .2217 .2217 .2217 | . 523 - 028 - 013 - 020 - 024 - 058 - 109 - 1154 - 178 - 223 - 230 - 260 - 260 - 285 - 515 - 180 - 152 - 180 - 168 - 199 - 227 | 075 078 078 078 137 164 192 200 213 259 259 318 385 385 385 385 385 227 227 227 | 312 | 284 203 202 202 232 264 292 309 337 364 399 432 364 399 432 364 359 362 362 362 363 | |
| LOWER SURFACE | 1.25 2.50 5.00 7.50 10.00 20.00 25.00 35.00 40.00 45.00 65.00 70.00 65.00 70.00 85.00 90.00 90.00 90.00 | 289 2826 2158 2158 20027 | .000 .010 .0009 .0022 .045 .0778 .0078 .0078 .0078 .264 .2714 .884 .885 .886 .264 | 049 028 028 008 010 .030 .046 .076 .076 .139 .200 .277 .308 .313 532 532 532 532 532 532 | 009 001 0014 0016 0017 0023 0017 0023 0034 0051 118 1263 3341 3333 - 456 - 458 468 476 347 | .133 .045 .047 .022 .001 .002 .001 .008 .027 .046 .0117 .181 .270 .329 .330 .424 .427 .430 .431 .431 .432 | . 157 . 019 . 072 . 039 . 027 - 004 . 009 . 015 . 020 . 030 . 041 . 067 . 118 . 200 . 240 . 217 - 337 - 329 - 329 - 323 . 213 | 1960254229538000437281849 10000175617788000437281849 1000017561778878878878878878878878878878878878878 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| | DEDGENE | | · . | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|--|---|---|---|--|---|---|
| | PERCENT | 0.135b/2 | 0.25b/2 | 0.40ь/2 | 0.55b/2 | 0.70ь/2 | 0.85b/2 | 0.95b/2 |
| | M = 1. | $\begin{array}{cccccccccccccccccccccccccccccccccccc$ | .359 | .262 | .511 | .214 | .203 | 013 939 |
| UPPER SURFACE | 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.0000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.000 10.00000 10. | 0246 - 0461 - 0912 - 01123 - 1170 - 1189 - 1289 - 2249 - 22954 - 22954 - 22954 - 22954 - 22954 - 22954 - 22954 - 22954 - 22954 | 6018265024402245023342356223562231532173620234442254 | 7 28 2 29 9 2 29 9 2 29 9 2 29 9 2 29 9 2 3126 3126 346 9 346 9 421 444 0 8 425 341 288 2 30 4 | | 892 7848 7193 3917 4456 4701 5527 5567 5669 5622 4640 4640 4640 | 88 63 | 998447552879915700448655 998686775866704787555559 |
| LOWER SURFACE | 1.25 2.500 10.500 15.000 25.000 25.000 35.000 45.000 45.000 65.000 70.000 80.000 80.000 95.000 | . 401 .3944 .3076 .2237 .11509 .11095 .0822 .1452 .4799 .4990 .4990 .4990 .3323 | .394 .326 .257 .2187 .1187 .1524 .1005 .1266 .1266 .1230 .314 .387 .3881 883 | . 400 .333 .266 .238 .208 .184 .159 .155 .155 .155 .154 .238 .309 .338 .334 | . 455 . 330 . 245 . 207 . 1759 . 1144 . 136 . 137 . 174 . 286 . 3428 . 3428 . 3428 . 34454 . 4648 . 4668 . 3344 | .453 .347 .287 .205 .806 .1712 .122 .123 .139 .167 .215 .277 .3103 420 432 440 445 .311 | . 465 .314 .281 .204 .148 .131 .114 .099 .091 .101 .131 .183 .203 .181 346 348 342 345 .162 | .389 .2399 .1331 .0641 1027 1386 1542 2789 2789 3145 3228 3228 |
| | M = 1.0 | | 206 | 103 | .346 | .047 | .056 | 181 |
| UPPER SURFACE | 1.00 2.50 5.00 10.00 20.00 20.00 30.00 40.00 50.00 60.00 60.00 60.00 60.00 60.00 60.00 60.00 60.00 | 0337 - 0849 - 11578 - 22544 - 22544 - 22544 - 2266 - 22866 - 23831 - 33477 - 33477 - 33547 | . 20 6 | . 103 959 948 866 797 412 352 353 385 486 427 467 467 467 489 474 489 501 472 332 338 488 | . 346 - 1.023 - 1.013 940 879 819 748 548 445 445 463 520 550 5418 578 | . 047 - 1.049 - 1.053 974 9418 8215 673 545 517 535 559 587 613 639 639 639 639 643 639 643 639 643 643 643 643 643 644 644 644 654 | . 056 - 1.062 - 1.005 - 1.011 975 914 869 815 815 759 665 659 659 657 444 598 574 4448 | - 1.078 - 1.078 - 1.970 - 9792 - 9864 - 8879 - 8868 - 7257 - 7757 - 7757 - 7767 - 6886 - 6671 - 6652 - 6552 - 574 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 25.00 25.00 45.00 45.00 45.00 65.00 65.00 80.00 90.00 90.00 | 4 382 4 4343 3 365 3 265 2 2 2 2 4 1 1 7 5 7 1 1 1 8 2 0 1 1 1 2 2 2 4 1 1 5 7 8 8 9 1 2 2 2 4 1 2 2 2 4 1 3 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 | . 498 . 498 . 352 . 352 . 2705 . 1206 . 1770 . 1812 . 2631 . 405 . 308 . 907 . 907 . 907 . 907 . 907 . 265 | . 482 . 351 . 320 . 284 . 225 . 1925 . 1925 . 1925 . 1927 . 218 . 260 . 327 . 336,0 . 359 - 498 - 498 - 498 - 498 - 498 - 483 . 367 | . 304 . 3267 . 2787 . 22787 . 2132 . 1192 . 1179 . 1178 . 1298 . 2996 . 348 4451 4460 . 353 | . 413 .3497 .2625 .193 .173 .1618 .1699 .28908 .323 417 445 445 447 | .3860 .342 .2291 .258 .2001 .181 .157 .136 .1227 .1217 .150 .1982 .200 .200 .360 350 350 350 350 363 | - 13899289928992899289928992899289928992899 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| ĺ | DEDCEME | | | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|---|---|---|---|---|---|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | N. = 1 | $.00 \cdot \alpha = 7.9^{\circ}$ | | • | | | | |
| UPPER SURFACE | 1.00 1.25 2.55 5.50 10.0 | 344 369 389 511 354 377 391 | 374 395 417 430 430 775 390 361 305 | 0 3 0 - 1 . 0 4 8 - 1 . 0 4 8 99 4 89 7 7 4 6 9 4 3 4 4 3 8 4 5 1 4 7 3 5 1 7 5 2 7 5 3 1 3 6 4 8 3 6 8 3 7 8 3 8 | . 201 - 1.107 - 1.095 - 1.095 - 1.095 - 1.0959838838875675564560563 | 583 533 639 617 574 503 | 9362 9259 9085 9085 8851 7158 6293 6879 | 751 719 723 |
| LOWER SURFACE | 1.25 2.50 7.50 10.00 15.00 20.00 25.00 40.00 45.00 55.00 60.00 75.00 80.00 95.00 GAP | 973 402 329 239 179 | .430 .3899 .2967 .2255 .235 .257 | .544 .418 .418 .364 .307 .277 .257 .257 .2312 .247 .344 .381 .499 .4997 .4997 .4969 .388 | . 532 . 460 . 389 . 3144 . 336 . 2261 . 2211 . 2211 . 2261 . 3165 . 3870 - 421 - 452 - 452 . 375 | . 23 9 . 21 4 . 20 2 . 19 3 . 19 8 . 24 9 . 30 5 . 34 8 . 34 9 | . 422 . 399 . 346 . 250 . 225 . 199 . 175 | .430 .340 .283 .231 |
| | M = 1 | | ·- · | | | 1 | , | |
| UPPER SURFACE | 1.25 2.50 5.00 7.50 10.00 25.00 35.00 45.00 45.00 66.00 67.00 75.00 85.00 85.00 90.00 95.00 | 024 038 048 072 077 097 154 184 202 188 206 207 194 | 07751111155518322166236226311369186218162 | 518 - 046 - 028 - 0372 - 068 - 138 - 1574 - 128 - 2562 - 2883 - 2863 - 1683 - 1572 - 1683 - 1572 - 1693 - 172 - 189 | 189 211 236 | 125 141 155 167 220 235 279 316 342 379 395 342 235 229 346 226 226 | 321 286 284 268 269 | 417 4034 262 250 242 240 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 15.000 20.000 30.000 40.000 45.000 55.000 65.000 75.000 80.000 90.000 | 184 -119 -0815 -0322 -0122 -0437 -0627 -0627 -087 -1088 -2445 -45930 -52930 -435 | 006 0005 0232 0444 0844 1024 1024 1057 3650 3339 78421 6664 554 | 059 041 050 054 074 087 .099 123 .222 .296 .331 .347 489 481 .365 | . 022 - 0014 - 0037 - 037 - 047 - 055 - 069 - 085 - 111 - 154 - 293 - 352 - 424 - 424 - 434 - 365 | .089 .062 .042 .043 .049 .086 .1154 .2206 .362 .362 .389 394 3947 | .018 .084 .071 .039 .045 .059 .059 .084 .1053 .233 | 1752 10958 10958 10958 10022 11078 10788 1 |

TABLE IV

| Ī | 22222 | | - | PRESSUR | E COEFFICIENT, | P, AT: | | |
|---------------|---|---|--|--|--|--|---|--|
| | PERCENT CHORD | 0.135b/2 | 0.25b/2 | .0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95b/2 |
| | M = 1.00 .00 1.25 2.50 5.00 7.50 | 069 .098 .009 064 | .392 603 568 276 237 | .301 762 732 576 271 261 | .546 838 834 745 634 409 | .253 877 884 791 752 | .245 883 855 842 796 771 | .035 906 913 809 822 796 |
| UPPER SURFACE | 10.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.000 20.0000 20.000 20.000 20.00000 20.00 | 1099 11591 11750 11750 11803 11918 1 | | 2673 2279 2888 3142 35678 367862 376862 3821 26633 26633 26633 | 310 327 341 349 359 404 432 445 455 454 489 489 334 334 | 570 357 392 410 425 477 508 493 509 4612 388 | 733 6667 4293 4484 5149 5588 5863 5863 54513 403 | - 6961 - 7003 - 7003 - 5070 - 4449 - 4486 - 55328 - 5528 - 5528 - 5555 - 5548 - 5555 - 5548 - 5555 - 5548 - 5555 - |
| LOWER SURFACE | 1.25 2.500 7.500 10.00 15.00 25.00 30.00 40.00 40.00 55.00 60.00 65.00 | 3 4 4 3 5 1 7 2 2 5 6 2 2 5 6 2 1 7 4 1 1 1 1 3 1 1 0 9 5 1 1 6 1 3 2 2 0 5 7 4 7 | 376 316 247 208 1180 1147 1122 1110 1143 1192 245 420 345 420 | .402 .375 .275 .222 .227 .199 .189 .187 .198 .274 .343 .371 .369 | . 482 . 358 . 274 . 241 . 241 . 213 . 194 . 180 . 177 . 186 . 258 . 3282 . 364 | .486 .385 .322 .273 .244 .168 .163 .165 .181 .207 .253 .317 | .510 .351 .27321 .22444 .1913 .157 .1345 .1231 .2240 .2240 | . 371 . 2857 . 1809 . 1012 . 00576 |
| | 75.00 80.00 85.00 90.00 95.00 GAP | $\begin{array}{cccccccccccccccccccccccccccccccccccc$ | 745 828 756 646 464 .266 | 449 457 456 455 450 .380 | 389 398 408 417 421 .371 | 373 381 389 395 402 .350 | 310 308 307 313 .203 | 286 291 287 284 |
| UPPER SURFACE | 1.25 2.50 5.00 10.00 25.00 30.00 30.00 40.00 45.00 55.00 60.00 60.00 70.00 80.00 95.00 | 084 .052 052 131 147 191 214 220 230 252 275 278 294 313 304 | 247 | 446 370 299 273 285 | 3889469328738745644143784034434434453793493472135933554 | 496 527 526 381 579 559 | 541 561 536 | 719 654 637 620 620 |
| LOWER SURFACE | 40.00 45.00 50.00 55.00 | .4399 .3728 .32737 .22091 .11772 .11815 .235496 .92031 .94032 .94032 .94032 .94032 .94032 | .503 .362 .3162 .3163 .245 .247 .1995 .2987 .2987 .2987 .2987 .2987 .2987 .2987 .2997 .4099 | 441 447 446 433 | . 566 . 4400 . 3116 . 315 . 273 . 2251 . 232 . 219 . 219 . 249 . 2897 . 4001 . 385 . 3768 . 3768 . 3768 . 3788 . 3788 . 3788 | .449 .385 .3355 .300 .2621 .2007 .207 .2088 .3277 .3577 .3577 .3775 .399 | . 410 .3727 .2956 .2186 .21975 .11686 .1775 .16837 .2355 .2323 .2323 .2323 .2323 .2323 | . 441379 . 24379 . 215486 . 00579 . 00 |

TABLE IV

WING WITH LOWER SURFACE SPOILER (WITH GAP)

| | PRESSURÉ COEFFICIENT, P, AT: | | | | | | | |
|---------------|--|---|--|---|---|--|---|---|
| | CHORD | 0.135b/2 | 0.25b/2 | 0.40b/2 | 0.55b/2 | 0.70b/2 | 0.85b/2 | 0.95ъ/2 |
| | M = 1.0 | | | | | | | |
| UPPER SURFACE | 1.250 2.500 10.000 | 100 010288866 | 097 955 955 9677 9455 | . 005 - 991 - 980 - 940 - 940 - 868 - 790 - 618 - 454 - 4454 - 4454 - 4454 - 459 - 459 - 459 - 459 - 468 - 474 - 488 - 4701 - 316 - 298 - 308 | .240 -1.045 -1.032 -981 -9939 -9038 -798 -798 -569 -516 -520 -517 -516 -527 -516 -527 -516 -527 -517 -515 | 0762 - 1.0648 9969 99530 99530 8880 8799 7761 6698 6597 55316 5537 55866 | - 053 - 1.000 - 1.020 - 1.994 9974 9974 9976 8651 8651 8651 6641 6657 5949 5549 | 289 - 1.077 - 1.07290338991899585928108298367457497528733 |
| LOWER SURFACE | 1.25 2.500 7.500 10.000 25.000 25.000 35.000 40.000 50.000 70.000 80.000 95.000 95.000 | 3238 455418231557287287287287287287287287287287287287287 | 4392626427900872 935147397666839572 447397666839548 9466203 447633 447633 | .572 .519 .450 .419 .345 .345 .3294 .269 .269 .285 .317 .415 .440 .442 .443 .442 | .564 4421 .3872 .3872 .3872 .3872 .3899 .28758 .2848 .2487 .2695 .2949 .4103 .4103 .4103 .4105 .407 | .57 20 30 20 20 20 20 20 20 20 20 20 20 20 20 20 | .577 .448 .448 .377 .345 .2859 .2339 .2193 .183 .185 .199 .239 .275 .239 .275 .239 .275 .239 .241 .241 .241 .241 .241 .241 .241 .241 | .487 .459 .374 .3166 .197 .025 .0017 0465 096 150 289 314 289 317 330 336 355 |

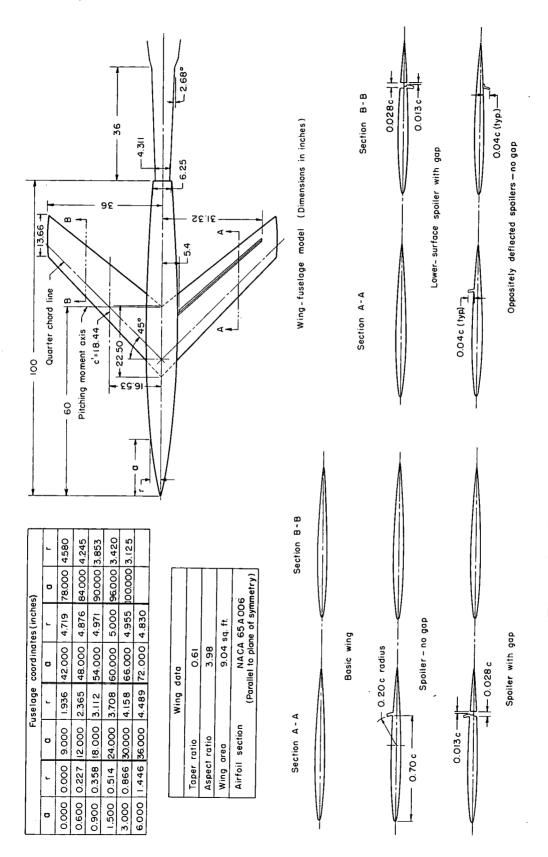


Figure-1.- Geometry of test configurations.

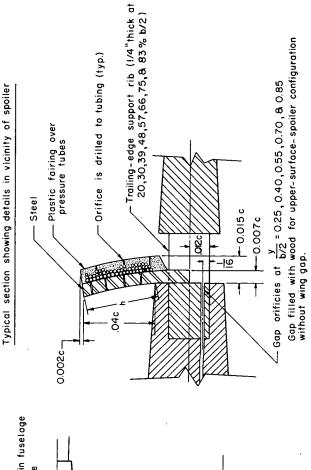
 $\frac{y}{b/2} = 0.85$ 405 643 835 178 463 80 0.159 Spoiler-orifice locations, z/h $\frac{y}{b/2} = 0.70$ 880 687 810 .435 160 413 0.188 $\frac{y}{b/2} = 0.55$ 808 .668 866 030 .422 428 0.177 $\frac{y}{b/2} = 0.40$ 448 720 868 0.207 035 445 841 $\frac{y}{b/2} = 0.25$ 455 704 090 448 865 0.201 716 Front Front Front Front Back Back Back

station are given with the pressure coefficient

data in Tables I - IX

Chordwise locations of wing orificies at each

Typical section showing details in vicinity of spoiler



Orifices are located in fuselage 0.1" from wing surface .95b/2 .85b/2 .70b/2 .55b/2 40 b/2 25b/2 0.135 b/2 Wing orifice stations 10.70c

Figure 2.- Pressure-orifice locations and details of spoiler and wing gap.

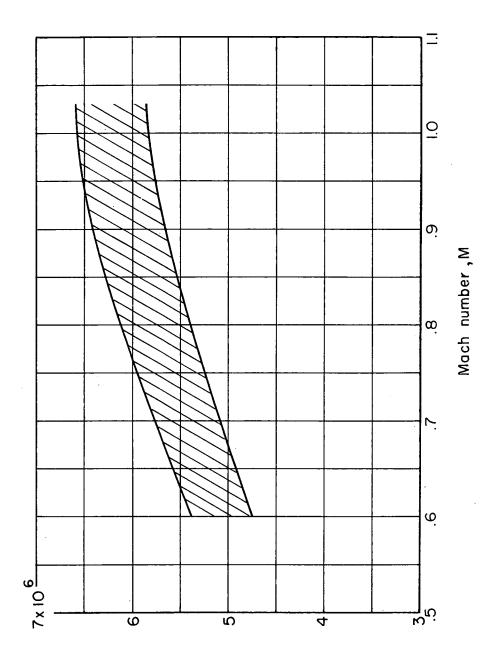


Figure 3.- Variation of Reynolds number based on mean aerodynamic chord with Mach number.

Reynolds number

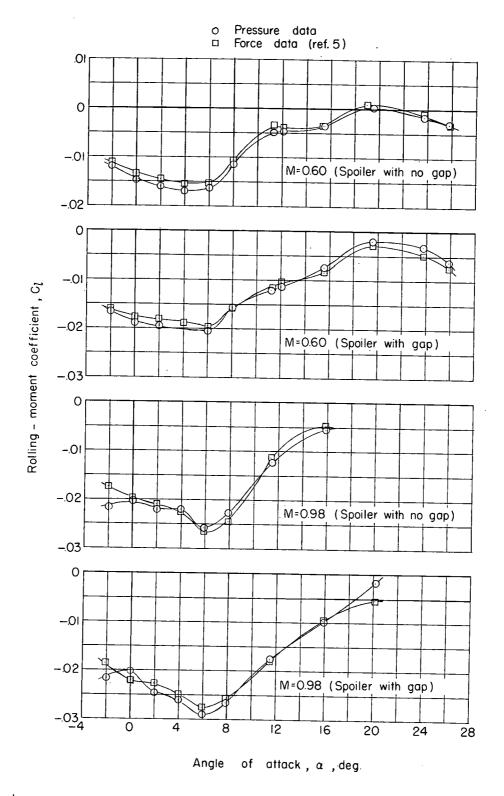


Figure 4.- Comparison of rolling-moment characteristics as obtained from pressure data and from force data for the model with upper-surface spoilers.

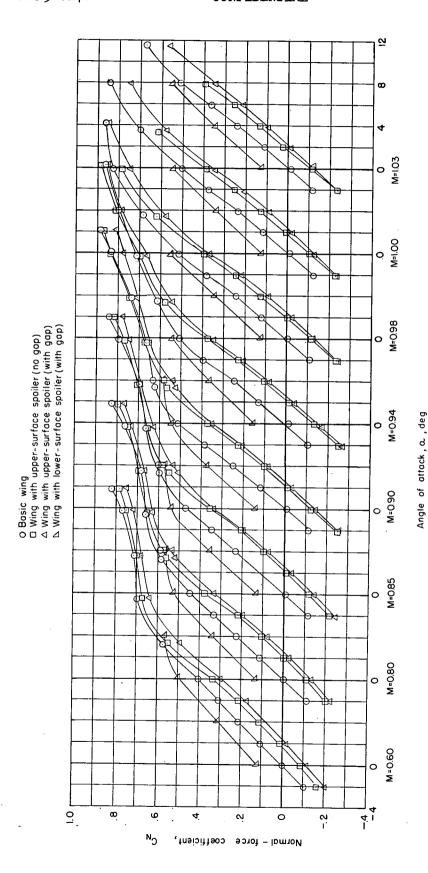
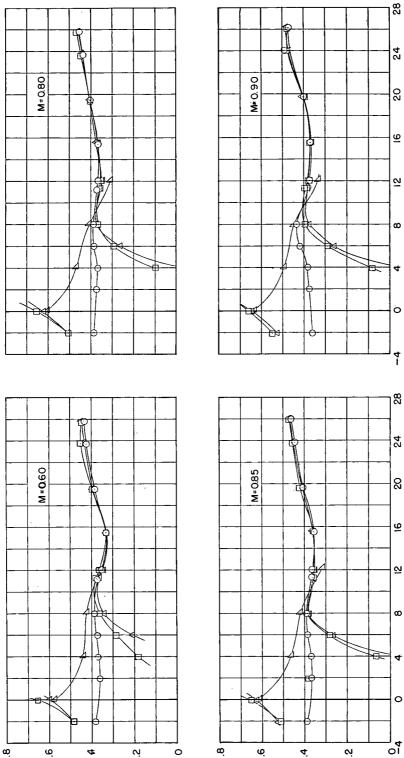


Figure 5.- Wing normal-force characteristics for the basic wing and various spoiler configurations. M = 0.60 to M = 1.03.



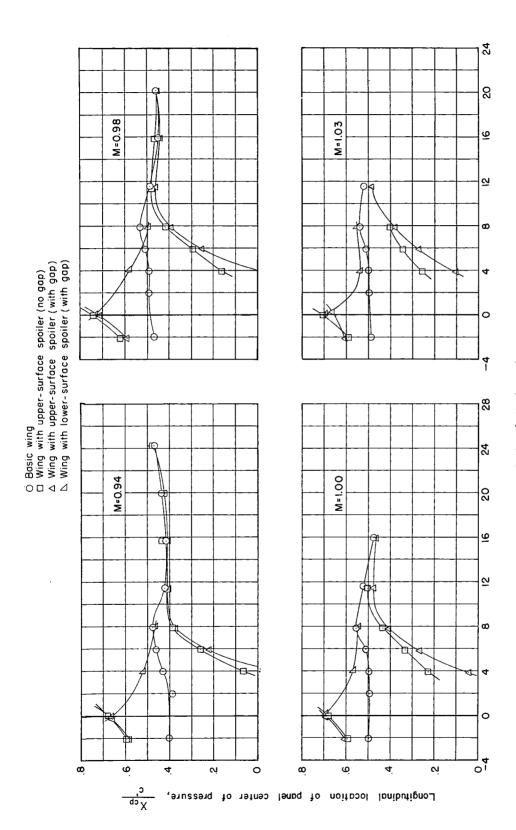


Angle of attack, a., deg

(a) M = 0.60, 0.80, 0.85, 0.90.

Figure 6.- Longitudinal position of wing center of pressure for the basic wing and various spoiler configurations.

Longitudinal location of panel center of pressure,

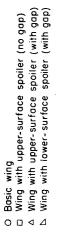


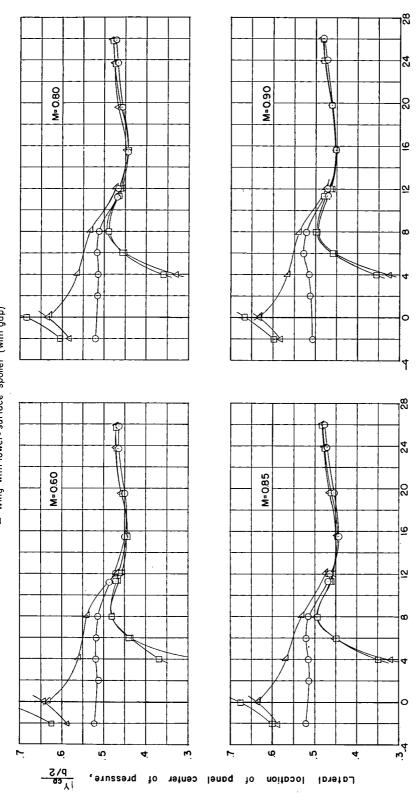
Angle of attack,a., deg

(b) M = 0.94, 0.98, 1.00, 1.05.

Figure 6.- Concluded.

CONFIDENTIAL



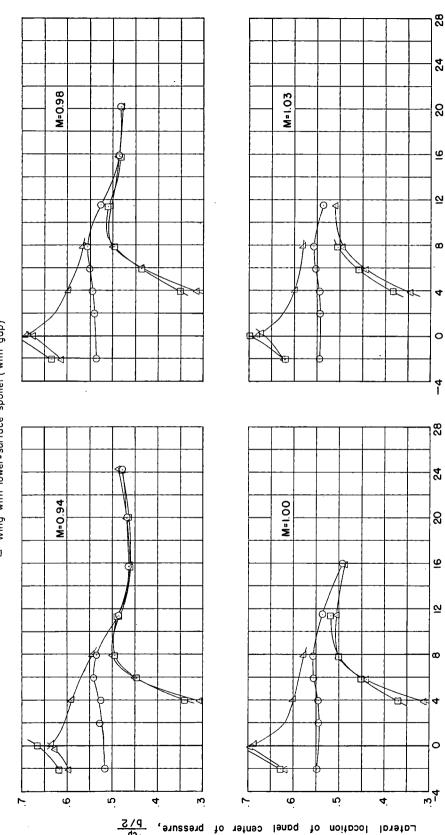


Angle of attack,a., deg

(a) M = 0.60, 0.80, 0.85, 0.90.

Figure 7.- Lateral position of wing center of pressure for the basic and various spoiler configurations.

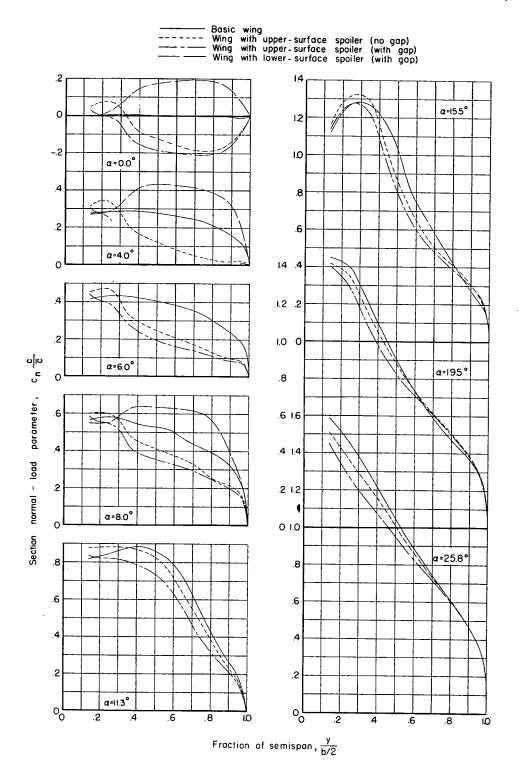
O Basic wing
□ Wing with upper-surface spoiler (no gap)
△ Wing with upper-surface spoiler (with gap)
△ Wing with lower-surface spoiler (with gap)



Angle of attack, α , deg

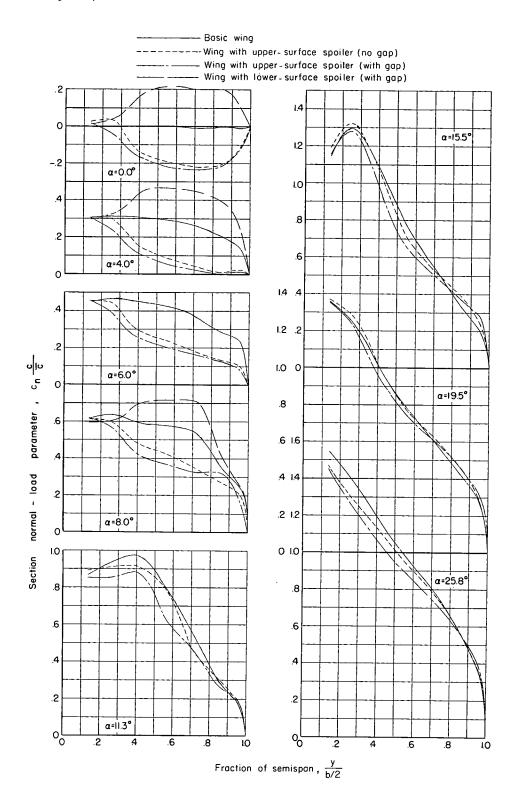
(b) M = 0.94, 0.98, 1.00, 1.05.

Figure 7.- Concluded.



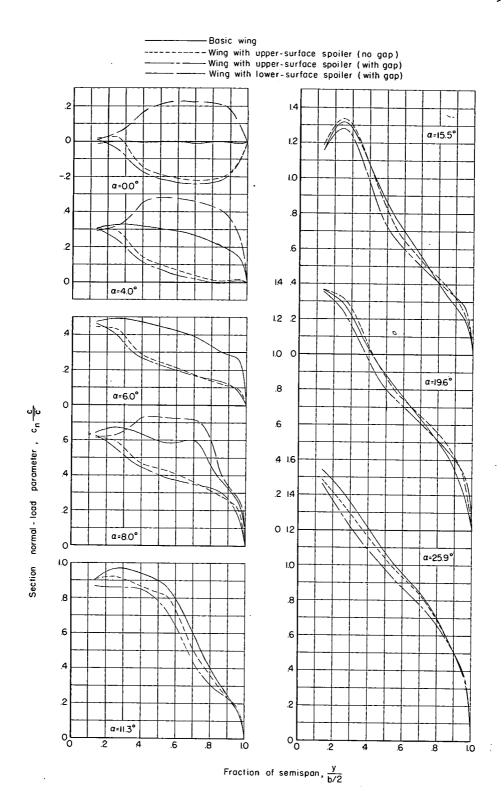
(a) M = 0.60.

Figure 8.- Wing semispan load distributions for the basic wing and various spoiler configurations.



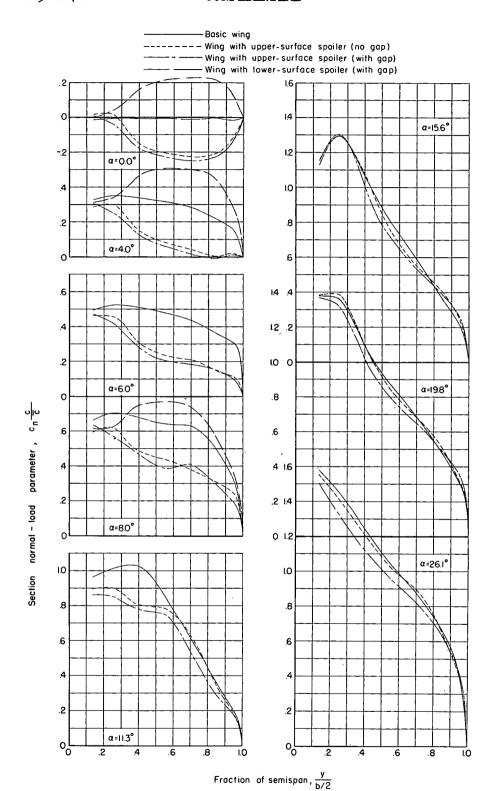
(b) M = 0.80.

Figure 8.- Continued.



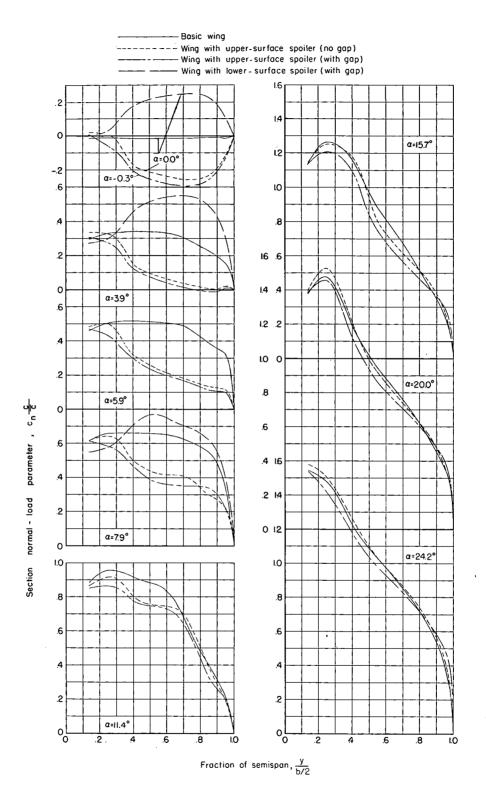
(c) $M_{.}=0.85$.

Figure 8.- Continued.



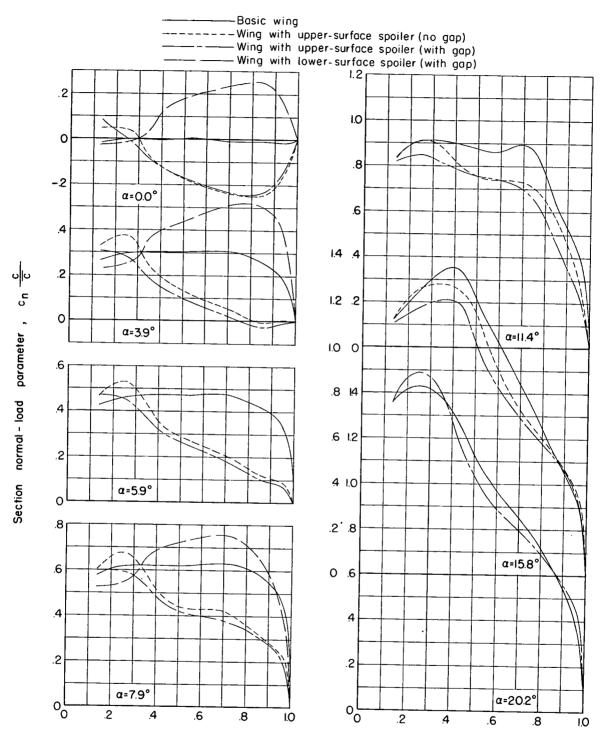
(d) M = 0.90.

Figure 8.- Continued.



(e) M = 0.94.

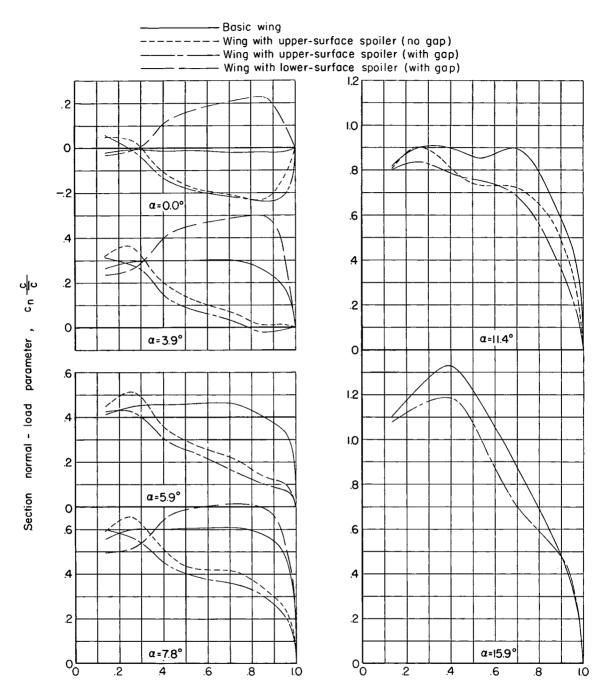
Figure 8.- Continued.



Fraction of semispan, $\frac{y}{b/2}$

(f) M = 0.98.

Figure 8.- Continued.

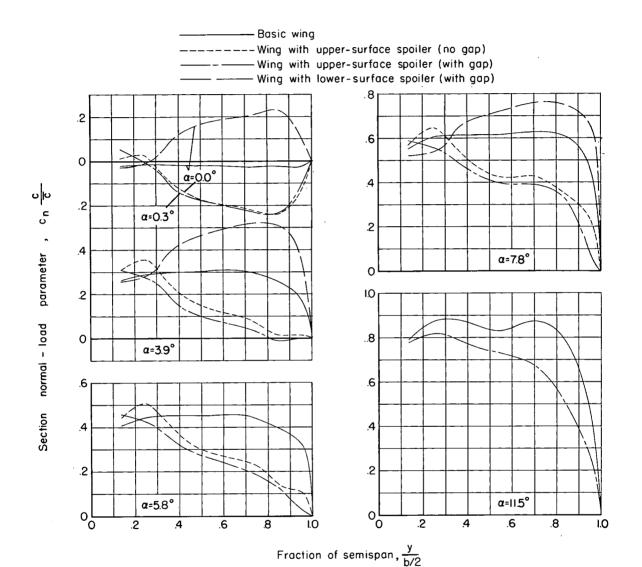


Fraction of semispan, $\frac{y}{b/2}$

(g) M=1.00.

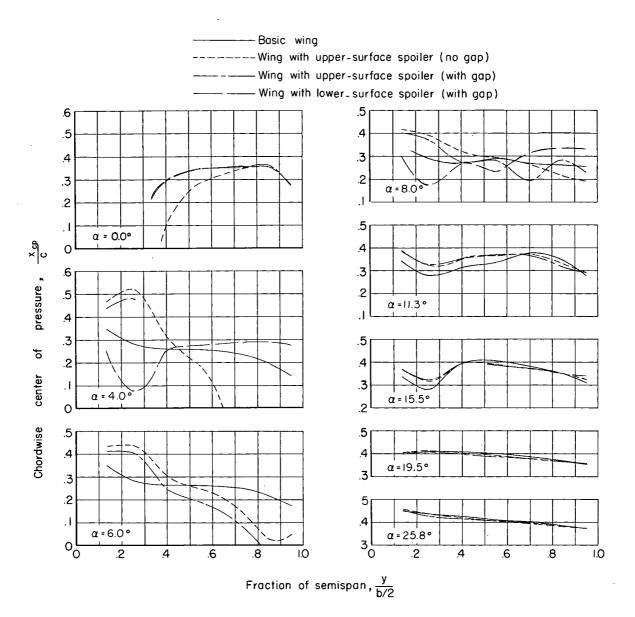
(g) M = 1.00.

Figure 8.- Continued.



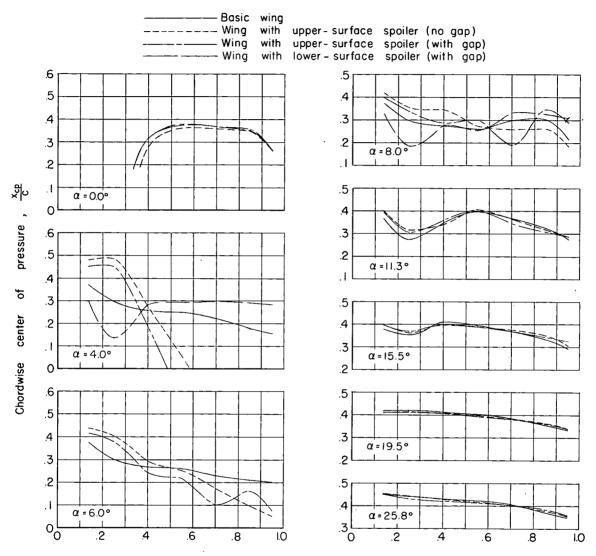
(h) M = 1.03.

Figure 8.- Concluded.



(a) M = 0.60.

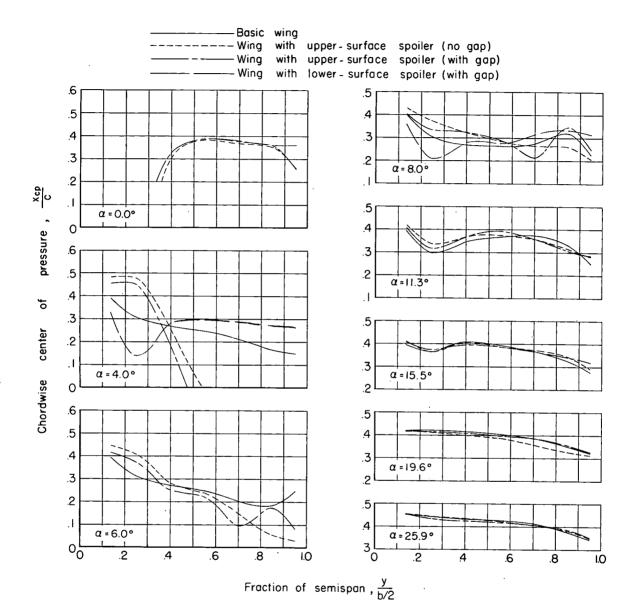
Figure 9.- Wing-section center of pressure across the semispan for the basic wing and various spoiler configurations.



Fraction of semispan, $\frac{y}{b/2}$

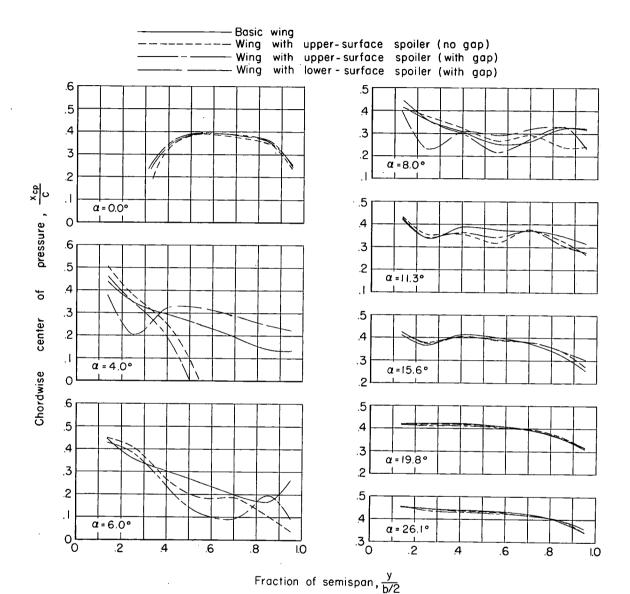
(b)
$$M = 0.80$$
.

Figure 9.- Continued.



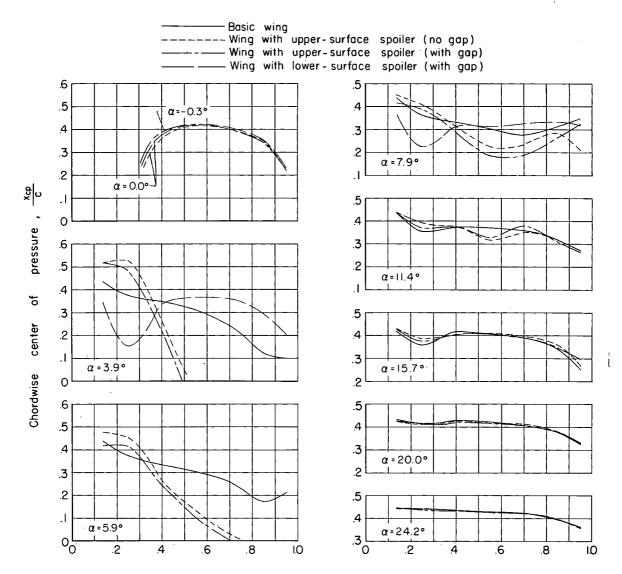
(c) M = 0.85.

Figure 9.- Continued.



(d) M = 0.90.

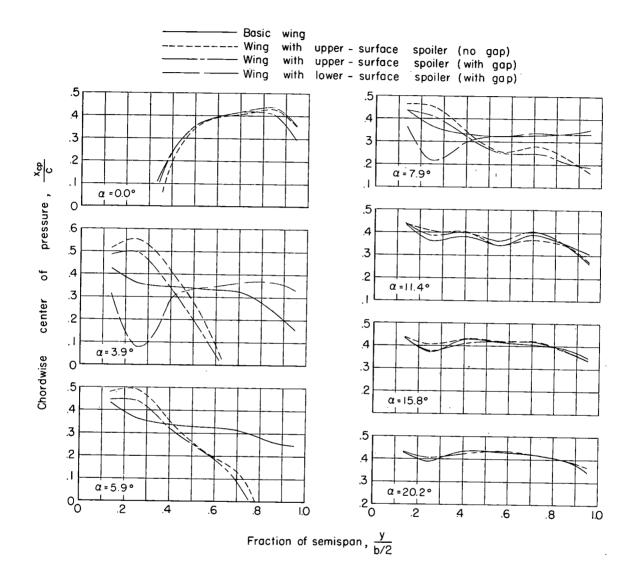
Figure 9.- Continued.



Fraction of semispan, $\frac{y}{b/2}$

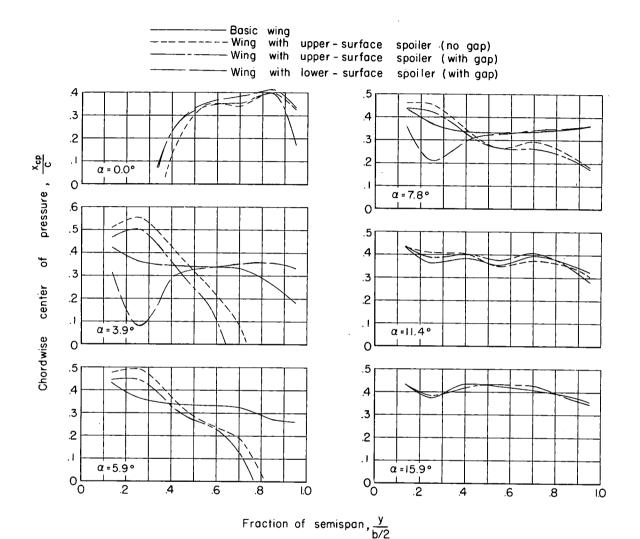
(e) M = 0.94.

Figure 9.- Continued.



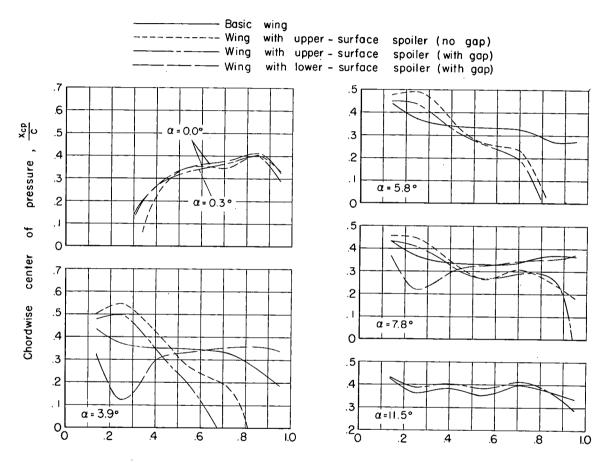
(f) M = 0.98.

Figure 9.- Continued.



(g) M = 1.00.

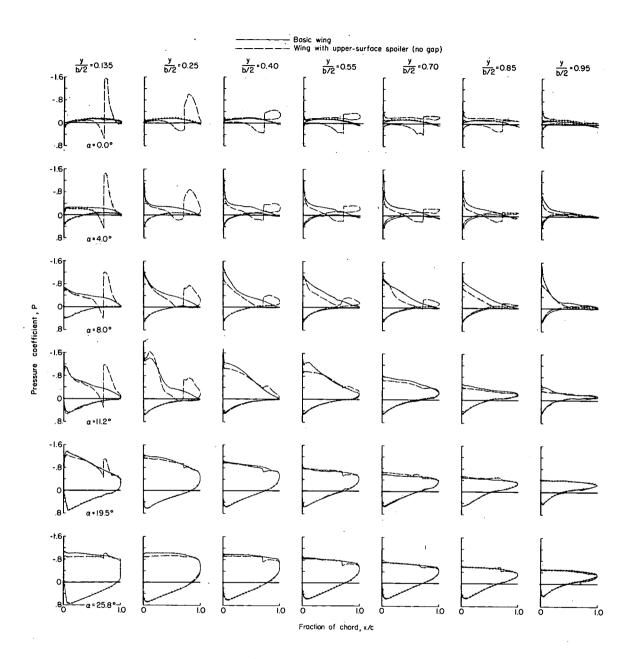
Figure 9.- Continued.



Fraction of semispan, $\frac{y}{b/2}$

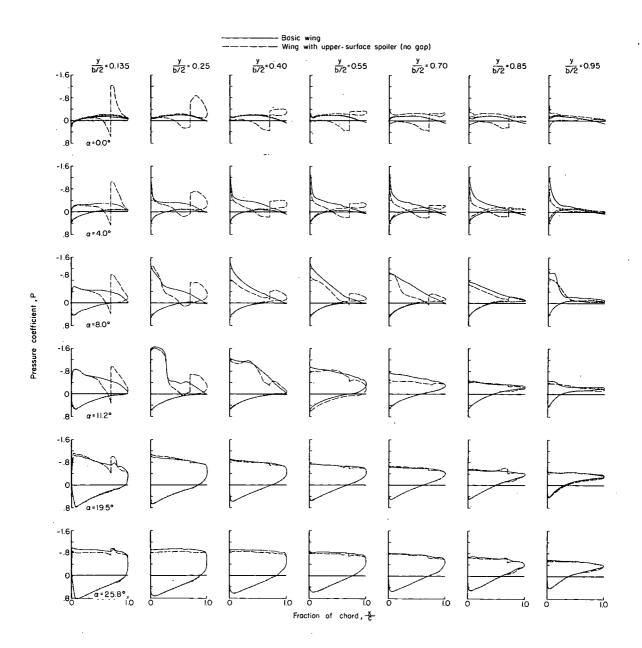
(h)
$$M = 1.03$$
.

Figure 9.- Concluded.



(a) M = 0.60.

Figure 10.- Chordwise pressure distributions on the wing; basic wing compared with the upper-surface spoiler (no gap) configuration.



(b) M = 0.80.

Figure 10.- Continued.

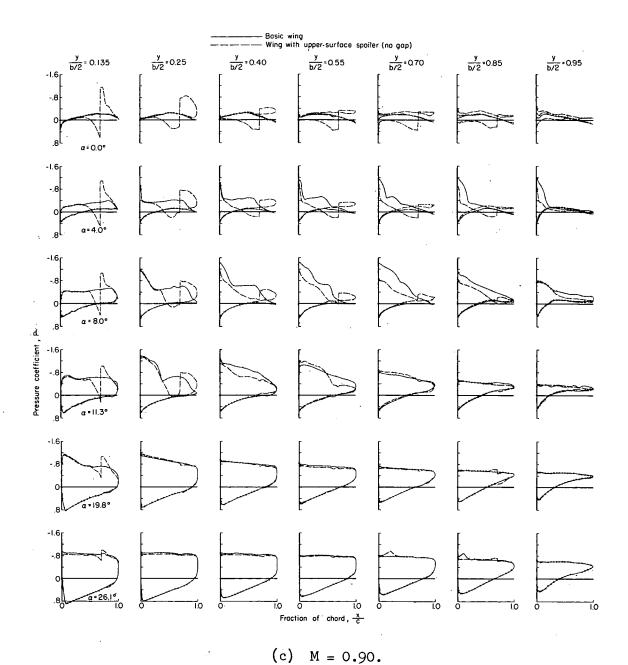
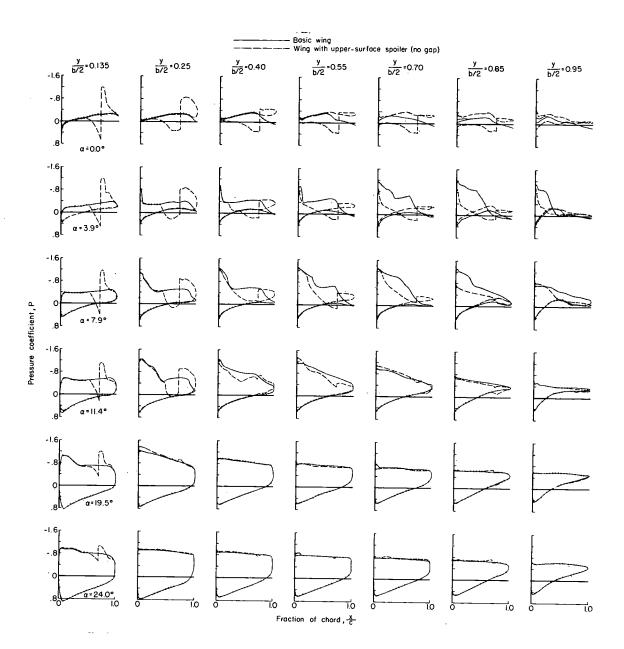
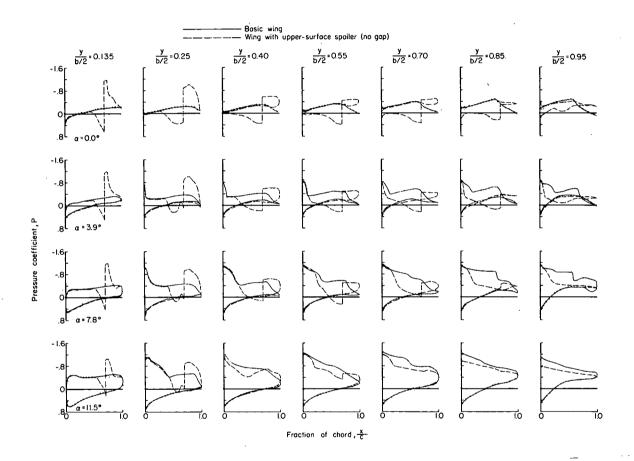


Figure 10.- Continued.



(d) M = 0.94.

Figure 10.- Continued.



(e) M = 1.00.

Figure 10.- Concluded.

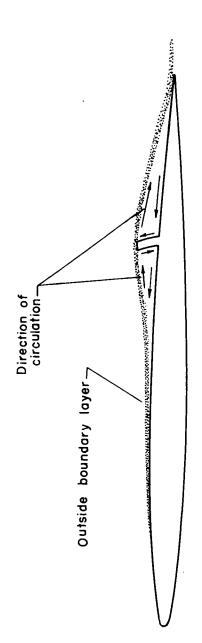


Figure 11.- Concept of flow in the boundary layer at a typical wing-spoiler section for a Mach number of 0.60.

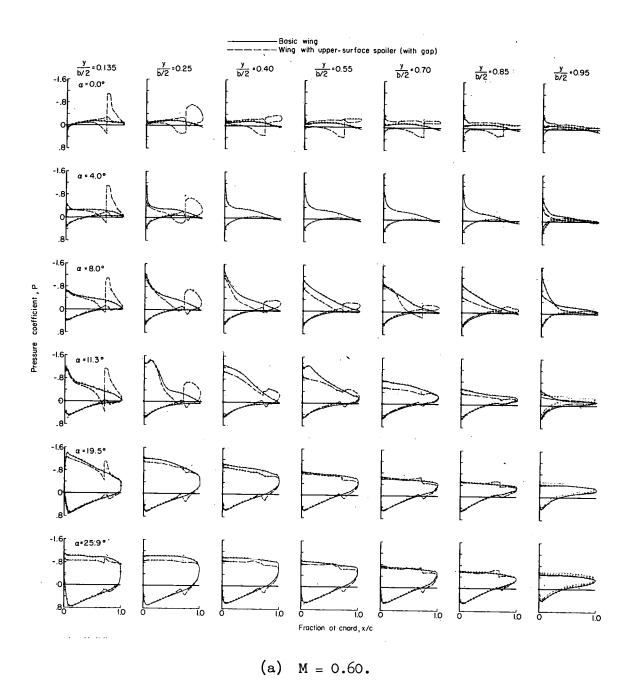
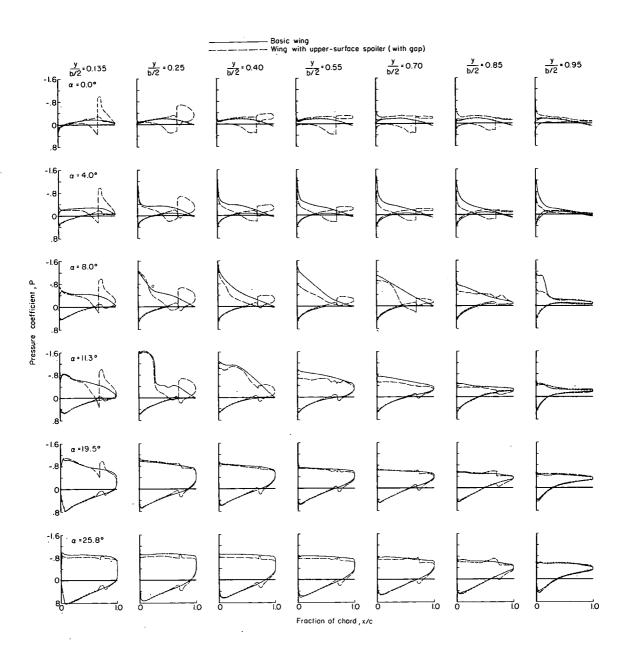
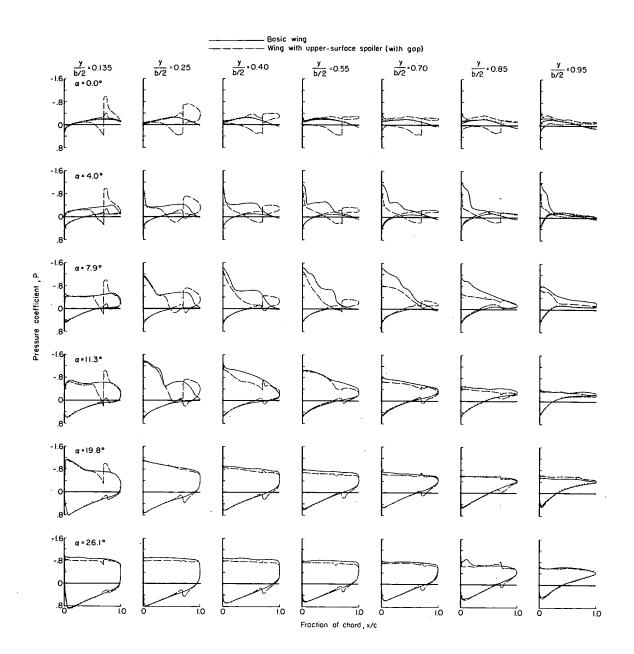


Figure 12.- Chordwise pressure distributions on the wing; basic wing compared with the upper-surface spoiler (with gap) configuration.



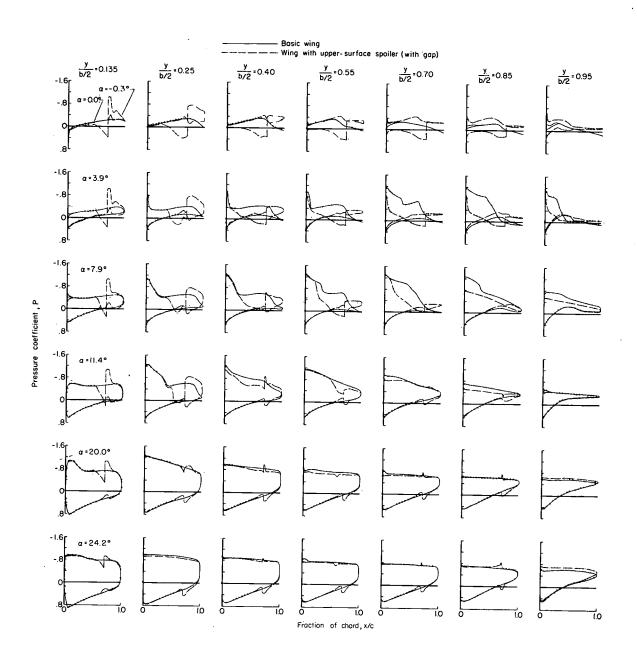
(b) M = 0.80.

Figure 12.- Continued.



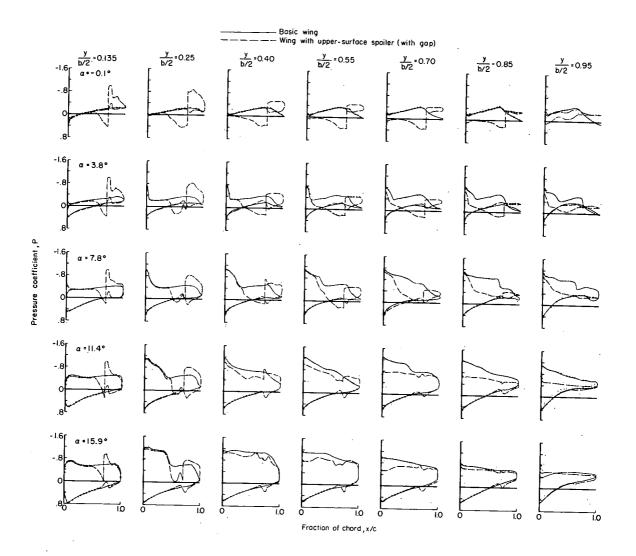
(c) M = 0.90.

Figure 12.- Continued.



(d) M = 0.94.

Figure 12.- Continued.



(e) M = 1.00.

Figure 12.- Concluded.

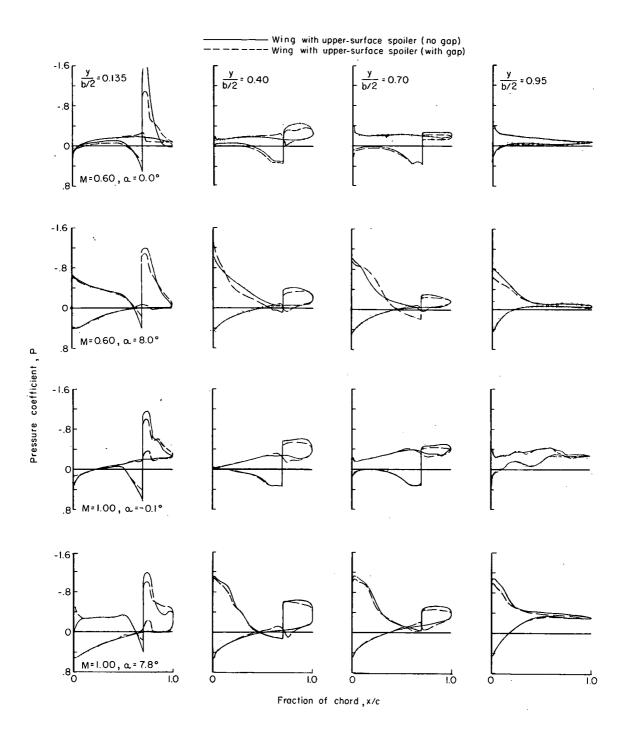


Figure 13.- Effect of wing gap behind an upper-surface spoiler on wing chordwise pressure distributions.

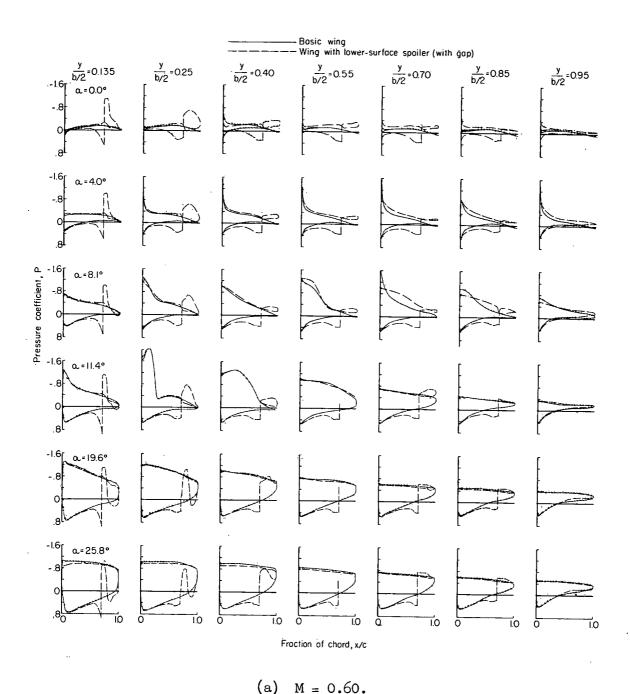
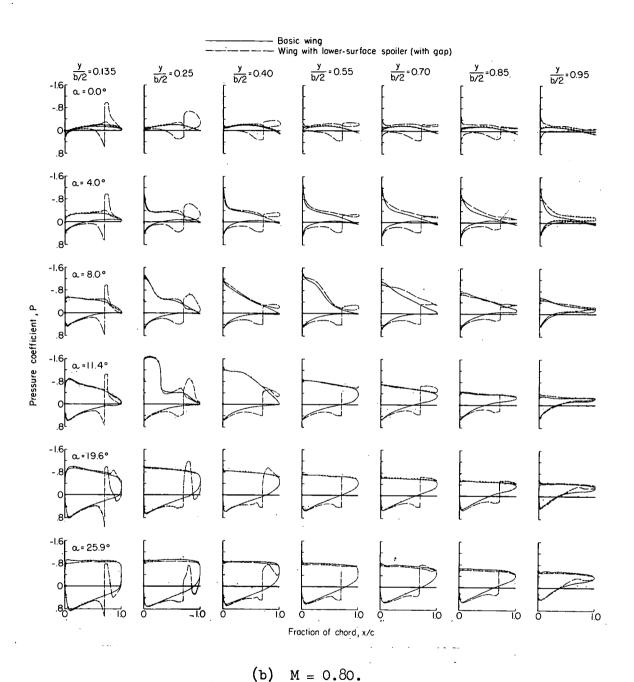
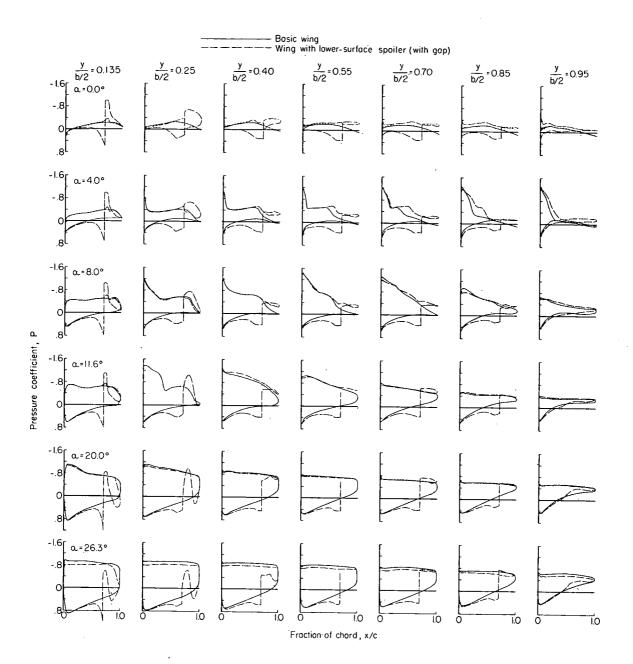


Figure 14.- Chordwise pressure distributions on the wing; basic wing compared with the lower-surface spoiler (with gap) configuration.



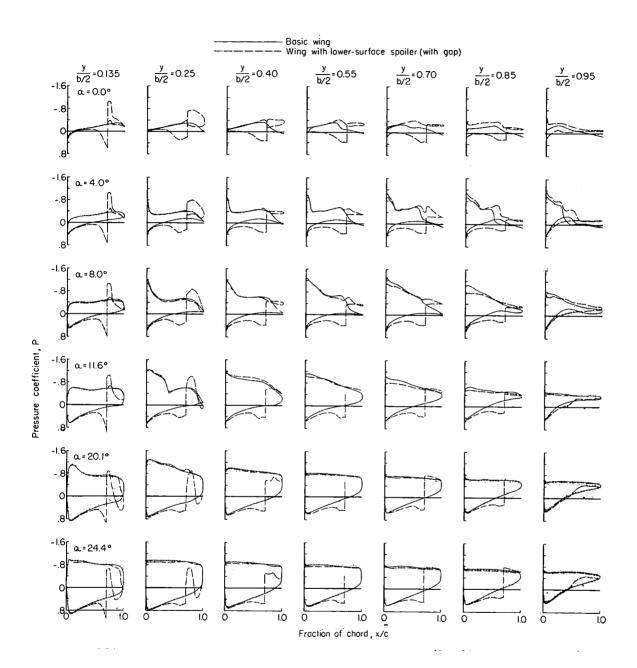
(...,

Figure 14.- Continued.



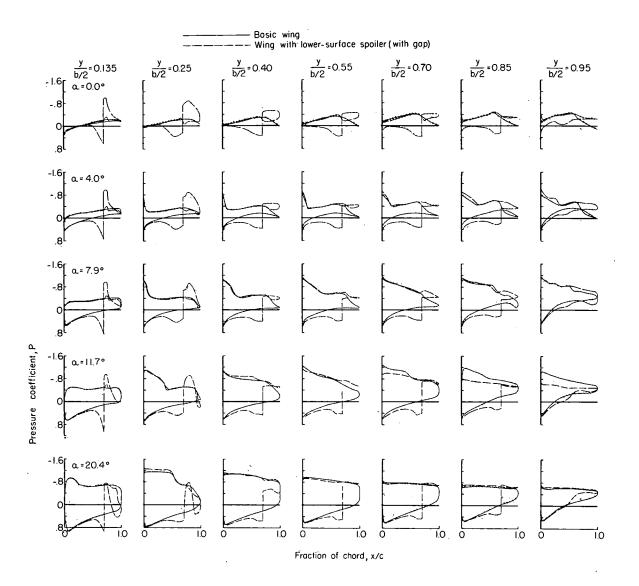
(c) M = 0.90.

Figure 14.- Continued.



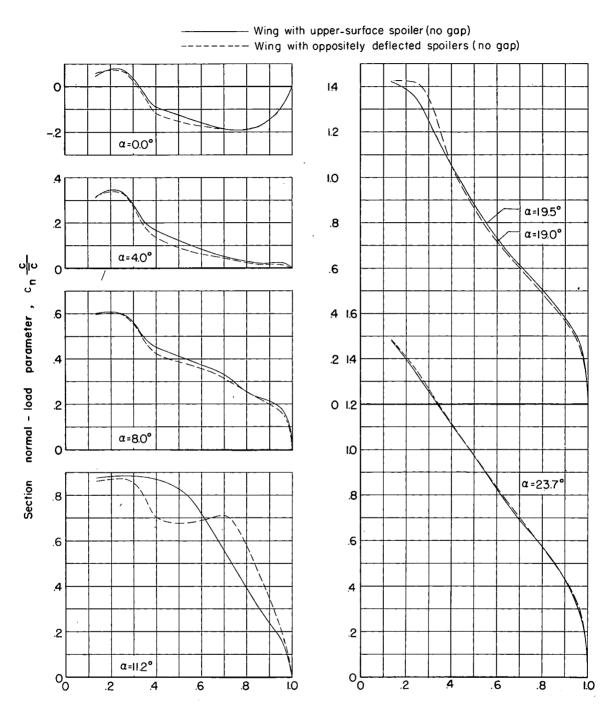
(d) M = 0.94.

Figure 14.- Continued.



(e) M = 1.00.

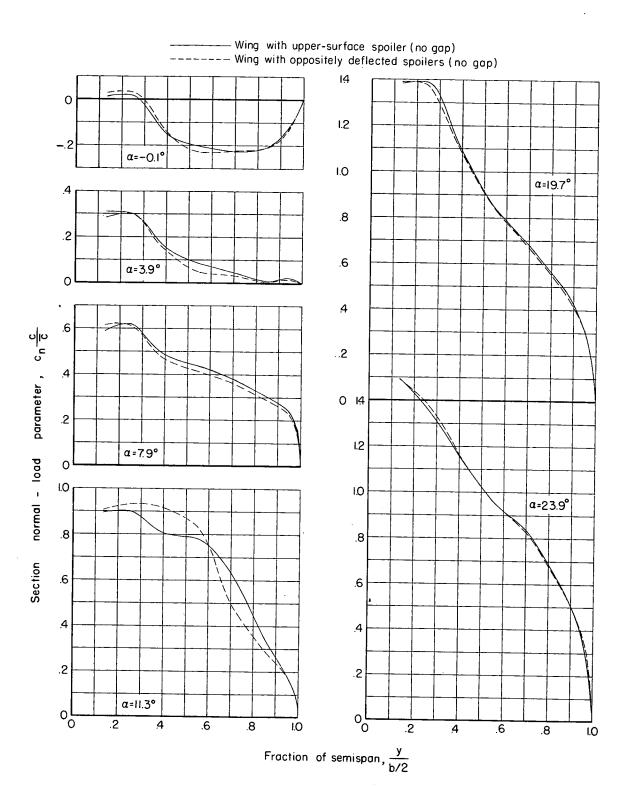
Figure 14.- Concluded.



Fraction of semispan, $\frac{y}{b/2}$

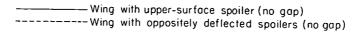
(a) M = 0.60.

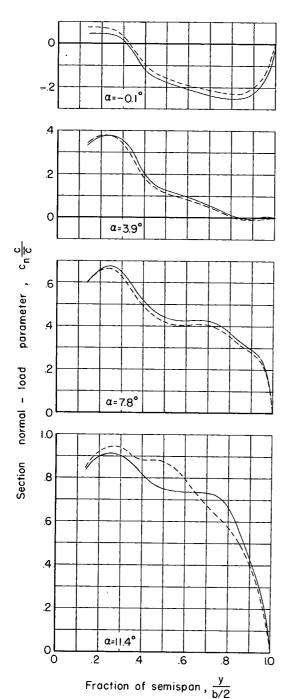
Figure 15.- Effect of a lower-surface spoiler on the opposite wing semispan load distribution for the upper-surface spoiler (no gap) configuration.

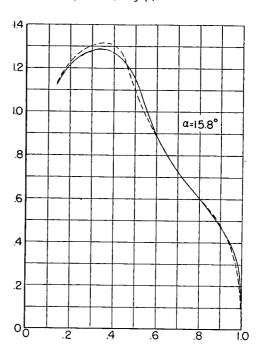


(b) M = 0.90.

Figure 15.- Continued.







Fraction of semispan, $\frac{y}{b/2}$

(c) M = 0.98.

Figure 15.- Concluded.

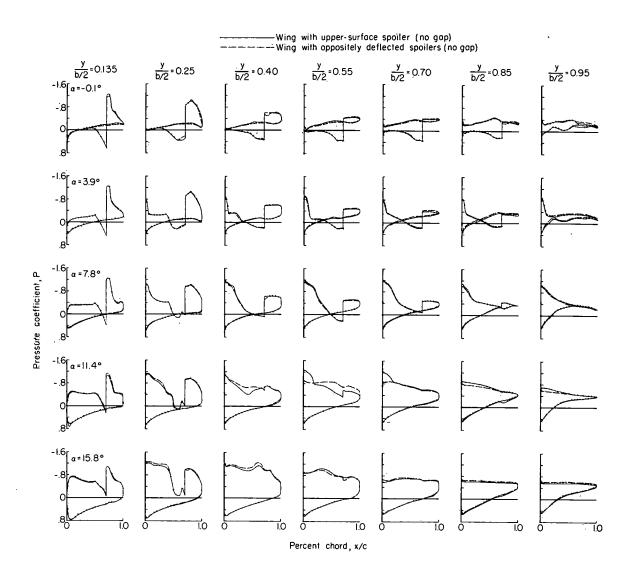
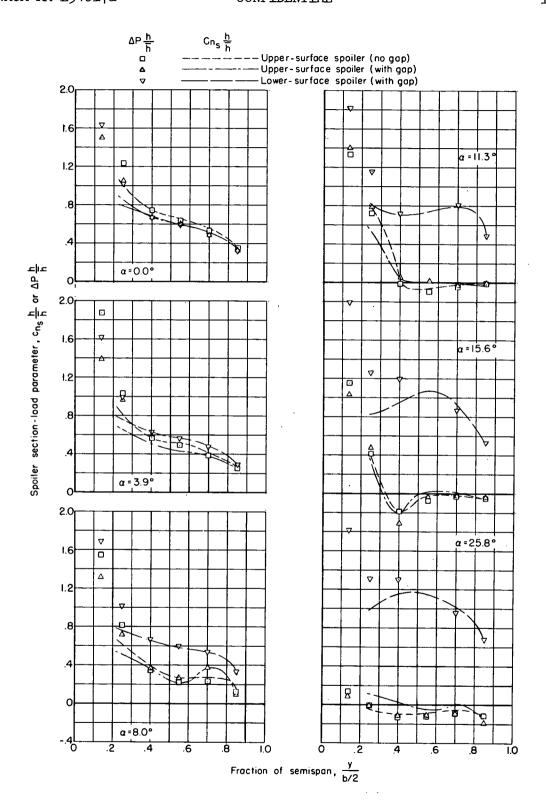
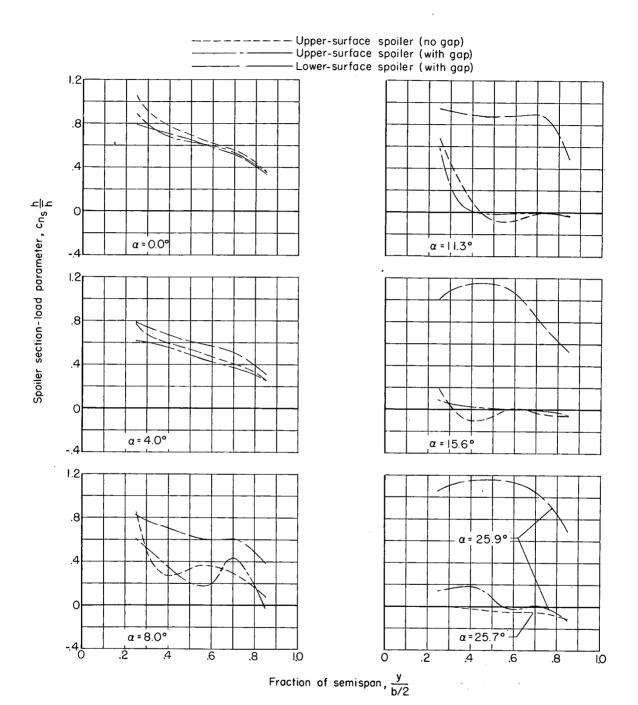


Figure 16.- Effect of a lower-surface spoiler on the opposite wing chordwise pressure distributions for the upper-surface spoiler (no gap) configuration. $\dot{M} = 0.98$.



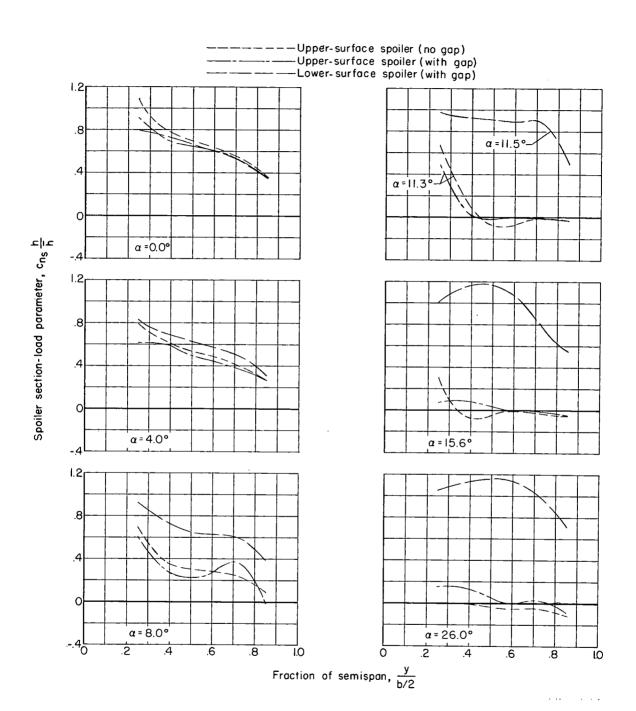
(a) M = 0.60.

Figure 17.- Spoiler load distributions.



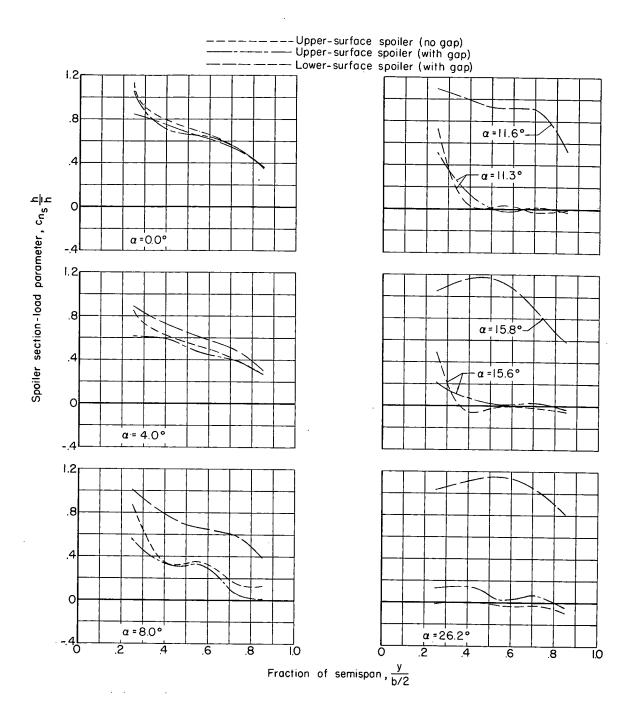
(b) M = 0.80.

Figure 17.- Continued.



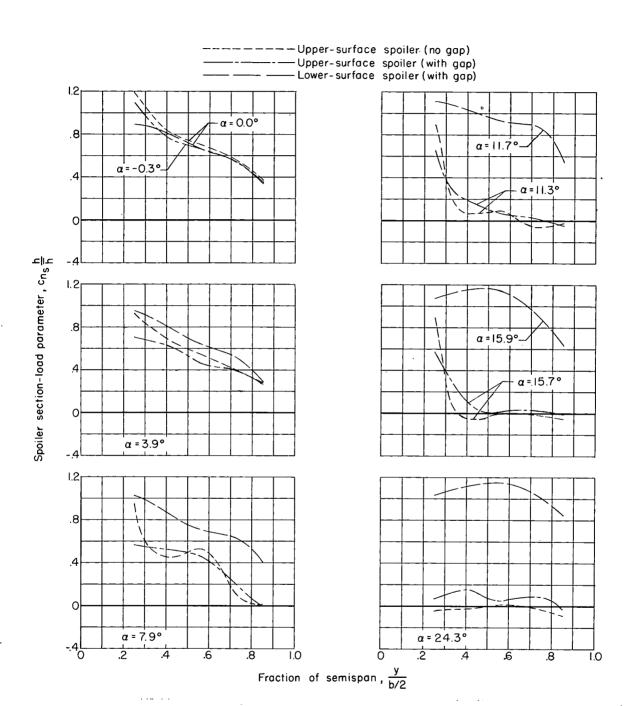
(c)
$$M = 0.85$$
.

Figure 17.- Continued.



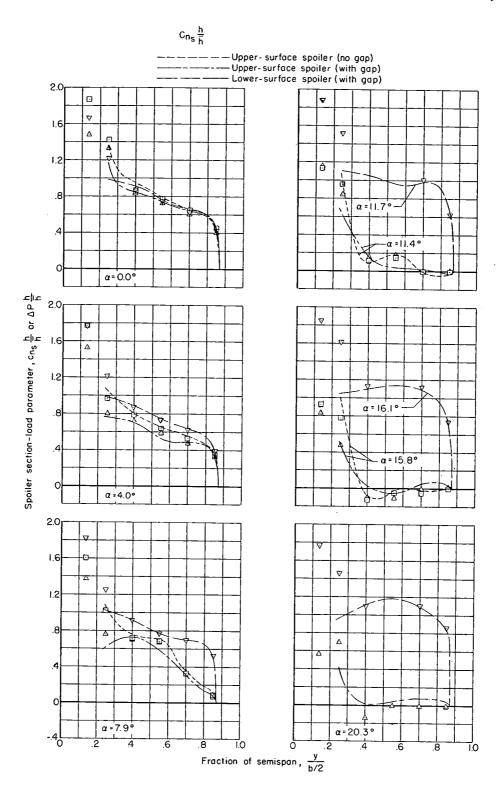
(d) M = 0.90.

Figure 17.- Continued.



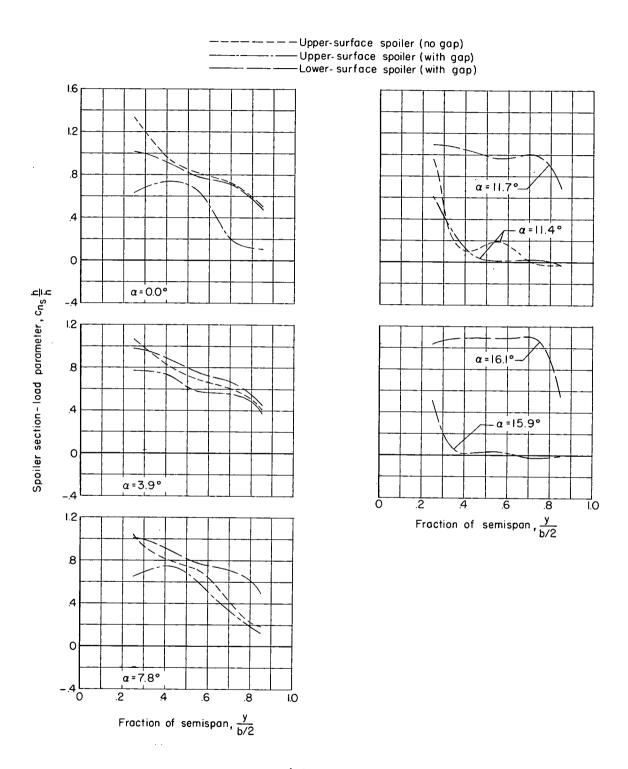
(e) M = 0.94.

Figure 17.- Continued.



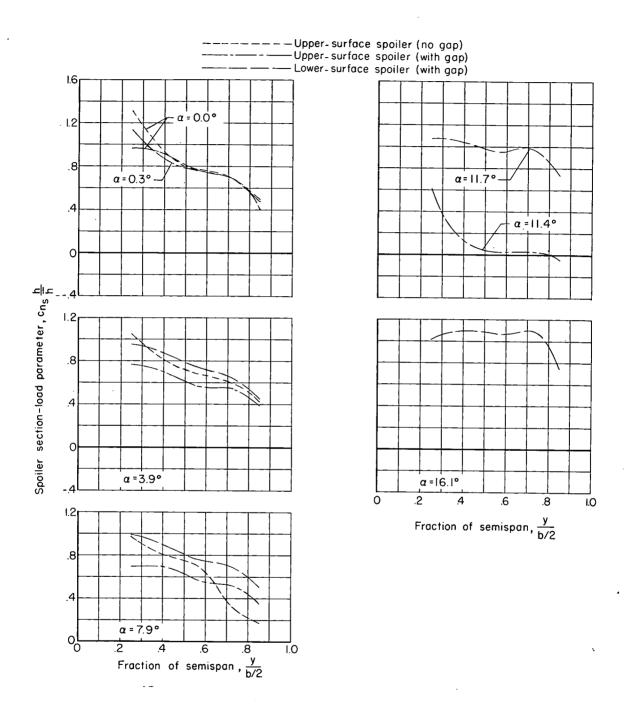
(f) M = 0.98.

Figure 17.- Continued.



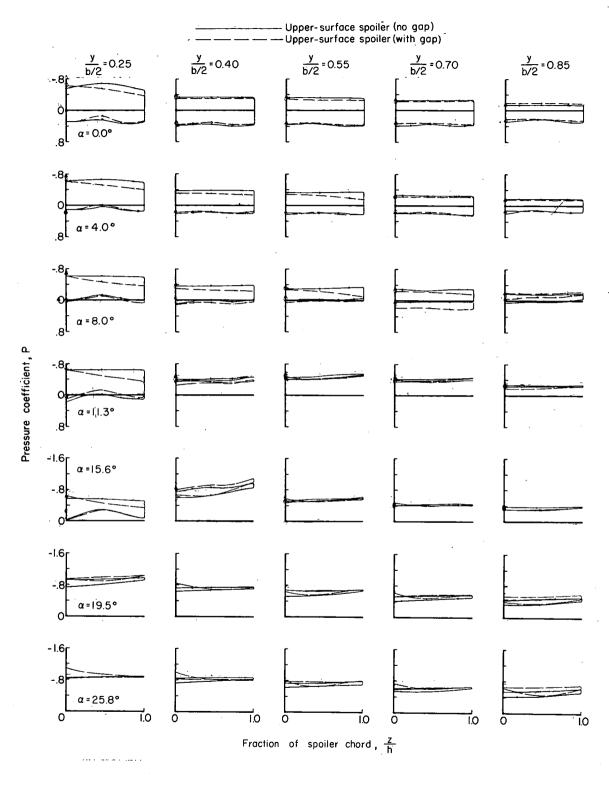
(g) M = 1.00.

Figure 17.- Continued.



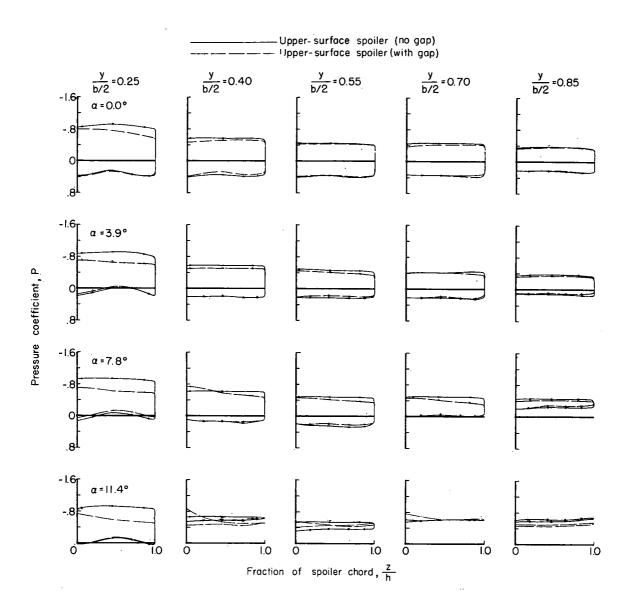
(h) M = 1.03.

Figure 17.- Concluded.



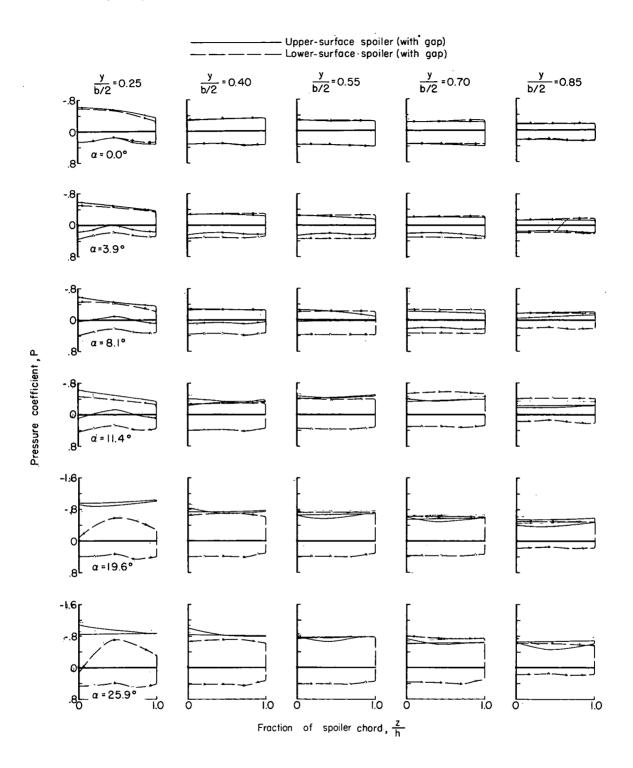
(a) M = 0.60.

Figure 18.- Section pressure distributions on the spoiler; upper-surface spoiler (with gap) compared with the upper-surface (no gap) configuration.



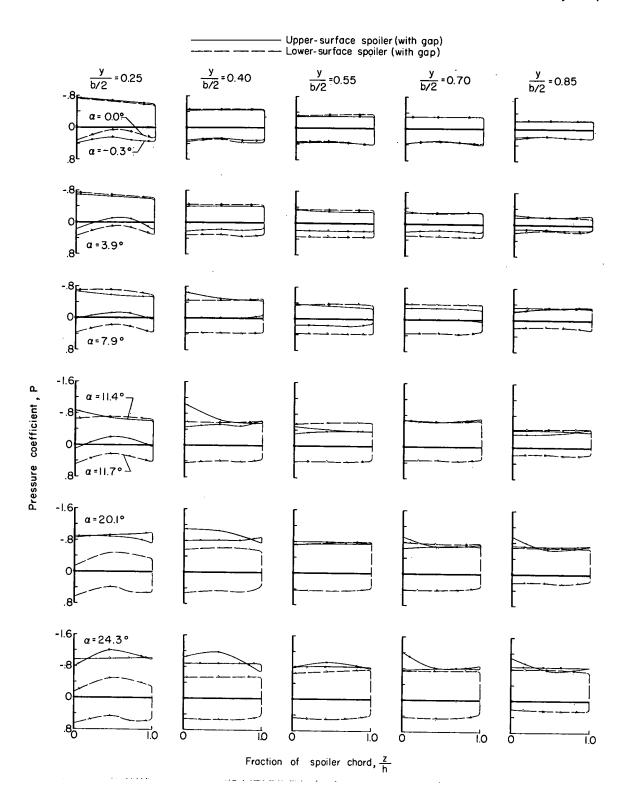
(b) M = 1.00.

Figure 18.- Concluded.



(a) M = 0.60.

Figure 19.- Section pressure distributions on the spoiler; upper-surface spoiler (with gap) compared with the lower-surface spoiler (with gap) configuration.



(b) M = 0.94.

Figure 19.- Concluded.